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# AN INVESTIGATION OF STRESS DETERMINATION FOR AIRCRAFT FATIGUE LIFE ESTIMATION FROM IN-FLIGHT STRAIN DATA

George Michael Horne

# NAVAL POSTGRADUATE SCHOOL
## Monterey, California

# THESIS

An Investigation of Stress Determination for Aircraft Fatigue Life Estimation from In-Flight Strain Data

by

George Michael Horne

September 1976

Thesis Advisor: G. H. Lindsey

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20. Abstract (continued)

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Neuber's theory was evaluated by comparison of experimental stress concentration factors with theoretical values for plates with central holes and was found to be a valid basis for obtaining local stress from nominal strain.

Stress relaxation behavior was obtained for two cyclic loading histories of plate specimens in an effort to extend the monotonic local stress vs. nominal strain relationships into practical use for fatigue life estimation of aircraft structures.

# An Investigation of Stress Determination for Aircraft Fatigue Life Estimation from In-Flight Strain Data

by

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Lieutenant, United States Navy
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Submitted in partial fulfillment of the
requirements for the degree of

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# ABSTRACT

A thorough knowledge of localized stresses due to geometric effects is necessary for accurate fatigue life estimation in aircraft structures. The Department of Aeronautics, Naval Postgraduate School, Monterey, California, has developed a strain monitoring system that provides data on nominal stresses experienced by aircraft structures, which can be applied to obtain local stresses at a stress concentration, provided a local stress vs. nominal strain relationship is available. A theory proposed by Neuber lends itself to development of a method by which local stress can be obtained with knowledge of nominal strain and material properties alone.

Neuber's theory was evaluated by comparison of experimental stress concentration factors with theoretical values for plates with central holes and was found to be a valid basis for obtaining local stress from nominal strain.

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TABLE OF CONTENTS

LIST OF FIGURES

# I. INTRODUCTION AND LITERATURE SURVEY

The ability to predict the fatigue life of aircraft structures accurately and with reliability is of prime concern to structural engineers.

Prior to any attempt to derive a valid fatigue life theory one basic requirement must be satisfied. A thorough knowledge of localized stresses due to geometric effects is necessary and, because these stresses can not be measured directly, an accurate method of determining them analytically must be found which is applicable to a variety of loading situations and configurations. From a more realistic and practical standpoint fatigue life estimation would be greatly facilitated if stresses could be calculated based on actual inflight strain histories, data that are quite easily obtained in realistic situations but difficult to simulate in a laboratory. Specifically, if a relationship between the nominal strain in a structural component and the local stress at a stress raiser could be developed based on structural configuration, then the easily measured nominal strain would provide a local stress. Knowledge of this local stress is critical to fatigue life estimation, since fatigue failures originate at the stress raiser.

In a survey of the literature to determine if a satisfactory method for determining local stress exists, and if such a method would be adequate with only nominal strain and the material properties known, one relationship was frequently encountered. This relationship, postulated by Neuber (Ref. 1), states that the geometric mean of the stress concentration factor, $K_\sigma$, and the strain concentration factor, $K_\varepsilon$, is equivalent to the elastic stress concentration factor, $K_t$, even in nonlinear stress-strain regions, or

$$K_t^2 = K_\sigma K_\varepsilon$$

where

$$K_\sigma = \frac{\text{local stress}}{\text{nominal stress}} = \frac{\sigma}{s}$$

and

$$K_\varepsilon = \frac{\text{local strain}}{\text{nominal strain}} = \frac{\varepsilon}{e}$$

Numerous examples of the adequacy of this relationship in calculating stress-strain curves were found. Crews used a modified form of the relationship in loading sequence tests (Ref. 2), and in a study of stress-strain behavior at notch roots (Ref. 3), and found the stresses thus calculated correlated very closely with experimentally determined stresses. Wetzel studied the accuracy of the relationship with three different types of data taken in experiments involving smooth specimen simulation of fatigue behavior of notches (Ref. 4). Morrow *et al*., also confirm the validity of the equation by comparison of fatigue failures at different stress concentration factors (Ref. 5). Since the Neuber relationship involved the factors of interest in this investigation and in light of past results, it appeared that Neuber's theory might prove to be a satisfactory basis on which to establish a method for calculating local stress using nominal strain.

During the course of this study the requirement for a stress-strain relationship was expected. However, due to the dependence of fatigue on cyclic loading a monotonic stress-strain relationship alone appeared to be insufficient and a cyclic-stress-strain curve would be required in addition to the monotonic one. Landgraf et al., conducted a literature survey to determine whether an exact definition of a cyclic stress-strain curve existed and found none (Ref. 6). They did, however, propose an incremental step test method for determining a cyclic stress-strain curve whereby a uniaxial specimen is taken into tensile yield, then compressive yield, then cycled in tension and compression to decreasing values of strain. The locus of the maximum values of strain obtained from the resulting hysteresis loops forms the cyclic stress-strain curve. This method was adopted for use in this study.

Because the local stress-nominal strain relationships expected to be developed from this study would eventually be applied to actual aircraft structures under fatigue analysis, a specimen representing a realistic structural component was sought. A plate with a central hole was decided upon due to its commonality in almost all aircraft structures. The plate specimen was expected to provide local and nominal stress and strain data which would be representative of that found in actual structural components.

During the literature survey a dependence of fatigue on loading history was shown by Crews (Ref. 2), Naumann (Refs. 7 and 8), Schijve (Refs. 9 and 10), and Potter (Ref. 11). In order to establish a basic foundation of knowledge from which to extend into more sophisticated loading histories, two simple loading situations were proposed. Single and dual, or repetitive high-low, amplitude loading programs were chosen to compare the effects of different loading histories on stress relaxation behavior, a necessary factor in the determination of local stress in cyclic loading situations. In the tests proposed, the requirement for compressive loading of the plates was ruled out on the basis that in actual aircraft structures such loads, in the higher stress regions to be encountered, would cause buckling and change the nature of the investigation entirely.

In summary, this investigation was directed toward determining a method whereby local stress can be calculated from knowledge of nominal strain and material properties alone. Local stress relaxation behavior was examined in order to extend the local stress vs. nominal strain relationship to the determination of local stress under cyclic loading and thereby give it practicality in future fatigue life determination studies.

# II. STRESS-STRAIN DATA ON UNIAXIAL SPECIMENS OF 7075 T-6 ALUMINUM

## A. INTRODUCTION

In order to provide a sound data base for comparison with data obtained in tests on plate specimens with central holes (Figs. 1 and 2), uniaxial specimens of 7075 T-6 aluminum (Figs. 3 and 4) were subjected to three different tests. The first test was designed to obtain monotonic and cyclic stress-strain curves. The second and third were single and dual amplitude cyclic loading tests, which were designed to obtain monotonic stress-strain curves and stress relaxation data under two different types of loading. The single amplitude cyclic loading test was designed to repeatedly load the specimen to a predetermined strain, which remained constant throughout the test. The dual amplitude cyclic loading test was designed to alternately load the specimen to two predetermined strains, one approximately twice the magnitude of the other.

The data obtained from the uniaxial specimens were considered indicative of the properties of the material at the location of stress concentration factor (Ref. 3) and would provide a consistent and readily duplicated basis for determining behavior of other specimens of the same material but of arbitrary geometry and stress concentration factors.

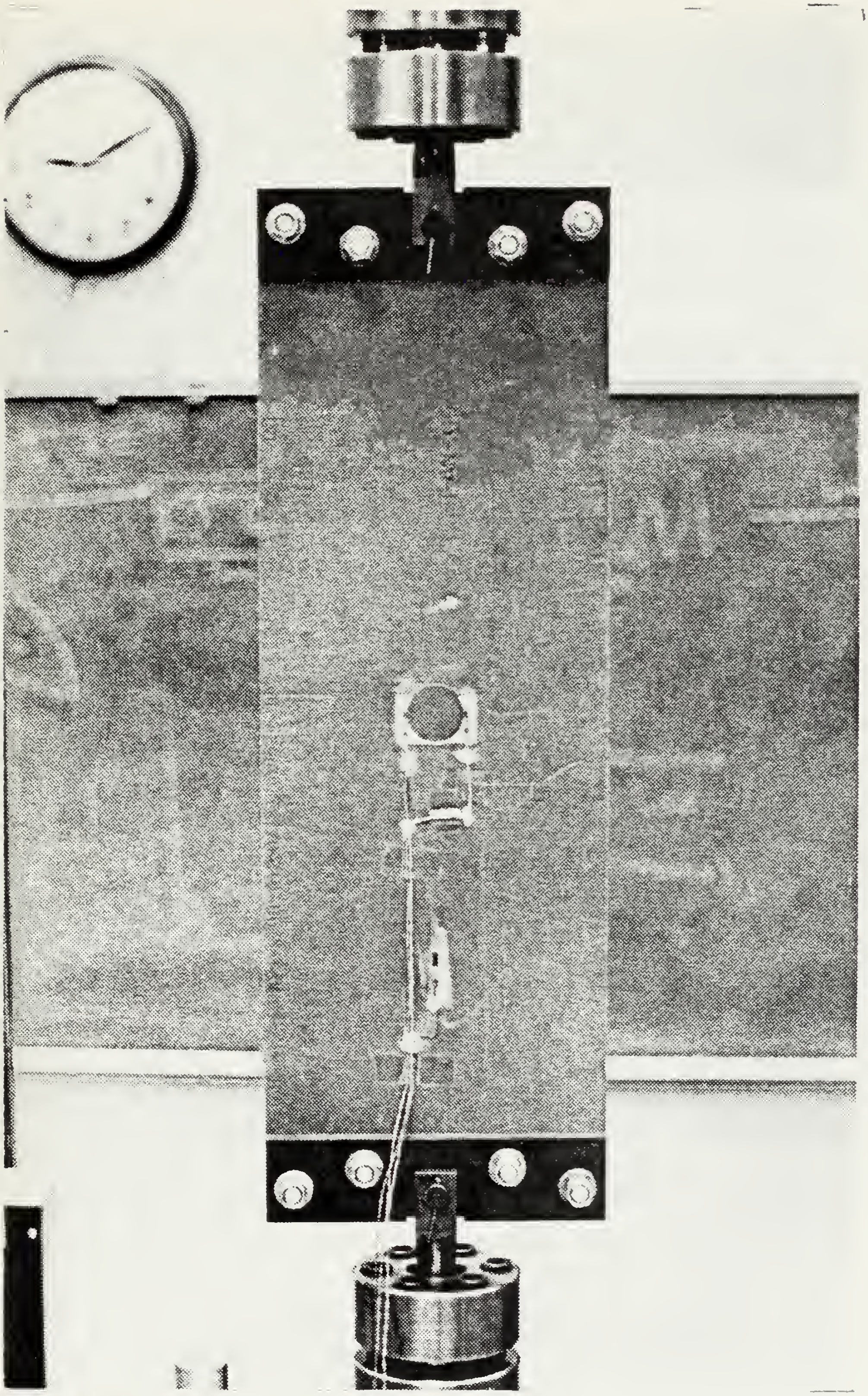

Figure 1

Photo of Plate Specimen

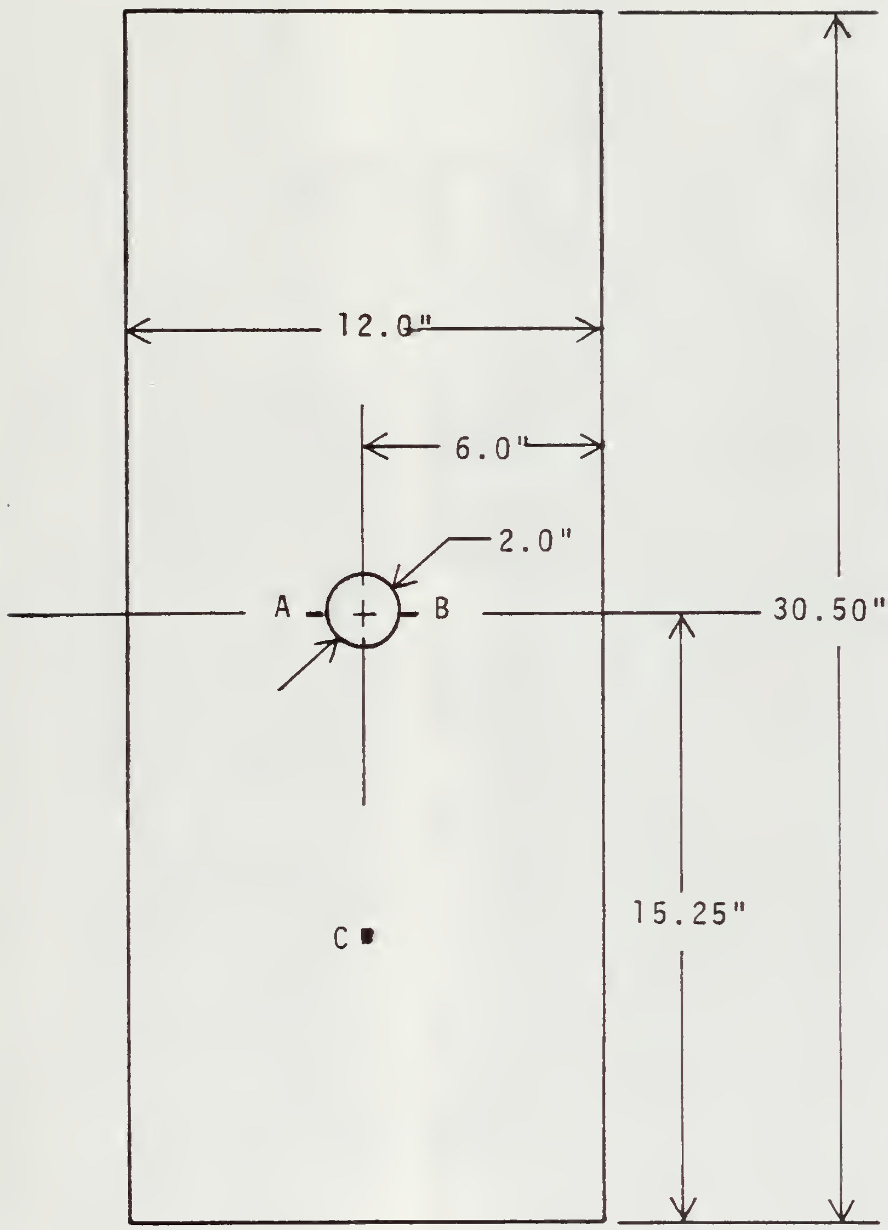


Figure 2

Plate specimen with central hole
constructed of 7075 T-6 aluminum

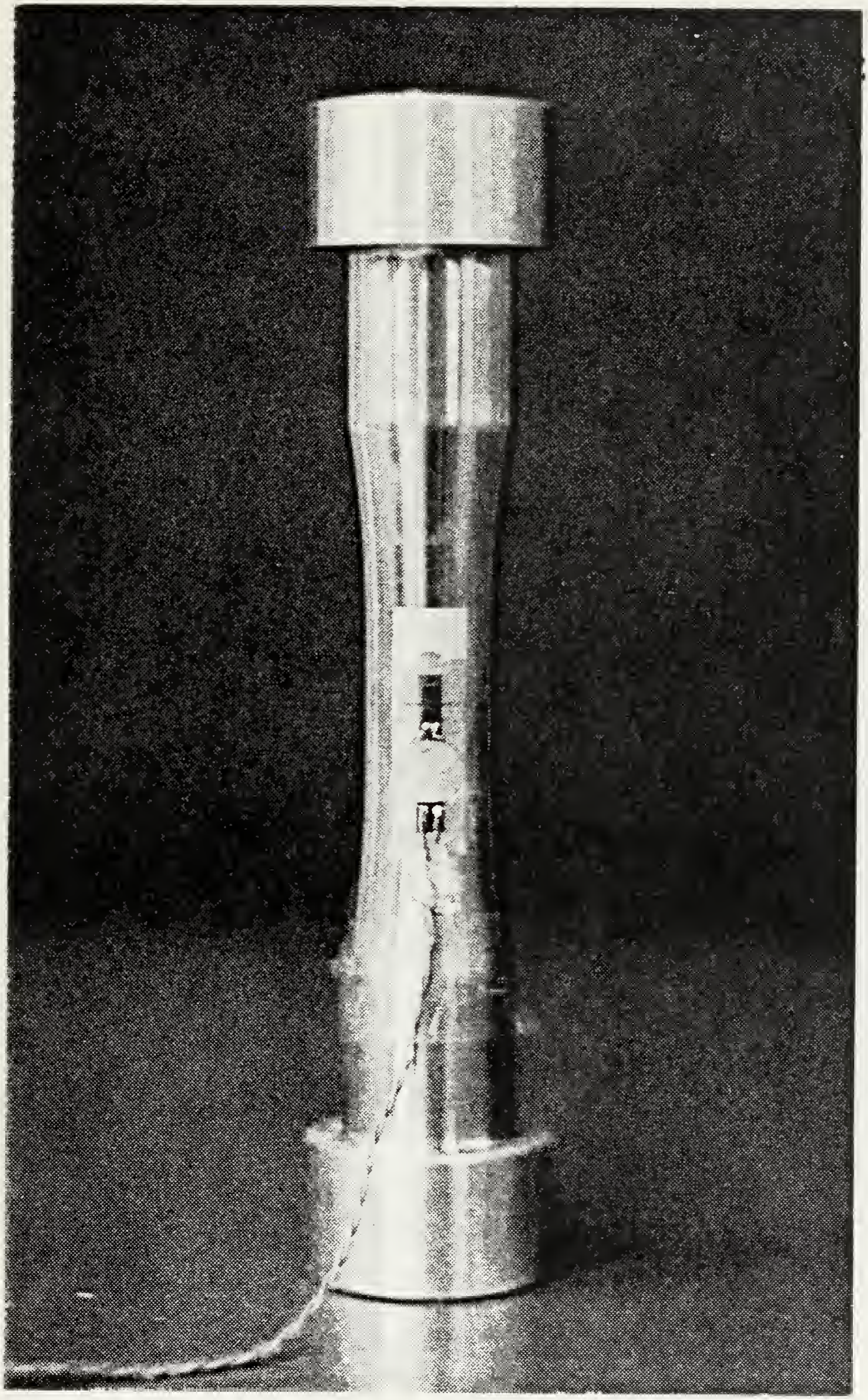

Figure 3

Photo of Uniaxial Specimen

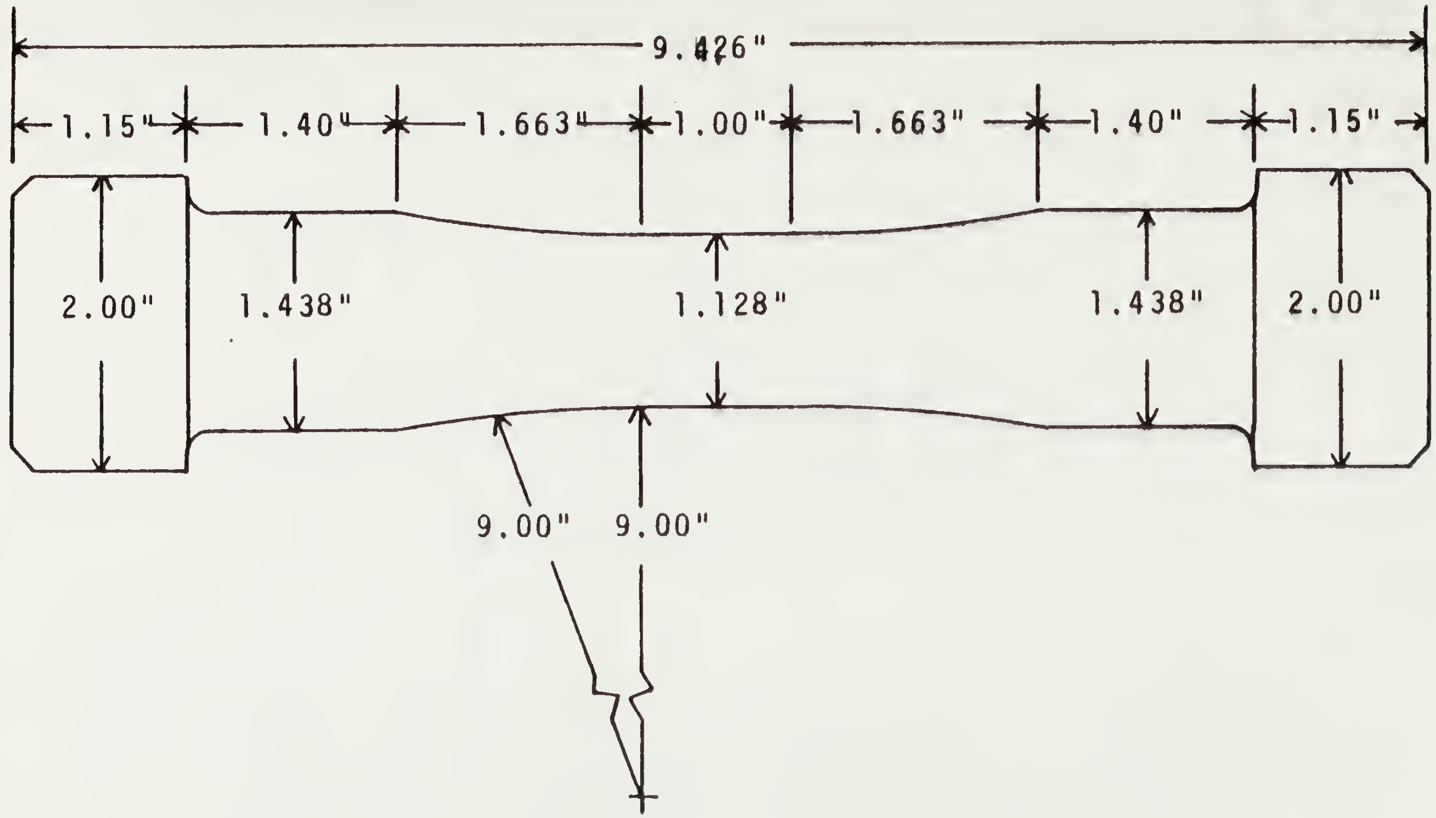


Figure 4

Uniaxial Specimen
7075 T-6 Aluminum

The testing of the uniaxial specimens was done using an MTS Systems Corporation closed-loop, servo-controlled testing system (Figs. 5 and 6). The system was driven under strain control by an internal function generator or by an Electronics Associates, Incorporated PACE TR-20 analog computer. Outputs of voltages representing load and strain on the specimen being tested were input to a Hewlett-Packard X-Y recorder and a Hewlett-Packard dual trace strip chart recorder.

The uniaxial specimens used in the tests were constructed of 7075 T-6 aluminum in accordance with ASTM recommendations (Ref. 12). Each specimen had a test section cross-sectional area of one square inch in order that load might be interpreted directly as stress on the specimen. Strain gages were mounted as shown in Figure 3.

Prior to the actual tests, an alignment check, as described by ASTM (Ref. 12), was performed on the load cell test bed to ensure that strains due to bending from misalignment would not be introduced into the specimens. The maximum percent bending ranged from 2.37 to 4.43 percent (Table 1), with the average being 3.30 percent. This is within the 5.0 percent maximum allowable bending moment recommended by ASTM.

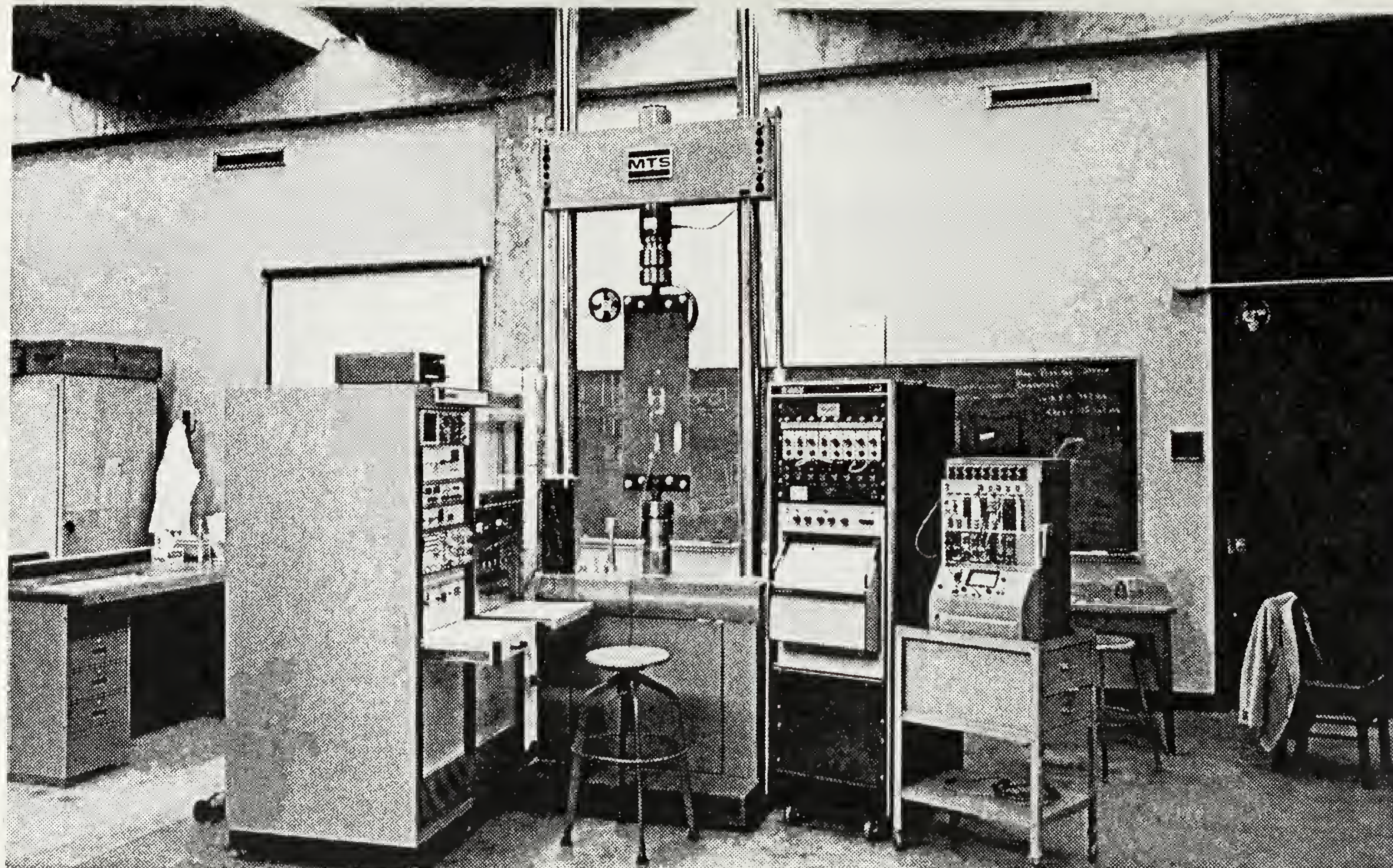


Figure 5
Photo of MTS System

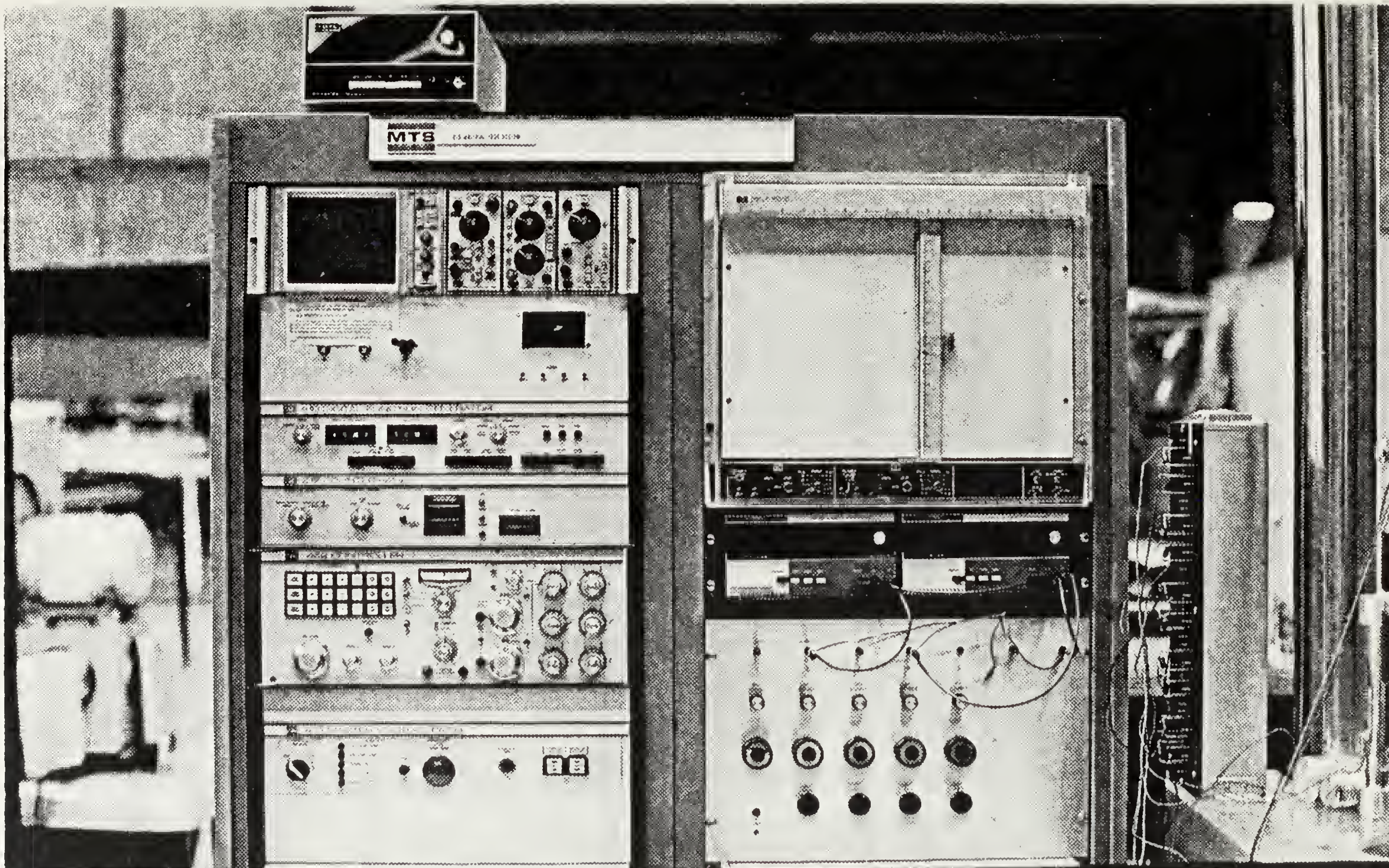


Figure 6
Photo of MTS System

Specimens were mounted and secured in the test system in accordance with current MTS Corporation instructions (Ref. 13). Care was taken to ensure that each specimen was not damaged nor yielded prior to any test.

Before each test was begun, strain gages mounted on the specimen were zeroed and calibrated while the specimen was hanging unattached in the machine, and according to standard practice. The load cell output voltage was also zeroed at this time. All recorders in use for a particular test were calibrated with a known input voltage to ensure accurate reproduction of voltage outputs from the test system.

## B. CYCLIC STRESS-STRAIN CURVE TESTS

### 1. Description of Test

To obtain the desired monotonic and cyclic stress-strain curves for 7075 T-6 aluminum, an incremental step test similar to that proposed by Landgraf et al. (Ref. 6) was utilized.

The closed-loop, servo-controlled material testing system used did not have a function generator capable of providing a periodic, decreasing amplitude function required for the incremental step test. To obtain such a function to drive the testing system, an analog computer and the beat phenomena, obtained from summing two sinusoidal functions, were employed (Ref. 14).

Two sinusoidal functions

$$X_1(t) = R_1 \ \mathrm{Cos}(\omega_1 t)$$

and

$$X_2(t) = R_2 \ \mathrm{Cos}(\omega_2 t)$$

generated by an analog computer and summed provide a resultant output of

$$X(t) = R_1 \ \mathrm{Cos}\omega_1 t + R_2 \ \mathrm{Cos}[(\omega_1 - \Delta\omega)t]$$

or

$$X(t) = R \ \mathrm{Cos}(\omega_1 t + \phi)$$

where

$$R = R_1 + R_2, \ \omega_1 > \omega_2, \ \omega_1 \neq \omega_2, \ \Delta\omega = \omega_1 - \omega_2,$$

and

$$\emptyset = \mathrm{Tan}^{-1} \left[- \frac{R_2 \mathrm{Sin}(\Delta\omega t)}{R_1 + R_2 \ \mathrm{Cos}(\Delta\omega t)}\right].$$

By appropriate selection of the variables $R_1$, $R_2$, $\omega_1$, and $\omega_2$ the resultant output will oscillate at $\omega_1$ between $R_1 + R_2$ and $R_1 - R_2$ at a rate of $\Delta\omega$ (Fig. 7). Beginning at the maximum amplitude and continuing for approximately one half the beat period, $\frac{\Delta t}{2}$, a cyclic decreasing amplitude function is obtained.

Both the analog computer and the material testing system operate on $\pm$ 10.0 VDC. To utilize the full capability of the system, a maximum amplitude of R = $\pm$ 10.0 VDC and a minimum of R = 0.0 VDC were desired. Thus, $R_1$ = 5.0 VDC and $R_2$ = 5.0 VDC were chosen for maximum input amplitudes. A beat period of $\Delta t$ = 80 s/c was selected and considered

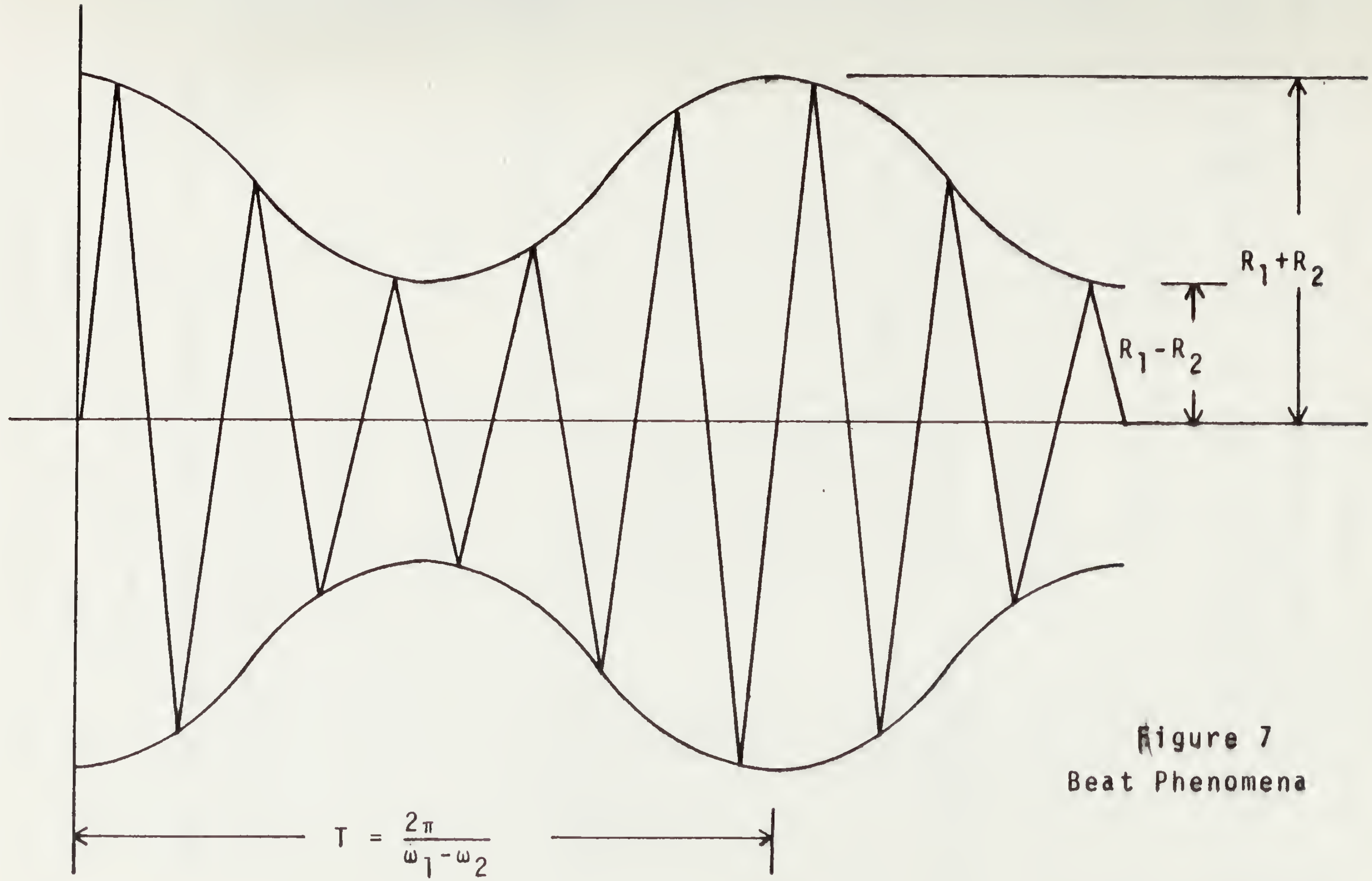


Figure 7
Beat Phenomena

adequate to remain within testing system and recorder limitations. Likewise, a cyclic frequency of $f = \frac{1}{4}$ c/s was desired to provide the cyclic output function period of 4 c/s. Thus, $\omega_1 = \frac{\pi}{2}$ rad/s was assumed, fixing $\omega_2 = \frac{19}{40}\pi$ rad/s since $\Delta w = \frac{\pi}{40}$ rad/s. The two input functions used were

$$X_1(t) = 5.0 \text{ Cos } (\tfrac{\pi}{2}t)$$

and

$$X_2(t) = 5.0 \text{ Cos } (\tfrac{19}{40}\pi t).$$

To produce these functions, differential equations for analog solution were programmed as follows:

$$\ddot{X}_1(t) = - 2.4674 \ X_1(t)$$

and

$$\ddot{X}_2(t) = - 2.2268 \ X_2(t).$$

In constructing the scaled analog solution, the actual equations used were

$$\ddot{X}_1(t) = - 0.24674 \ X_1(t)$$

and

$$\ddot{X}_2(t) = - 0.22268 \ X_2(t).$$

These provided a more satisfactory beat period of $\Delta t = 252.95$ s/c. Thus the cyclic period became 12.65 s/c or twenty oscillations in one beat period.

The output of the analog computer was supplied as input to the controller of the material testing system under strain control. Initially the output amplitude of the analog computer was set to zero by zeroing the initial conditions on the input functions. By manually increasing the initial conditions on each input function to the values calculated for solution, the output was increased to + 10.0 VDC, putting the specimen into tensile yield. Reversing the procedure and manually decreasing the initial conditions to zero, reversing the polarity of the output and manually increasing the initial conditions to the calculated values, produced a - 10.0 VDC output which placed the specimen in a compressive yield condition with the strain equal to the initial tensile yield strain. This was done to provide a symmetric cyclic stress-strain curve.

The analog computer was activated with the specimen in compressive yield and allowed to cycle until the load-strain curve plotted by the X-Y recorder became linear through the origin, at which time the test was terminated (Fig. 8). This occurred after approximately nine cyclic oscillations.

Output voltages representing load and strain on the specimen were input to an X-Y recorder to provide a series of load, or stress, versus strain curves throughout the test. The uniaxial specimen was constructed with a test section cross-sectional area of one square inch, thus allowing load to be interpreted directly as stress, and load-strain curves as stress-strain curves.

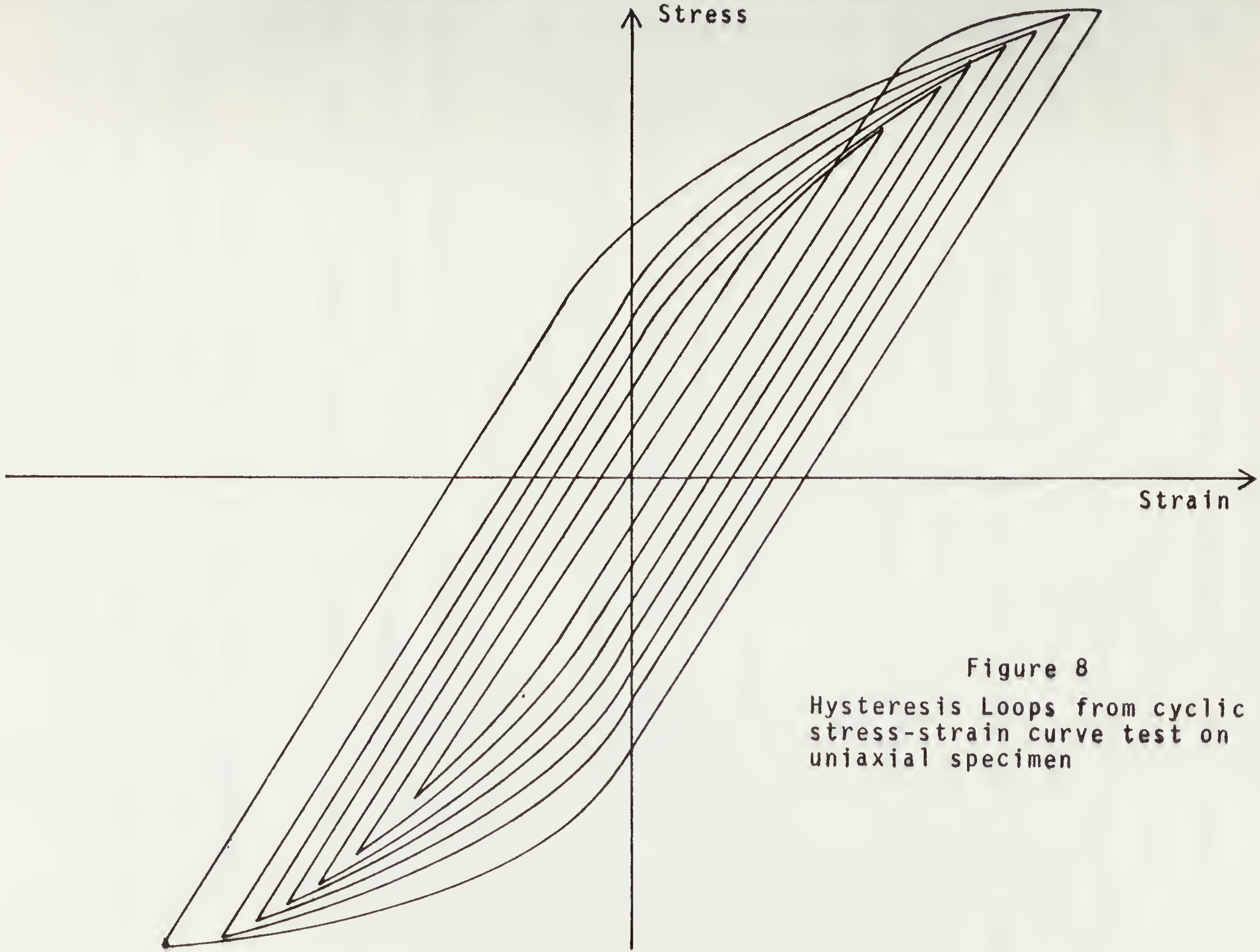


Figure 8

Hysteresis Loops from cyclic stress-strain curve test on uniaxial specimen

## 2. Test Results

The X-Y recorder plot of output voltages of stress and strain provided a series of hysteresis loops, each with a maximum strain amplitude less than the preceding loop (Fig. 8). The locus of the tips of these loops, when plotted in terms of stress in $Lbf/in^2$ and strain in $\mu in/in$, is the desired cyclic stress-strain curve (Fig. 9 and Table 2). The slightly "S" shaped curve is symmetrical about the origin in tension and compression. The modulus of elasticity calculated from the linear portion of the curve was $E = 10.18 \times 10^6$ $Lbf/in^2$.

The initial loading of the specimen provided voltage outputs of stress and strain with which to construct the monotonic stress-strain curve (Figs. 9 and 10 and Table 3). The modulus of elasticity calculated from this curve was $E = 10.67 \times 10^6$ $Lbf/in^2$.

The two percent yield stress obtained from the monotonic stress-strain curve is 78,000 $Lbf/in^2$. This yield stress and the modulus of elasticity compare favorably with the theoretical values generally accepted to be 77,000 $Lbf/in^2$ and $E = 10.3 \times 10^6$ $Lbf/in^2$.

Values of stress and strain from the monotonic curve were used to produce a stress x strain ($\sigma\varepsilon$) vs. stress curve (Fig. 11 and Table 3). This curve was for later use in the plate tests.

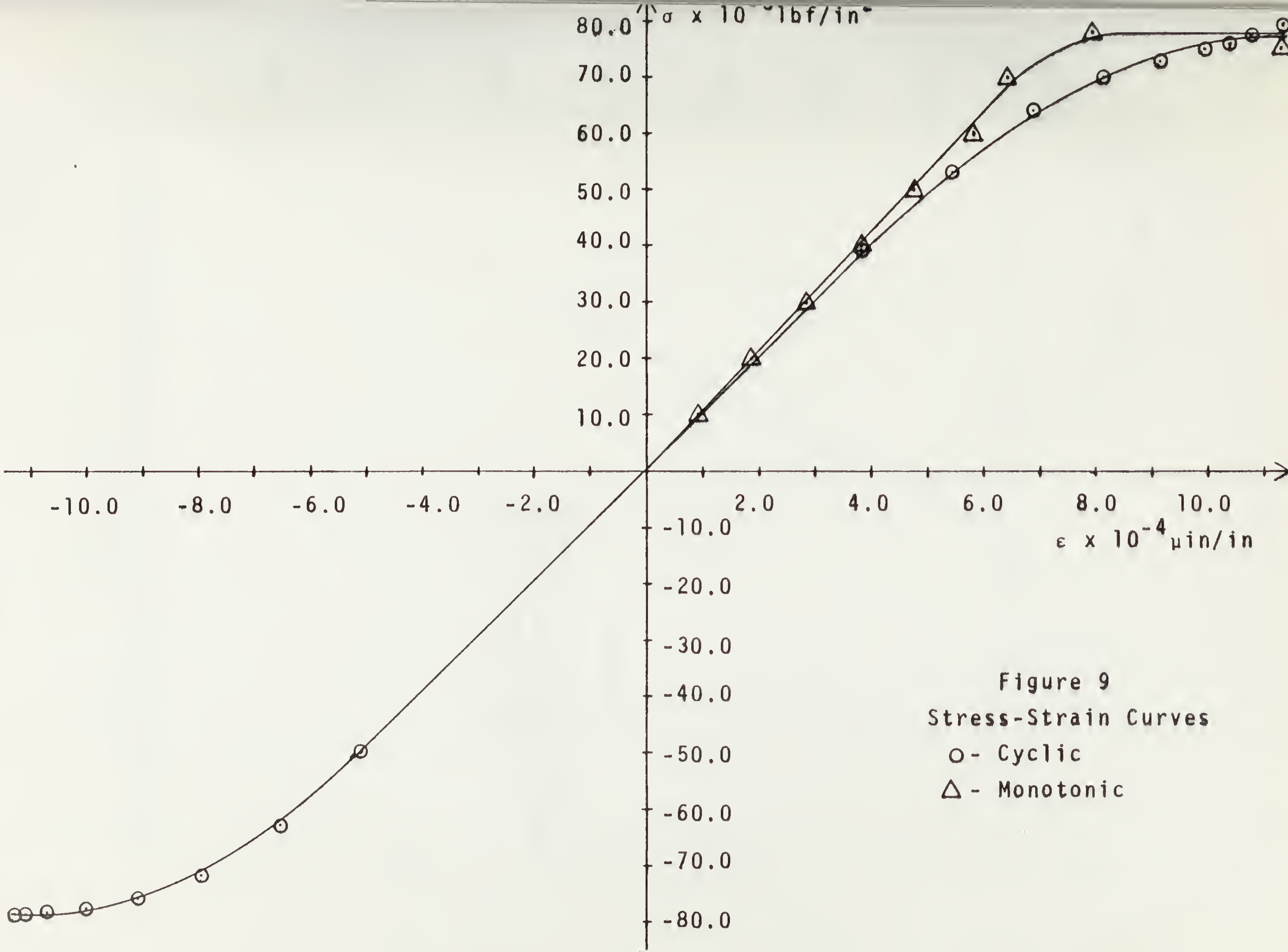


Figure 9
Stress-Strain Curves
○ - Cyclic
△ - Monotonic

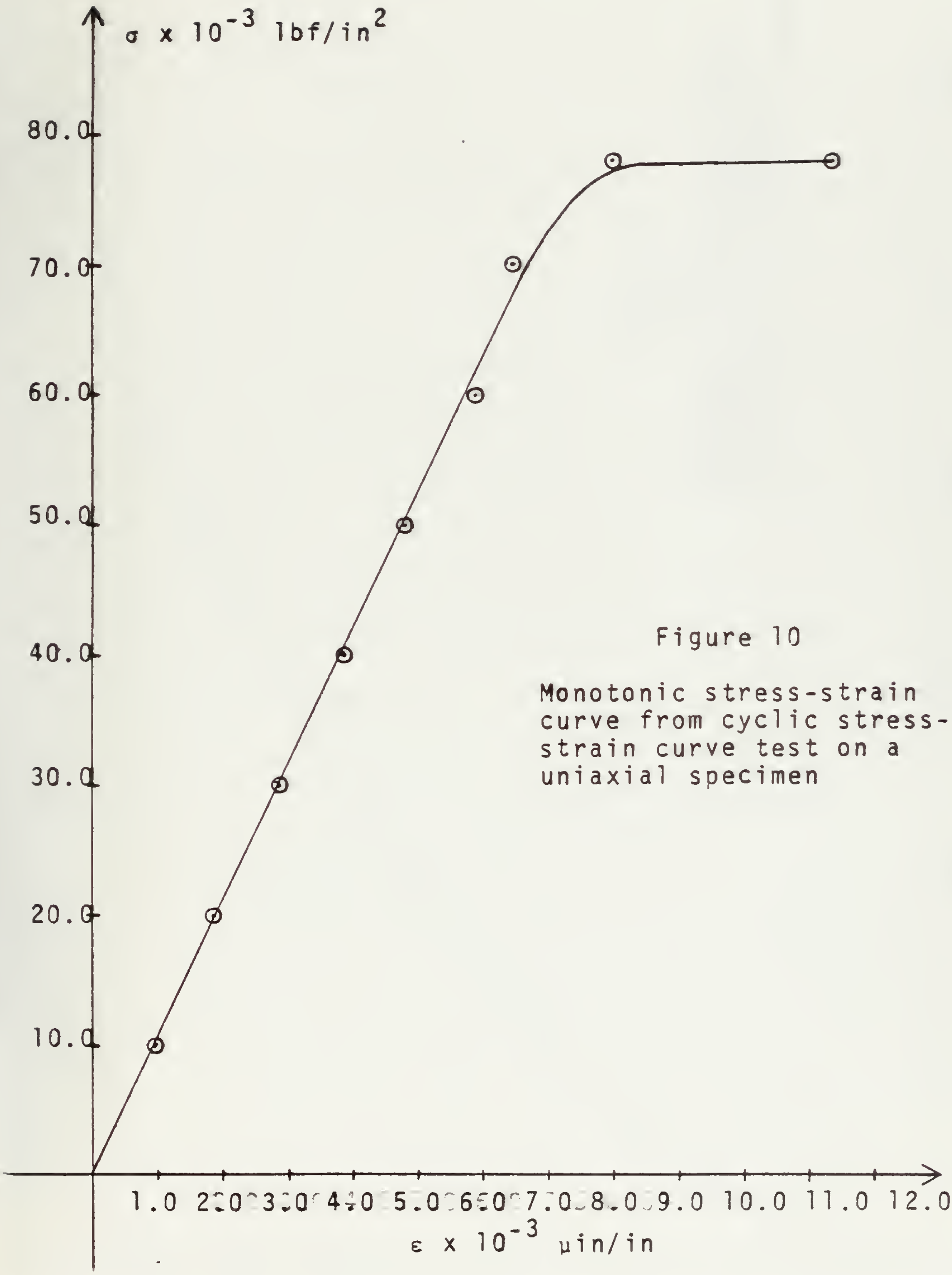


Figure 10

Monotonic stress-strain curve from cyclic stress-strain curve test on a uniaxial specimen

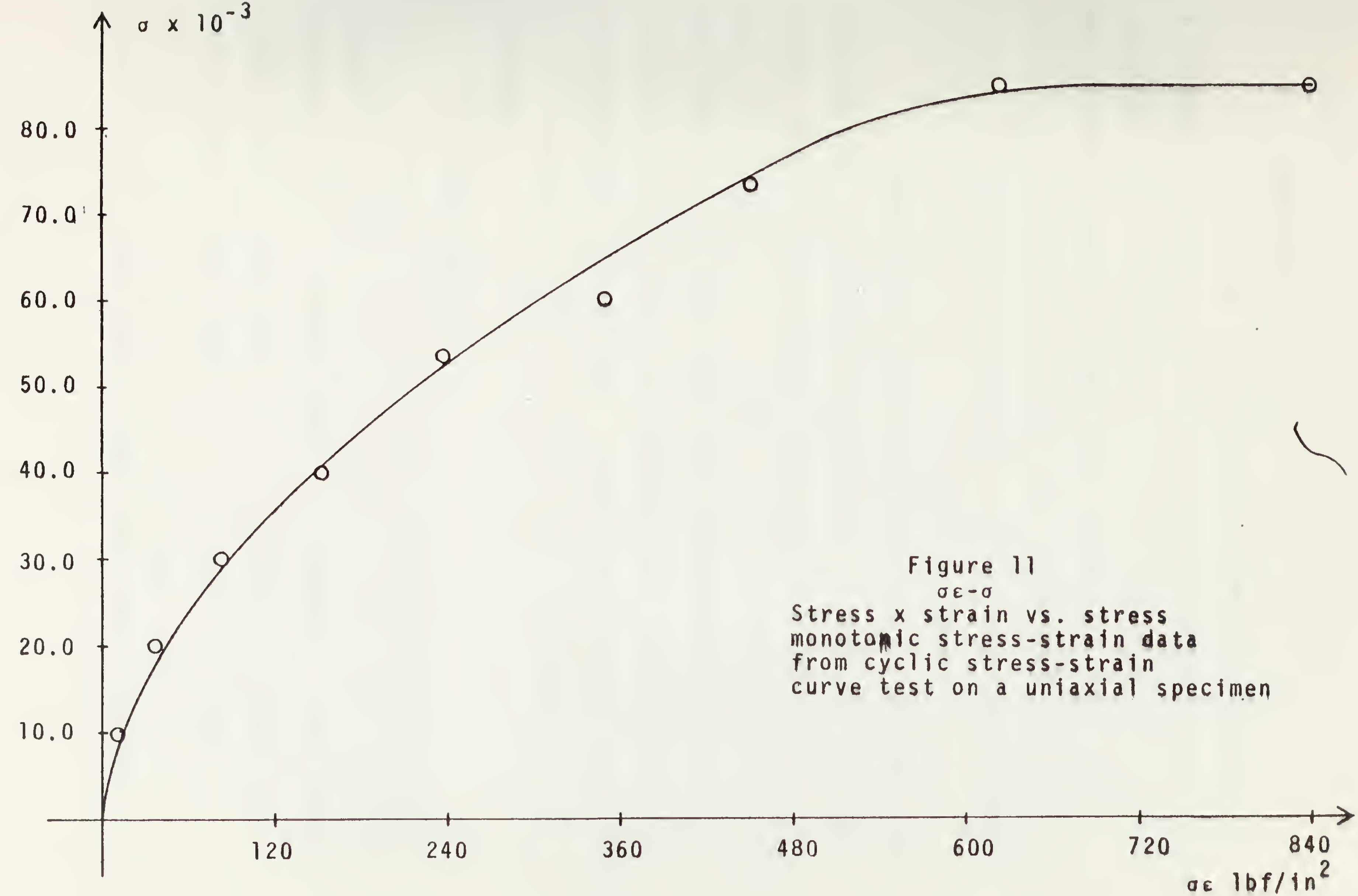


Figure 11

$\sigma\varepsilon - \sigma$

Stress x strain vs. stress
monotomic stress-strain data
from cyclic stress-strain
curve test on a uniaxial specimen

## C. SINGLE AMPLITUDE CYCLIC LOADING TEST

### 1. Description of Test

A knowledge of the stress relaxation behavior in a uniaxial specimen of 7075 T-6 aluminum subjected to single amplitude cyclic loading was required for comparison with relaxation behavior in the plate specimens. The initial loading cycle furnished stress and strain data for construction of a monotonic stress-strain curve, which was compared with other similar curves previously mentioned.

The function generator installed in the MTS system was capable of producing a haversine function to drive the system under strain control. Maximum amplitude of the haversine function was set to provide 7.3 VDC of the 10.0 VDC available. This input amplitude corresponded to 11167 $\mu$ in/in strain in the specimen. The specimen was cycled 275 times to the maximum strain value. A dual trace strip recorder and an X-Y recorder were used to plot load and strain output voltages.

### 2. Test Results

The plot of stress and strain provided by the X-Y recorder allowed calculation of data points from the initial loading cycle for construction of a monotonic stress-strain curve (Fig. 12 and Table 4). The modulus of elasticity calculated from the linear portion of the curve was $E = 10.0 \times 10^6$ lbf/in$^2$. The yield stress was found to be 80,000 lbf/in$^2$. Values of stress and strain from this curve

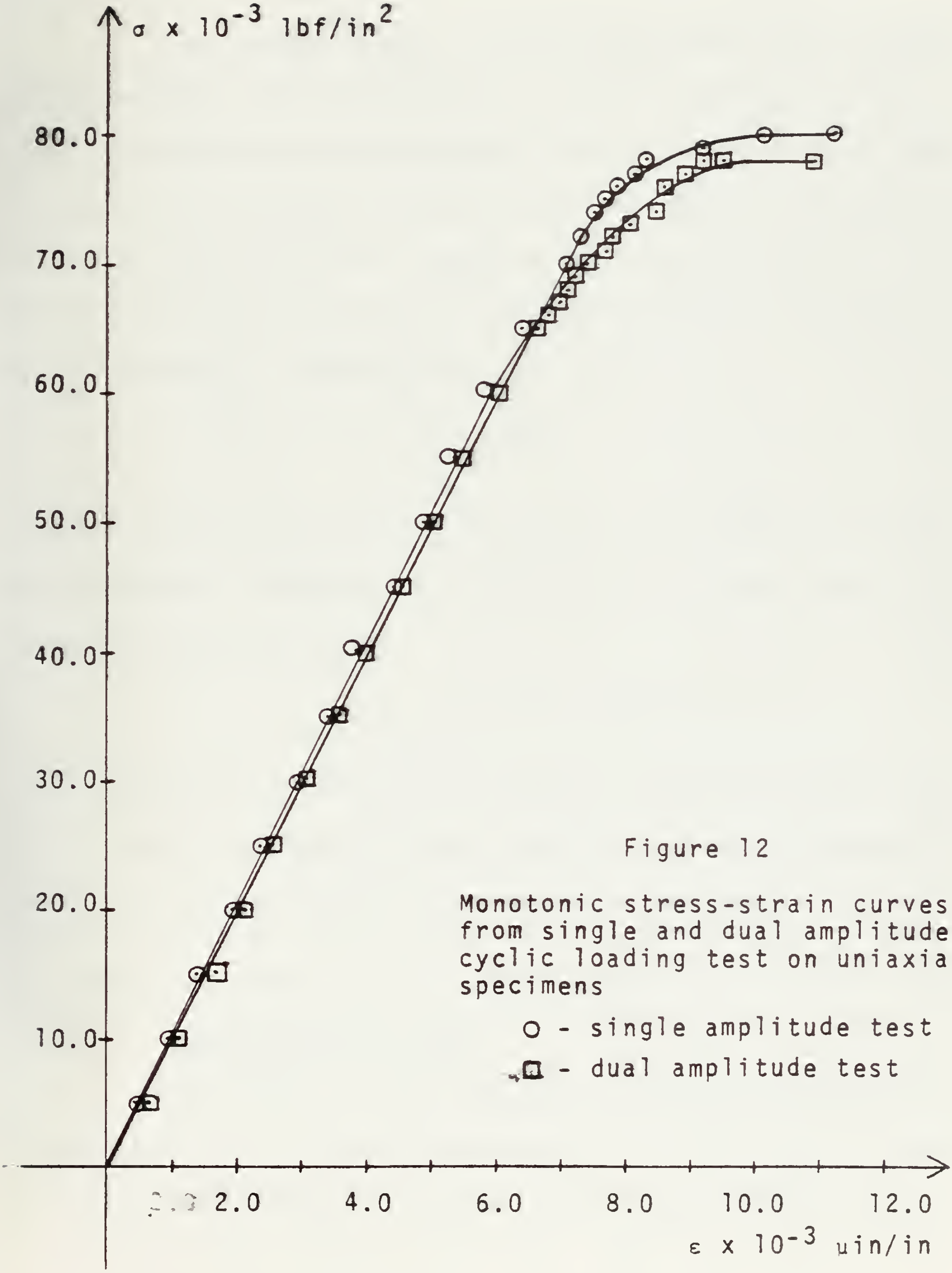


Figure 12

Monotonic stress-strain curves from single and dual amplitude cyclic loading test on uniaxial specimens

○ - single amplitude test

□ - dual amplitude test

were used to construct a stress x strain ($\sigma\varepsilon$) vs. stress curve (Fig. 13 and Table 4) for later use in the plate tests and for comparison with those of the other uniaxial specimen tests.

The stress output voltages obtained from the dual trace recorder were converted to $lbf/in^2$ (Table 5) and plotted against the cycle number, N, on semilog graph paper (Fig. 14) to indicate stress relaxation behavior graphically. The locus of data points appeared to form a straight line indicating stress relaxation behavior could be represented by an exponential equation of the form

$$\sigma = \sigma_o e^{-bN} .$$

A least squares exponential curve fit calculator algorithm was applied to the data on 253 points out of 275 taken. The resulting equation was

$$\sigma = 73160 \; e^{-(3.177 \times 10^{-3})N} .$$

A correlation coefficient of 0.994 was calculated for the fit. Thus the relaxed stress value at any cycle number, N, could be obtained with a high degree of accuracy.

## D. DUAL AMPLITUDE CYCLIC LOADING TEST

### 1. Description of Test

Knowledge of the effects of dual amplitude cyclic loading on stress relaxation behavior in uniaxial specimens of 7075 T-6 aluminum was desired for comparison with the

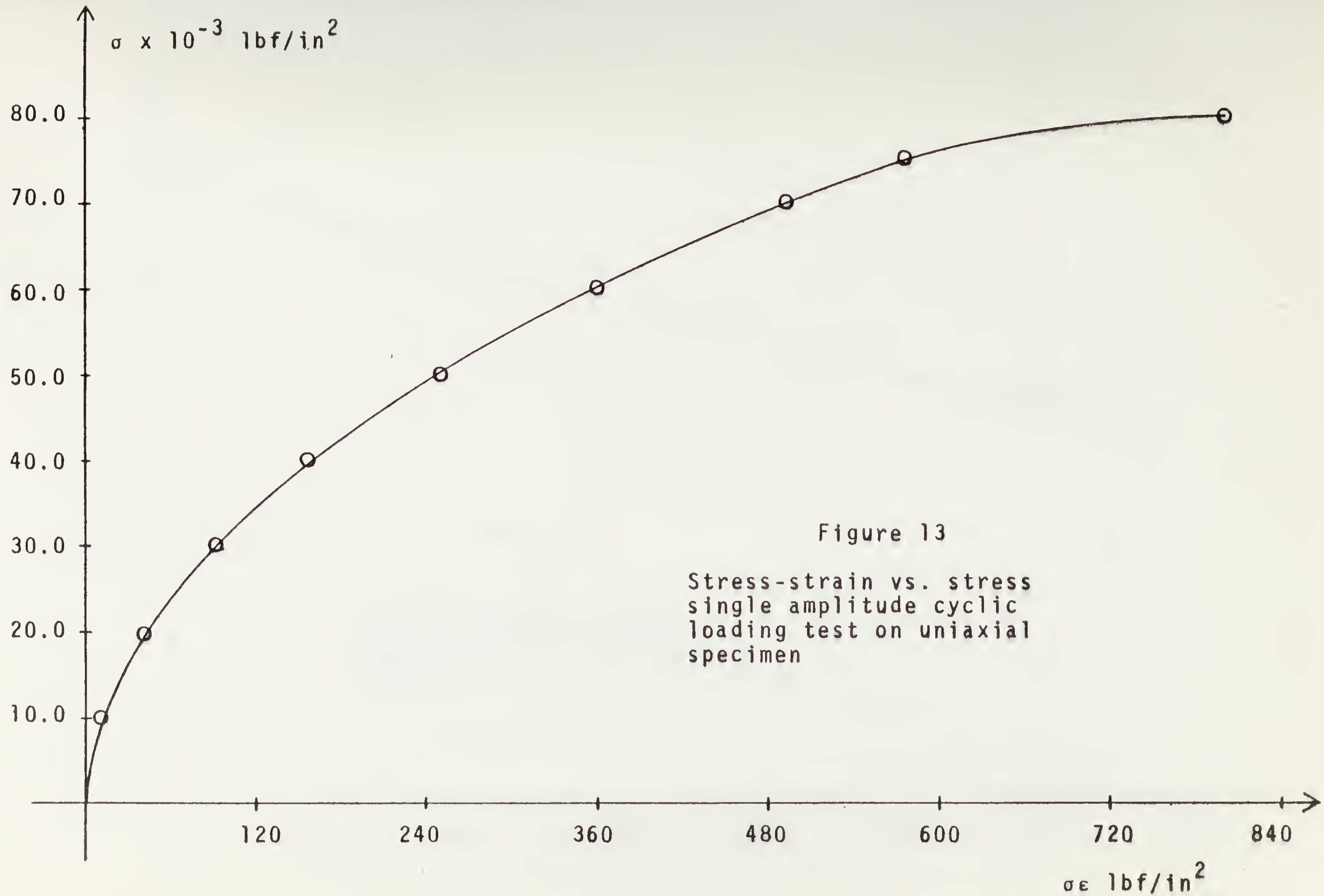


Figure 13

Stress-strain vs. stress single amplitude cyclic loading test on uniaxial specimen

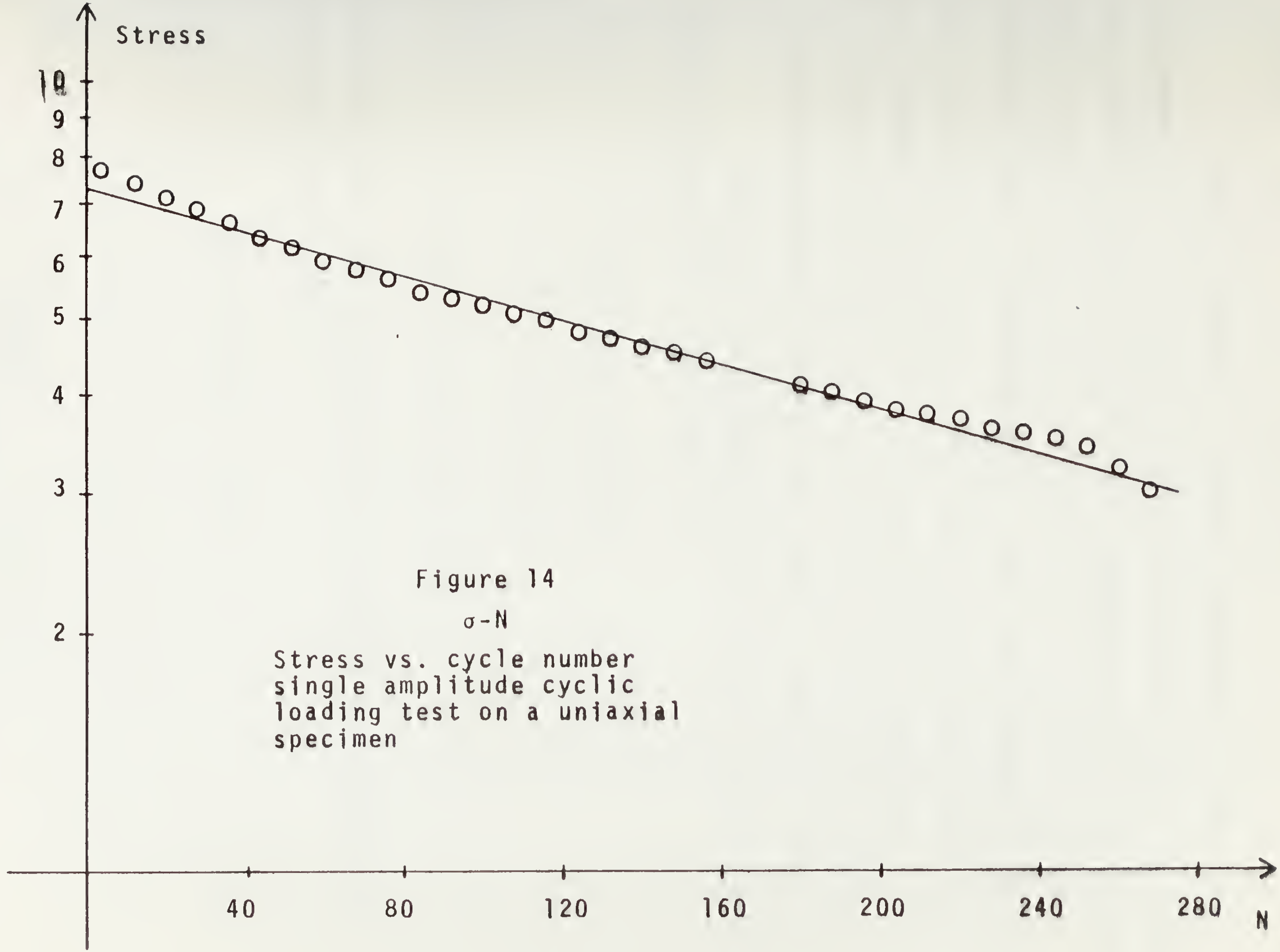


Figure 14

σ-N

Stress vs. cycle number single amplitude cyclic loading test on a uniaxial specimen results of the single amplitude cyclic loading test on a uniaxial specimen and the plate tests.

The MTS system's function generator did not have the capability of producing a dual amplitude, cyclic function. The analog computer and the beat phenomena used in the cyclic stress-strain curve test were applied to the problem in a modified form. A function with a low, positive amplitude of one half that of the high amplitude was desired (Fig. 15). For optimum utilization of the system this required a maximum high amplitude output voltage of + 10.0 VDC, thus fixing the maximum low amplitude output voltage of + 5.0 VDC.

From the development of the function for the cyclic stress-strain curve test (Ref. 14)

$$X_1(t) = R_1 \text{ Cos } (\omega_1 t)$$

$$X_2(t) = R_2 \text{ Cos } (\omega_2 t)$$

and

$$X(t) = R \text{ Cos } (\omega_1 t + \phi).$$

Summing of the two functions, $X_1(t)$ and $X_2(t)$, produced the resultant, $X(t)$, where

$$X(t) = X_1(t) + X_2(t)$$

Also,

$$R = [R_1^2 + R_2^2 + 2\ R_1\ R_2 \text{ Cos } (\Delta\omega t)]^{\frac{1}{2}},$$

and

$$\text{TAN } \emptyset = \frac{R \text{ Sin } \emptyset}{R \text{ Cos } \emptyset} = \frac{-R_2 \text{ Sin } (\Delta\omega t)}{R_1 + R_2 \text{ Cos } (\Delta\omega t)}$$

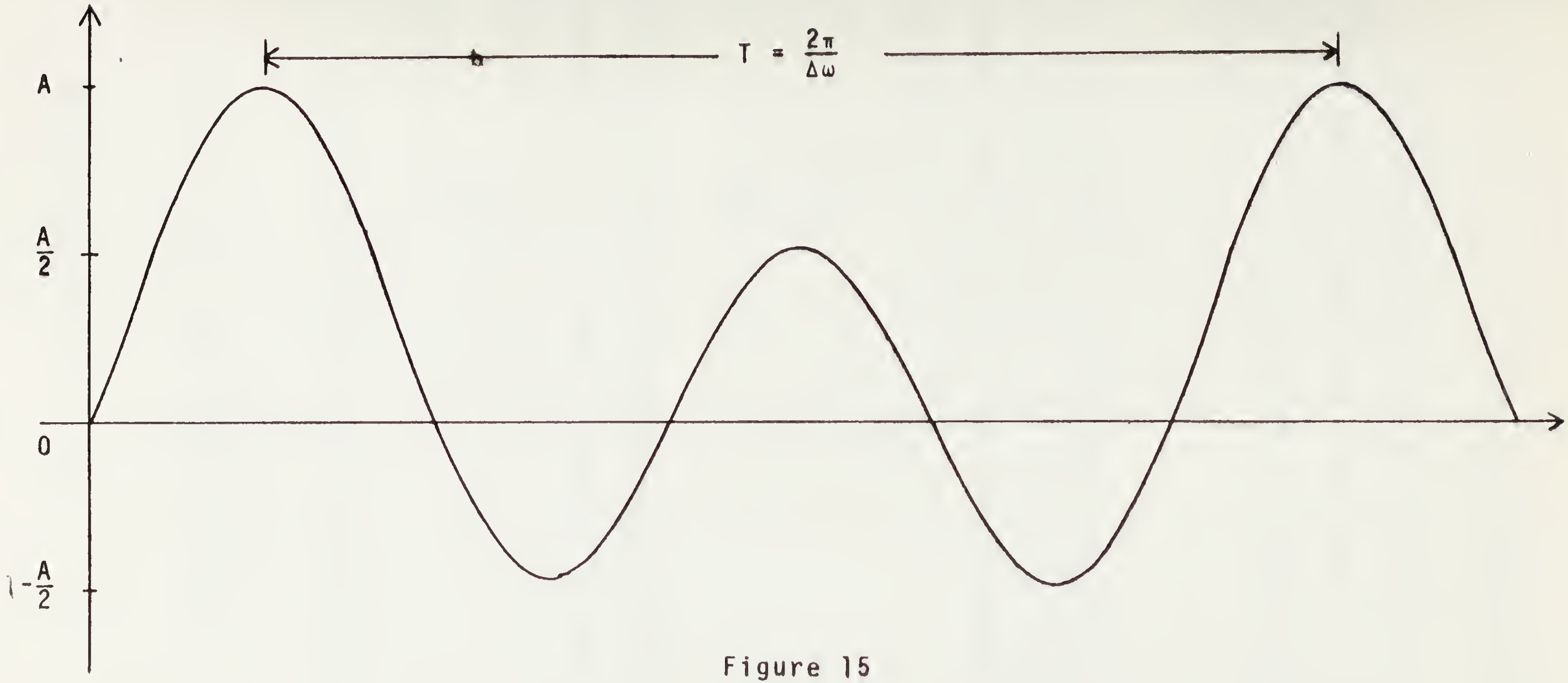


Figure 15

Dual amplitude input function

where

$$\Delta\omega = \omega_1 - \omega_2 \text{ and } T = 2\pi/\Delta\omega \text{ can be written.}$$

Consideration of Figure 15 and the conditions that $R = 10.0$ at the high amplitude output and $R = 5.0$ at the low amplitude output allowed constraint equations to be written in the form

$$X(t) + \Delta = R$$

where $\Delta$ was a constant voltage added to give an additional degree of freedom with which to force the resultant output into a dual amplitude wave form. The constraint equations obtained were

$$X(0) + \Delta = 10$$

and

$$X(\tfrac{T}{2}) + \Delta = 5.$$

An additional constraint equation was obtained from the negative portion of the desired waveform, where $R = -5.0$ was arbitrarily chosen such that

$$X(\tfrac{T}{4}) + \Delta = -5.$$

The application of

$$R(t) = [R_1^2 + R_2^2 + 2 R_1 R_2 \operatorname{Cos} (\Delta\omega t)]^{\frac{1}{2}}$$

$$\emptyset (t) = \operatorname{Tan}^{-1} [\frac{-R_2 \operatorname{Sin} (\Delta\omega t)}{R_1 + R_2 \operatorname{Cos} (\Delta\omega t)}]$$

and

$$X(t) = R(t) \operatorname{Cos} (\omega_1 t + \emptyset)$$

to the above constraint equations at $t = 0$, $t = \frac{T}{4}$, and $t = \frac{T}{2}$ gave rise to three equations in three unknowns for solution. For these calculations $\omega_1 = 2\Delta\omega$ was desired for only two amplitudes to be produced per cycle.

At $t = 0$,

$$\emptyset(0) = 0,\ R(0) = R_1 + R_2,\ \text{and } X(0) = R(0) = R_1 + R_2$$

and

$$R_1 + R_2 + \Delta = 10.$$

At $t = \frac{T}{4}$,

$$\Delta\omega t = \Delta\omega\frac{T}{4} = \frac{\pi}{2},\ \text{and } \omega_1 t = \pi,$$

then,

$$\emptyset(\tfrac{T}{4}) = \frac{-R_2}{R_1},\ R(\tfrac{T}{4}) = [R_1^2 + R_2^2]^{\frac{1}{2}} \text{ and}$$

$$X(\tfrac{T}{4}) = -[R_1^2 + R_2^2]^{\frac{1}{2}} \operatorname{Cos} [\pi + \emptyset\ (\tfrac{T}{4})].$$

By use of a trigonometric identity the equation

$$X(\tfrac{T}{4}) = -\ [R_1^2 + R_2^2]^{\frac{1}{2}} \operatorname{Cos} \emptyset\ (\tfrac{T}{4})$$

could be written. Then, if a right triangle is constructed with $R_1$ and $R_2$ as sides and $[R_1^2 + R_2^2]^{\frac{1}{2}}$ as the hypotenuse, $\emptyset(t)$ is the angle between the hypotenuse and $R_1$. Therefore,

$$\operatorname{Cos} \emptyset(t) = \frac{R_1}{[R_1^2 + R_2^2]^{\frac{1}{2}}},$$

which, when substituted into the equation for $X(\frac{T}{4})$, yielded

$$X(\frac{T}{4}) = - R_1 .$$

Then,

$$- R_1 + \Delta = - 5.0$$

could be written.

$$\text{At } t = \frac{T}{2},\ \Delta\omega t = \Delta\omega\frac{T}{2} = \pi, \text{ and } \omega_1 t = 2\pi,$$

then,

$$\emptyset(\frac{T}{2}) = 0,\ R(\frac{T}{2}) = R_1 - R_2, \text{ and } X(\frac{T}{2}) = R_1 - R_2 .$$

Then,

$$R_1 - R_2 + \Delta = 5.$$

Thus, three equations

$$R_1 + R_2 + \Delta = 10$$

$$- R_1 + \Delta = - 5$$

$$R_1 - R_2 + \Delta = 5$$

were available for solution to obtain $R_1$, $R_2$, and $\Delta$. Simultaneous solution of the equations gave values of $R_1 = 6.25$, $R_2 = 2.50$, and $\Delta = 1.25$. During the solution for these values it was noted that if $R_1 = 6.50$, $R_2 = 2.50$, and $\Delta = 1.0$ were substituted in the equations the only change in the output function would be the maximum amplitude of the negative cycle, such that

$$X(0) + \Delta = 10$$

$$X(\tfrac{T}{4}) + \Delta = -\ 5.5$$

and

$$X(\tfrac{T}{2}) + \Delta = 5.$$

Because such a change would not alter the original function's high and low positive amplitudes, which were of prime concern, and because the negative amplitude value was arbitrarily chosen as - 5.0 initially, the latter values were chosen for convenience in setting the initial conditions on the analog computer.

Having established the amplitudes required to generate the desired function, the frequencies $\omega_1$ and $\omega_2$ were considered next. The requirement to keep the periodic output function rate low to remain within system and recorder limitations led to the selection of $\omega_1 = \pi/5$ rad/s. Having assumed $\omega_1 = 2\Delta\omega$, $\Delta\omega = \pi/10$ rad/s and $\omega_2 = \pi/10$ rad/s followed. This established the beat frequency, f, at f = 0.05 c/s and $\dagger$ = 20 s/c. Thus, the period for one local oscillation, from high peak amplitude to the next corresponding low peak amplitude, was $\dagger_1$ = 10 s/c.

The input functions thus obtained were

$$X_1(t) = 6.5 \text{ Cos } (\Delta\pi/5t)$$

and

$$X_2(t) = 2.5 \text{ Cos } (\pi/10t).$$

To produce these functions, differential equations for analog solution were programmed as follows:

$$\ddot{X}_1(t) = -\ 0.3948\ X_1(t)$$

and

$$\ddot{X}_2(t) = -\ 0.09870\ X_2(t).$$

The two input functions were summed with $\Delta$ = 1.0 VDC at the final stage, prior to input of the resulting function to the controller of the MTS system, to provide alternating, maximum positive amplitude peak output voltages of + 10.0 VDC and + 5.0 VDC.

To prevent compressive yield in the specimen due to the - 5.0 VDC output on each cycle of the function, the reference voltage, or local zero, of the system was set such that, under strain control, the negative voltage output caused the specimen to be placed in a state of zero strain. Maximum strain was set to 7.0 VDC output of the 10.0 VDC available. This corresponded to 10737 $\mu$ in/in strain in the specimen on the high amplitude cycle and 6168 $\mu$ in/in strain on the low amplitude cycle.

As in the cyclic stress-strain curve test, initial conditions were set to zero at the start of this test and then brought up to the specified values manually with the system under the control of the analog computer. With all initial conditions set in, the specimen was in a maximum strain condition. At this point the analog computer was activated and allowed to cycle the specimen 140 times. Outputs of strain and load voltages were recorded on both the X-Y recorder and the dual trace strip chart recorder. As in the previous tests, load data were interpreted directly as stress.

2. Test Results

The output voltages of stress and strain plotted by the X-Y recorder provided data points from which a monotonic stress-strain curve was constructed (Fig. 12 and Table 6). The modulus of elasticity calculated for the curve was $E = 10.19 \times 10^6$ lbf/in$^2$. The yield stress was 78,000 lbf/in$^2$. Stress and strain data from this curve were used to construct a stress x strain ($\sigma\varepsilon$) vs. stress curve for comparison with those of other uniaxial specimen tests and for later use in the plate tests (Fig. 16 and Table 6).

The dual trace recorder provided stress output voltages from which maximum stress per cycle could be computed (Table 7). The stress data were plotted versus cycle number, N, on semilog graph paper (Fig. 17) to graphically represent stress relaxation behavior. The locus of the low amplitude data points as well as the high amplitude data points appeared to form a straight line, indicating equations for both stress relaxation behaviors would be of the form

$$\sigma = \sigma_0 e^{-bN}.$$

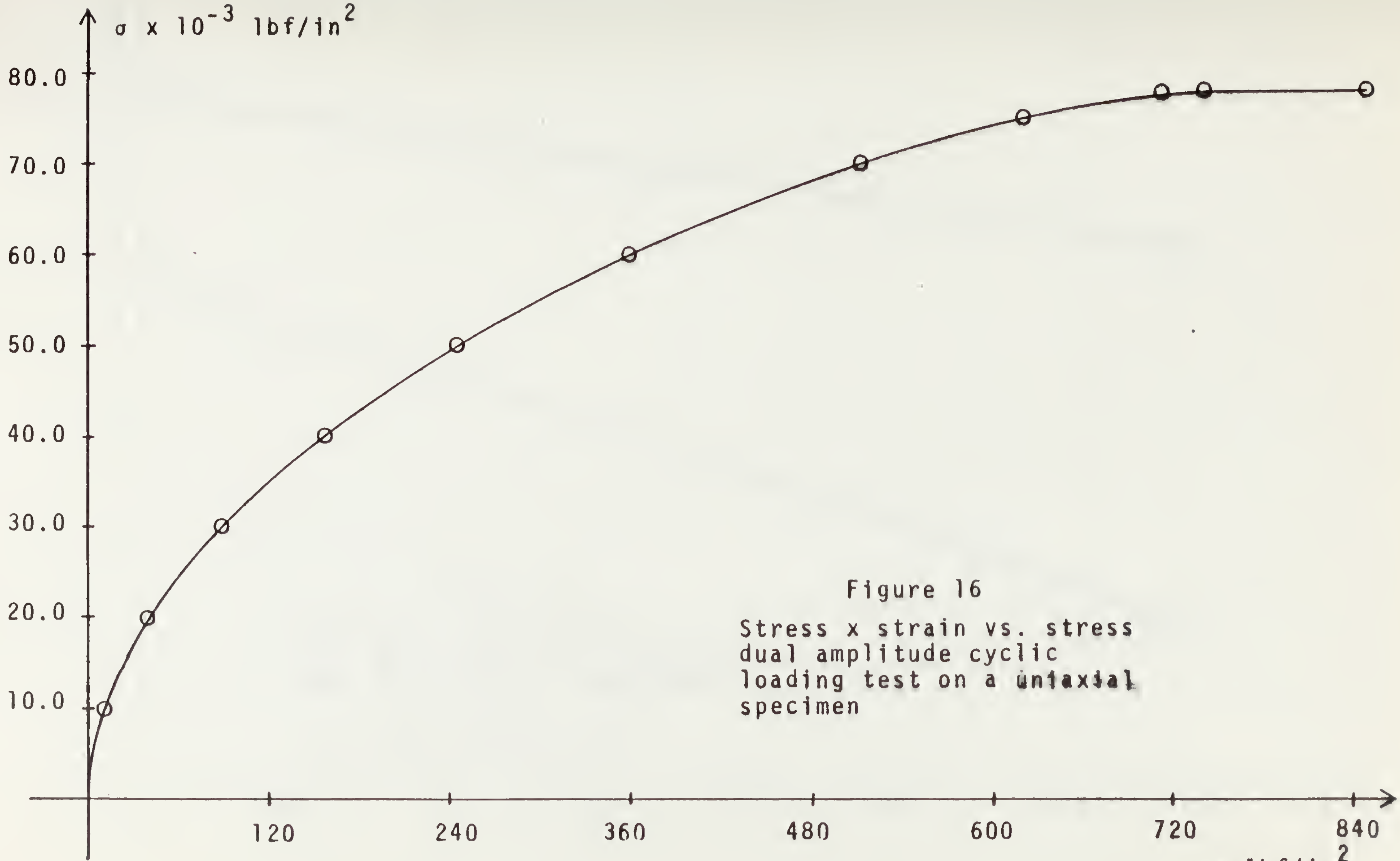


Figure 16

Stress x strain vs. stress dual amplitude cyclic loading test on a uniaxial specimen

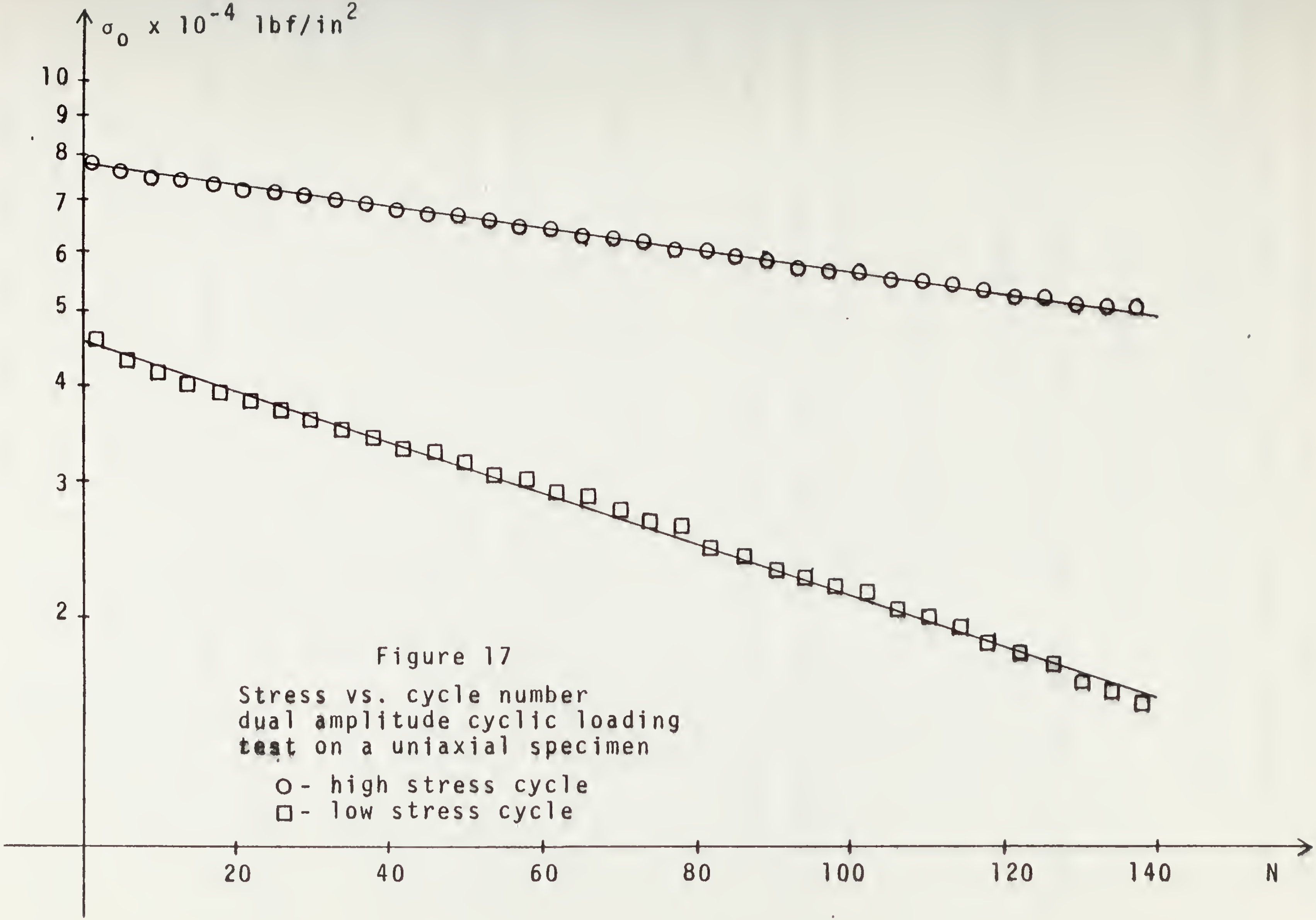


Figure 17

Stress vs. cycle number
dual amplitude cyclic loading
test on a uniaxial specimen

○ - high stress cycle
□ - low stress cycle

A least squares exponential curve fit was applied to 70 high stress points and to 70 low stress points. The resulting equation for the high stress relaxation behavior was

$$\sigma = 76930\ e^{-(3.168 \times 10^{-3})N}$$

with a correlation coefficient of 0.999. For the low stress situation the equation was

$$\sigma = 45600\ e^{-(7.572 \times 10^{-3})N}$$

with a correlation coefficient of 0.998. Thus the stress relaxation behavior in a dual amplitude loading program was defined in terms of the number of cycles and an initial stress value.

## E. DISCUSSION OF TEST RESULTS

The main objective of the uniaxial specimen tests was to provide consistent data in the form of monotonic and cyclic stress-strain curves, stress x strain ($\sigma\varepsilon$) vs. stress curves, and stress relaxation behavior for 7075 T-6 aluminum. These data were to be used for comparison and analysis of data taken in tests on plates with central holes.

Three monotonic stress-strain curves were obtained based on three separate tests of uniaxial specimens (Figs. 10 and 12). The moduli of elasticity for the three tests were $E = 10.67 \times 10^6$ lbf/in$^2$ for the curve obtained from the cyclic stress-strain curve test, $E = 10.0 \times 10^6$ lbf/in$^2$ from the single amplitude test, and $E = 10.19 \times 10^6$ lbf/in$^2$ from the dual amplitude test. These values are within a maximum of 6.28 percent of each other. The average value of the three moduli of elasticity, $E = 10.29 \times 10^6$ lbf/in$^2$, is almost identical to the published value for 7075 T-6 aluminum, $E = 10.3 \times 10^6$ lbf/in$^2$. The maximum deviation of individual values obtained from the published value is 3.48 percent. Comparison of curve shape indicates excellent correlation up to stresses of approximately 60,000 lbf/in$^2$, after which some deviation of the curves from each other is evident. The monotonic stress-strain curve from the dual amplitude loading test tends to decrease slope more rapidly above 60,000 lbf/in$^2$ and reaches a limit at a stress level of 78,000 lbf/in$^2$. The single amplitude loading test curve and the monotonic stress-strain curve from the cyclic stress-strain curve test decrease slope at approximately the same rate but have stress limits of 80,000 lbf/in$^2$ and 78,000 lbf/in$^2$, respectively.

Because only one cyclic stress-strain curve was developed (Fig. 9), a comparison for consistency could not be made. However, the modulus of elasticity obtained was $E = 10.18 \times 10^6$ lbf/in$^2$, which is consistent with the monotonic stress-strain curve values obtained, as it should be. The slope decreases more rapidly in comparison to the monotonic curves and remains below it indicating the material cyclically softens. This is not compatible with the results found by Landgraf _et al_ (Ref. 6), which indicates 7075 T-6 aluminum hardens under cyclic loading. The cyclic stress-strain curve obtained in this test was the only one available for use and therefore would be used, if necessary, while the differences in results were noted.

The stress x strain ($\sigma\varepsilon$) vs. stress curves (Figs. 11, 13 and 16) obtained from the three stress-strain curves conform favorably. The differences noted are due to the differences found in the stress and strain data and perpetuated in the mathematics used to construct them. As in the monotonic cases, and for the same reasons, the stress x strain ($\sigma\varepsilon$) vs. stress curves are acceptable for use in the plate tests.

A comparison of the stress relaxation behavior in the single and dual amplitude loading tests can be made by consideration of the equations obtained previously describing this behavior. The equations take the form

$$\sigma = \sigma_o e^{-bN} .$$

Of particular interest are the initial stress, $\sigma_o$, and the stress relaxation rate parameter, b. The single amplitude loading test produced $\sigma_o$ = 73160 lbf/in$^2$ and b = 3.177 x $10^{-3}$, while the high stress relaxation behavior of the dual amplitude loading test produced $\sigma_o$ = 76930 lbf/in$^2$ and b = 3.168 x $10^{-3}$. The initial stresses differ by 4.8 percent, possibly due to the mathematics of the curve fit routine, and the relaxation rate parameters differ by 0.28 percent. This correlation seems to be quite good and would indicate the type of loading history does not appreciably affect the relaxation behavior of the material when it is loaded repeatedly beyond the proportional limit. However, because only one low cycle stress was applied between the high stress cycles, further tests with considerably different loading histories would be desirable before concluding this to be the general behavior of the high stress relaxation.

The low stress level relaxation behavior provided an initial stress of $\sigma_o$ = 45,600 lbf/in$^2$ and a relaxation rate parameter of b = 7.572 x $10^{-3}$. The relaxation rate parameter, b, is significantly higher for the low stress portion of loading than for the high stress portion and indicates a possible association between initial stress and relaxation rate. Because only one dual amplitude loading test was performed at one low stress value, further dual amplitude tests with various low stress values are warranted prior to generalizing this association.

From the three uniaxial specimen tests a sound data base was obtained for use in the analysis of the plate tests.

# III. LOCAL STRESS-STRAIN BEHAVIOR

## A. INTRODUCTION

General stress analysis indicates aircraft structures are in a state of uniaxial stress in most cases but contain numerous local stress concentrations. The Department of Aeronautics, Naval Postgraduate School, Monterey, California, has developed a strain monitoring system which provides data on the nominal stresses experienced by aircraft structures (Refs. 15, 16, and 17) which could be applied to obtain local strain at the stress concentrations. Practicality prevents such monitoring of the numerous local stress concentrations, while fatigue life estimation requires knowledge of local stresses. A means of relating the readily available nominal strains and the local stresses is required for practical fatigue life estimation in aircraft structures.

In order to obtain relationships between local stress and nominal strain for real structures possessing geometric effects, plates with central holes to model those effects were subjected to single and dual amplitude cyclic loading tests. The local strain and nominal stress and strain data obtained in these tests were used to calculate local stress at the hole in order to determine the suitability of a method, based on a theory proposed by Neuber (Ref. 1), for computing local stress on the basis of knowledge of nominal strain and the material properties alone. In addition, these tests were expected to show the interactions, if any, between geometric configuration and loading history on the local stress relaxation behavior as compared with the behavior found in uniaxial specimen tests.

## B. CALCULATION OF LOCAL STRESS ON INITIAL LOADING

Because stress in a plate can not be measured directly, other analytical methods must be used to determine the stress at points (A) and (B) in Figure 2.

One such method involves a proposal by Neuber (Ref. 1), derived from a study of prismatical bodies. Neuber concluded that the geometric mean of stress concentration factor, $K_\sigma$, and strain concentration factor, $K_\varepsilon$, is equal to the elastic stress concentration factor, $K_t$. In equation form:

$$K_t^2 = K_\sigma K_\varepsilon$$

where $K_\sigma$ is local stress, $\sigma$, divided by nominal stress, S, and $K_\varepsilon$ is local strain, $\varepsilon$, divided by nominal strain, e.

This equation can be further reduced to

$$K_t^2 = \frac{\sigma\varepsilon}{Se}$$

or, by assuming nominal stress remains in the linear region and S = E e,

$$K_t^2 = \frac{\sigma\varepsilon}{E\ e^2} \quad .$$

In this form the stress concentration factor is calculated on the basis of the measured local and nominal strain and calculated values of local stress, and the modulus of elasticity from the appropriate stress-strain curve. The stress concentration factor thus calculated should be indicative of, in this case, all plates with central holes of the same proportionate dimensions and of the same material. With the stress concentration factor thus calculated

$$\sigma\varepsilon = E\ e^2 K_t^2$$

can be written. Therefore, with the stress concentration for a particular configuration known, the modulus of elasticity for the material, and a stress x strain ($\sigma\varepsilon$) vs. stress curve as calculated in the uniaxial specimen tests, the only requirement is knowledge of the nominal strain, an easily obtained quantity from a practical standpoint. In further discussions this method will be known as the Neuber method.

## C. EVALUATION OF STRAIN GAGE PLACEMENT

Prior to conducting tests on the 7075 T-6 aluminum plate specimens, a determination of the validity of strain data obtained from strain gages positioned at points (A) and (B) in Figure 18 was made. A plate specimen of 2024 aluminum was prepared with strain gages mounted at these points. Strain gage (A) provided maximum strain on the notch or hole edge and strain gage (B) provided an average of strain across the area it spanned.

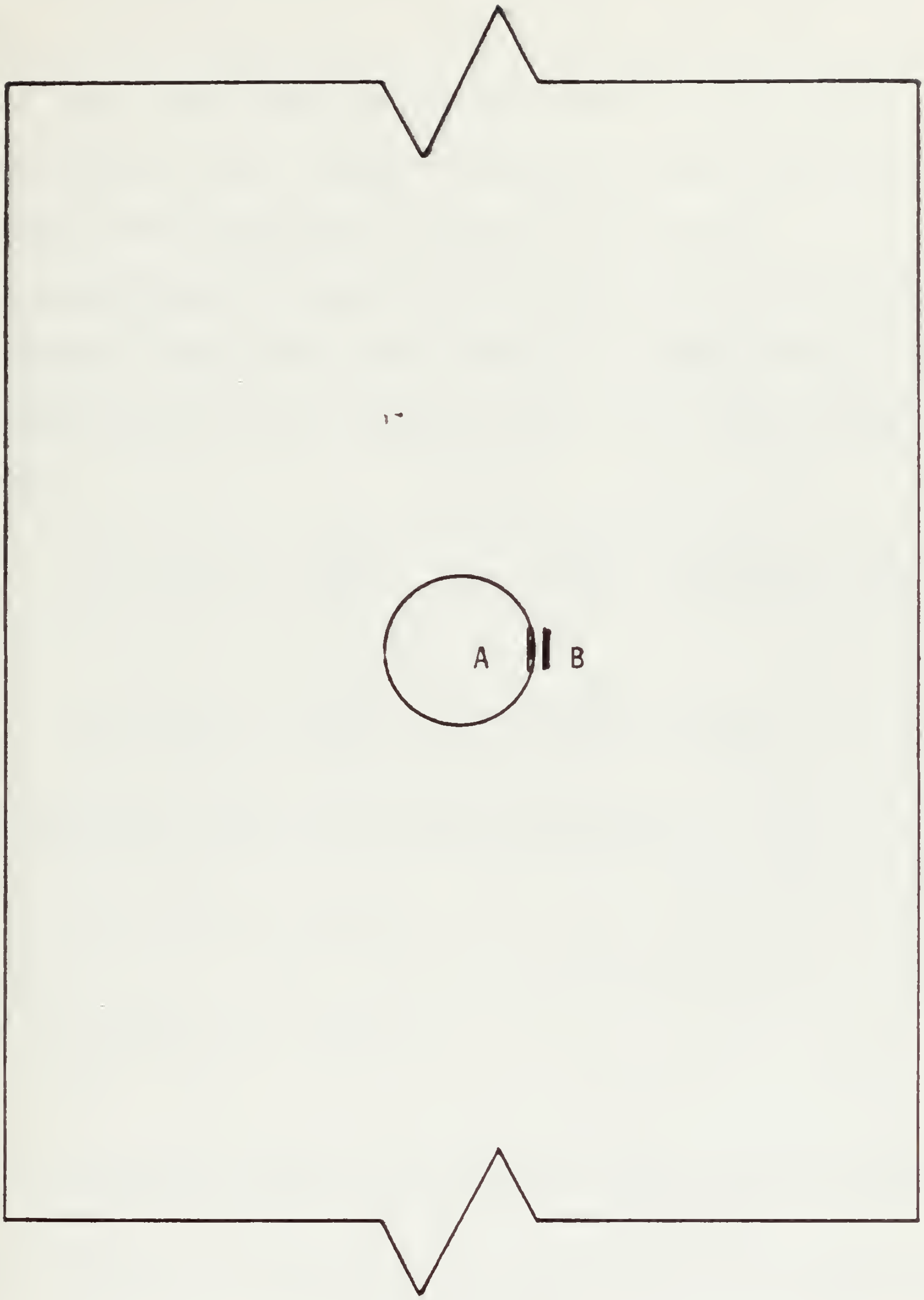


Figure 18

Center section of plate specimen
with locations of strain gages
used in strain gage placement test.

The plate was loaded into tension in load steps of 2000 lbs. up to 18,000 lbs. Strain data from each strain gage and load were recorded at each step. This step was repeated three times. Nominal stress on the plate was calculated from load data (Tables 8, 9, and 10).

To calculate the theoretical values of strain at the two locations the stress and strain solutions for an infinite plate with a hole in the center were used. The stress equations

$$\sigma_r = \frac{\sigma}{2}\left[\left(1 - \frac{a^2}{r^2}\right) - \left(1 - 4\frac{a^2}{r^2} + 3\frac{a^4}{r^4}\right)\text{Cos } 2\theta\right]$$

and

$$\sigma_\theta = \frac{\sigma}{2}\left[\left(1 + \frac{a^2}{r^2}\right) + \left(1 + 3\frac{a^4}{r^4}\right)\text{Cos } 2\theta\right]$$

were substituted into the strain equations

$$\varepsilon_r = \frac{1}{E}\left[\sigma_r - \nu\sigma_\theta\right]$$

and

$$\varepsilon_\theta = \frac{1}{E}\left[\sigma_\theta - \nu\sigma_r{}^1\right]$$

where

$\sigma_r$ - stress in the region of the hole in a direction perpendicular to that of the loading

$\sigma_\theta$ - stress in the region of the hole in a direction parallel to that of loading

$\sigma$ - nominal stress on the plate

a - radius of the hole

r - distance from the center of the hole to the point of interest

$\theta$ - angle measured from a horizontal bisector of the hole to the point of interest

$\varepsilon_r$ - strain in the region of the hole perpendicular to the direction of loading

$\varepsilon_\theta$ - strain in the region of the hole parallel to the direction of loading

E - Young's modulus of elasticity

$\nu$ - Poisson's Ratio

Of interest was the area along a line bisecting the hole and perpendicular to the loading direction where $\theta = 0$. The strain equation for this region is

$$\varepsilon_\theta = \frac{\sigma}{2}\frac{1}{E}\left[\frac{a^2}{r^2}(1 - 3\nu) + 3\frac{a^4}{r^4}(1 + \nu) + 2\right].$$

For maximum strain $r = a$, or

$$\varepsilon_{\theta_{max}} = 3\frac{\sigma}{E}\ .$$

For the average strain value, that which the gage measures, the strain was integrated over the radial distance from the inner edge of the strain gage to the outer edge and divided by that distance such that

$$\varepsilon_{\theta_{avg}} = \frac{\sigma}{2E(r_2 - r_1)}\int_{r_1}^{r_2}\left[\frac{a^2}{r^2}(1 - 3\nu) + 3\frac{a^4}{r^4}(1 + \nu) + 2\right]dr$$

or

$$\varepsilon_{\theta_{avg}} = \frac{\sigma}{2E(r_2 - r_1)}\left[a^2(3\nu - 1)\left(\frac{r_1 - r_2}{r_1\ r_2}\right) - a^4(1 + \nu)x\right.$$

$$\left.\left(\frac{r_1^3 - r_2^3}{r_1^3\ r_2^3}\right) + 2(r_2 - r_1)\right]$$

The physical placement of the strain gage on the sheet provided the following data:

$$r_1 = 1.023 \text{ in.}, \; r_2 = 1.062 \text{ in.}, \; a = 1.0 \text{ in.}$$

From data for 2024 aluminum $E = 10.6 \times 10^6$ lbf/in$^2$ and $\nu = 0.33$. Substitution of the above into the equations for maximum and average strain provides

$$\varepsilon_{\theta_{max}} = 0.28302 \, \sigma \; \mu\text{in/in}$$

and

$$\varepsilon_{\theta_{avg}} = 0.25401 \, \sigma \; \mu\text{in/in.}$$

Substitution of nominal stress values obtained in the actual tests into the above equations provided values of maximum and average strain to compare with the measured values of strain (Table 8, 9, and 10).

Of the three test series run, the second is considered the most accurate and reliable, with runs 3 and 1 next in accuracy in that order. In the comparison of theoretical and actual average strains in all three runs the measured strain exhibited greater deviation from the theoretical strain at the lower stress level of approximately 2000 lbf/in$^2$ (Table 11). In the two most reliable runs this deviation was approximately 2.5 percent. As stress increased, deviation decreased until stresses of around 9000 lbf/in$^2$ were reached. At this point the strain deviation again began to increase, reaching a maximum of less than 1.5 percent at over 17,000 $lbf/in^2$. Also noted was the tendency of measured strain to cycle about the theoretical strain. At low stresses the theoretical exceeded the measured strain. With increasing stress, measured strain increased, equaled theoretical at approximately 9000 $lbf/in^2$, then exceeded theoretical strain up to the maximum stress reached in each test.

The deviation of the measured strain from the theoretical strain was considered small and well within the accuracy of the entire material testing system, including the strain gages themselves, and therefore was considered adequate for the purpose of future tests.

The results of the maximum strain data comparisons were somewhat less desirable. In all runs, measured strain was greater than theoretical strain, and deviation continued to increase up to a maximum of approximately 5.25 percent. This greater deviation in the maximum strain tests than in the average strain tests was considered to be due to the placement of the strain gage on the curved, inside edge of the hole. Initial curvature of the strain gage was unavoidable due to its location, and the extension experienced by it was not entirely in the plane of the strain gage, as required for accurate strain reproduction. The trend of the deviation indicates that at higher stress levels than the levels encountered here, the deviation would be accordingly higher.

The increasing trend of the actual measured strain's deviation from the theoretical, and the close correlation of the average strain measured in the test with that calculated by theory, led to the decision to instrument plate specimens for this investigation for average strain data output rather than for maximum strain reproduction.

## D. CYCLIC LOADING TESTS

The plate specimens were tested in the MTS Corporation closed-loop, servo-controlled testing system used in the uniaxial specimen tests. The same function generator and analog computer were used in the plate tests. The functions used for strain control of the system were identical to those used in the uniaxial specimen single and dual amplitude cyclic loading tests. Output voltages representing load and strain were recorded on the Hewlett-Packard X-Y recorder and a Varian Corporation eight channel strip recorder. The X-Y recorder was used to record voltage outputs of nominal loads and local strain for test monitoring purposes only, and was not required for actual data analysis. The Varian recorder was calibrated to record one voltage input across two channels, thus doubling the resolution. This was done for six channels to provide more accurate recording of local strain data on two of the doubled channels, and nominal strain data on the third. Nominal load output voltages were recorded on one single channel strip.

Two plates with central holes (Figs. 1 and 2) were constructed from the same master sheet of 7075 T-6 aluminum. Care was taken to ensure that no stress raisers, such as scratches or notches, other than the hole itself were introduced. The cross-sectional area of each plate was 1.080 $in^2$. Strain gages were mounted at points (A), (B), and (C), as depicted in Figure 2. Strain gages at points (A) and (B) provided local strain at the point of highest stress in the plate and the strain gage at (C) provided nominal strain in the plate. The two local strain gages, one on either side of the hole at the point of maximum stress, were utilized to ensure that the data recorded were representative of a plate in uniaxial tension and not subject to undesirable loading such as shear introduced by improper clamping.

Prior to each test all strain gages on the specimen were zeroed and calibrated with the specimen free at one end. After attachment of the free end to the system's load cell, the strain gages were zeroed under load control and the load output voltage adjusted to zero to ensure zero load at zero strain. Recorders were calibrated prior to each test with a known voltage input to ensure accurate reproduction of voltage inputs from the test system.

1. <u>Single Amplitude Cyclic Loading Test</u>

Nominal stress and strain and local strain data from a plate with a hole under single amplitude cyclic loading were required for evaluation of the accuracy of Neuber's method, and for construction of monotonic local stress vs.

nominal strain curves for comparison with the curves obtained from a plate under a different loading history, and for later use in stress relaxation behavior studies.

The haversine function produced by the function generator in the MTS system and used in the uniaxial specimen test was employed in this test on the plate. The system was driven under strain control to a maximum amplitude of 7.3 VDC of the 10.0 VDC available. This voltage corresponded to 11167 μin/in strain in the specimen, according to strain gage (1) used as a control reference by the system; and to 11696 μin/in strain in strain gage (2), mounted opposite the reference strain gage and used for data consistency comparisons. The specimen was cycled 91 times to the maximum stress value.

The MTS system supplied one local strain and the nominal load voltage outputs to an X-Y recorder for test monitoring purposes. Voltage outputs for local strain, nominal strain, and nominal load were supplied to the Varian recorder for later data reduction.

2. <u>Single Amplitude Cyclic Loading Test Results</u>

The Varian recorder provided voltage representations of nominal load, nominal strain, and local strain from two gages. Nominal stress was obtained by dividing the nominal load by the cross-sectional area of the plate at the clamped ends. Nominal stress and local and nominal strains were computed for numerous points on the initial loading cycle Table 12).

A comparison of stress concentration factors calculated using Neuber's equation,

$$K_t^2 = \frac{\sigma \varepsilon}{Se} ,$$

with the theoretical stress concentration factor calculated for the plate with a central hole was made to evaluate Neuber's theory.

The theoretical stress concentration factor was calculated according to

$$K_t = \frac{3W}{W + D}$$

(Ref. 18), where W was defined as the width of the plate and D was defined as the diameter of the hole. From Figure 2, W = 12.0 in. and D = 2.0 in., therefore the theoretical stress concentration factor was calculated to be $K_t = 2.57$.

The stress and strain data taken from the initial loading cycle were used to calculate stress concentration factors for the plate to evaluate Neuber's equation. Two values of local stress due to material variations were obtained with each strain value from the local strain gages by entering the monotonic stress-strain curves constructed from: (1) the cyclic stress-strain test, and (2) the single amplitude cyclic loading test on the uniaxial specimens. With local and nominal stresses and strains known, the stress concentration factors could be calculated according to

$$K_t^2 = \frac{\sigma \varepsilon}{Se} .$$

This was done at seventeen points of local strain for both local strain gages and the two monotonic stress-strain curves mentioned (Table 13). The results were averaged such that $K_t$ = 2.59 for strain gage (1) and $K_t$ = 2.67 for strain gage (2) from the single amplitude cyclic loading test monotonic stress-strain curve, and $K_t$ = 2.61 for strain gage (1) and $K_t$ = 2.68 for strain gage (2) from the monotonic stress-strain curve constructed in the cyclic stress-strain curve test on the uniaxial specimen.

The maximum deviation of the stress concentration factors calculated from Neuber's theory was 4.10 percent. The average of the four values was $K_t$ = 2.64, within 2.56 percent of the theoretical value. The close correlation of the experimental stress concentration factors with the theoretical value indicates that Neuber's relationship is valid as a basis for calculating local stress.

Noting the apparent validity of Neuber's equation, data points were calculated for construction of monotonic local stress vs. nominal strain curves, based on Neuber's method, from the viewpoint of the analyst who has knowledge of modulus of elasticity, stress concentration factor, and nominal strain only ($\varepsilon_l$ is unknown). The stress concentration factors calculated from both local strain gages, the several moduli of elasticity, and nominal stress were used to obtain four local stress vs. nominal strain relationships for comparison. Neuber's method was applied, as previously outlined, to obtain local stress which was then plotted against the corresponding local strain (Fig. 19 and Table 14).

To provide a basis for comparison of the local stress vs. nominal strain from the two sets of data, two curves were constructed: one based upon the average value of the stress concentration factors, $K_t = 2.64$, and the average value of the three moduli of elasticity, $E = 10.29 \times 10^6$ lbf/in$^2$, and the other based upon the theoretical value of $K_t = 2.57$ and the published value of modulus of elasticity, $E = 10.3 \times 10^6$ lbf/in$^2$. The average curve, (A), in Figure 19, although slightly above, correlates well with the theoretical curve, (B). The scatter of all data points about these curves is quite low.

The maximum variation between the two sets of data points was 38.46 percent for strain gage (1) and 37.93 percent for strain gage (2), both at the lowest nominal strain (Table 14). The average variation for all points was 6.03 percent and 6.14 percent respectively, but omission of the initial data point with maximum variation on each test reduced this average to 4.00 percent and 4.15 percent, respectively.

The low scatter of the data points about the theoretical and average curves, as well as the relatively low average variation between the points based on the two sets of data, indicated either the average or the published material properties may be used to construct local stress vs. nominal strain curves for practical use.

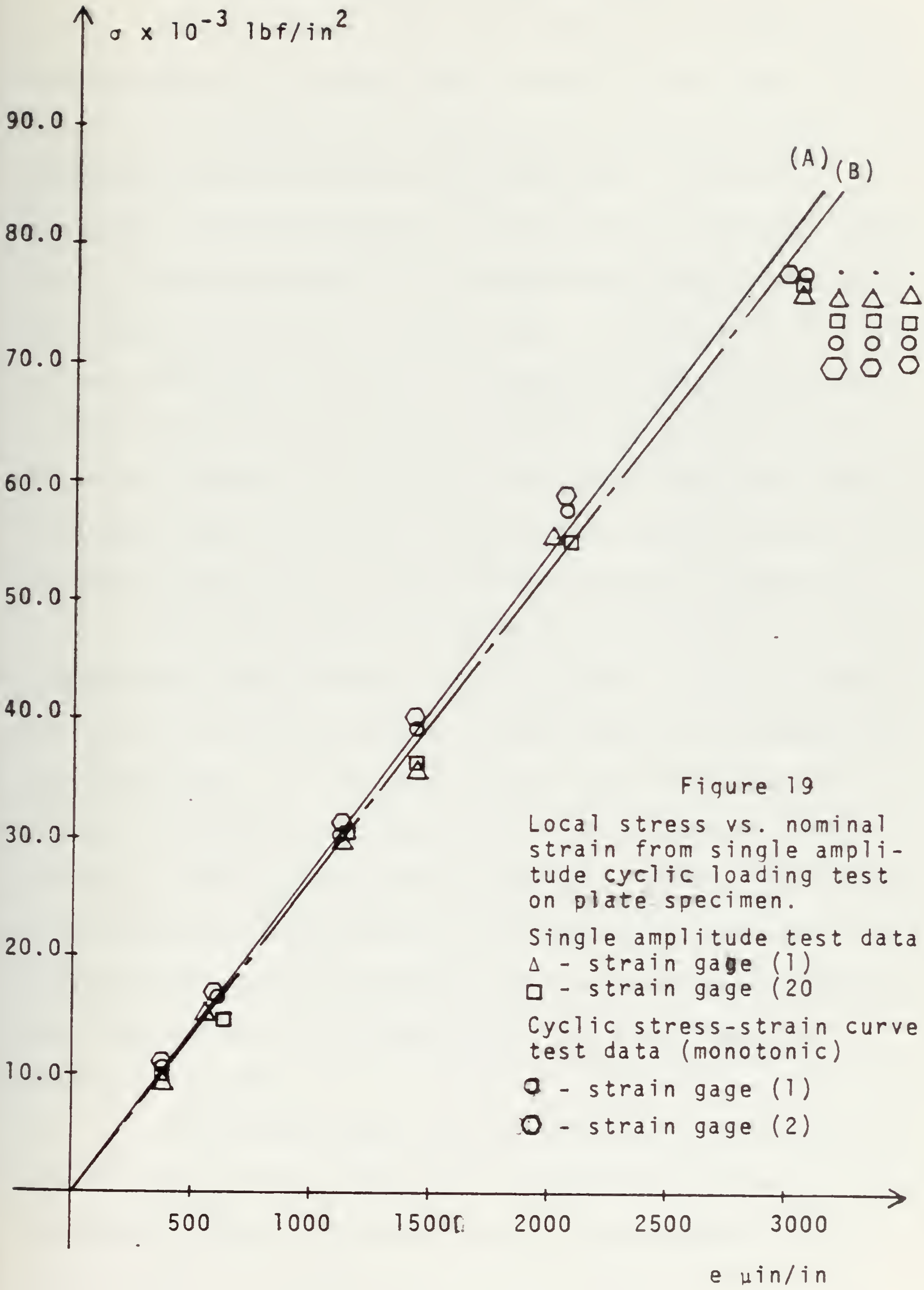


Figure 19

Local stress vs. nominal strain from single amplitude cyclic loading test on plate specimen.

Single amplitude test data
△ - strain gage (1)
□ - strain gage (20

Cyclic stress-strain curve test data (monotonic)
○ - strain gage (1)
⬡ - strain gage (2)

### 3. Dual Amplitude Cyclic Loading Test

Having established a data base for a simple cyclic loading history of a plate with a hole in the single amplitude cyclic loading test, stress and strain data from a different loading situation were desired. In order to establish a sound data base on more realistic loading situations, a plate specimen with a central hole was subjected to a dual amplitude cyclic loading test. This simple step toward the more realistic situation of random cycling was expected to provide a second evaluation of Neuber's equation, local stress vs. nominal strain curves for comparison with those of the single amplitude cyclic loading test, and additional data for use in the study of local stress relaxation behavior.

The dual amplitude function provided by the analog computer for strain control of the system in the dual amplitude cyclic loading uniaxial specimen test was repeated in this plate test. As in the previous test the system was driven to a maximum amplitude of 7.0 VDC of the 10.0 VDC available. This voltage corresponded to 10708 $\mu$in/in strain in the material, according to strain gage (1) used as a reference to control the system; and to 11178 $\mu$in/in strain according to local strain gage (2) used as a comparison against the first.

In this test the initial conditions to the analog computer were set to zero prior to the start of the test run. The plate specimen was brought up to a maximum amplitude strain in the first cycle by manual input of the initial conditions to full value. At the maximum amplitude the analog computer was activated and allowed to run for 114 cycles before test termination.

The system supplied one local strain and the nominal load voltage outputs to an X-Y recorder for monitoring purposes during the test run. Voltage outputs representing both local strains, the nominal load, and the nominal strain were supplied to the Varian recorder for reproduction.

4. Dual Amplitude Cyclic Loading Test Results

As in the single amplitude cyclic loading plate test, nominal load, nominal strain, and two local strains were recorded. Nominal stress was calculated by dividing nominal load by the cross-sectional area of the plate at the clamped end. A second evaluation of Neuber's equation was made by comparison of stress concentration factors calculated from experimental data with the theoretical value for the plate configuration, $K_t = 2.57$.

Stress concentration factors for the plate were first calculated using stress and strain data from the initial loading cycle (Table 15). Local strains were used to obtain local stresses directly from the monotonic stress-strain curves developed from (1) the cyclic stress-strain test and (2) the dual amplitude cyclic loading test on uniaxial specimens. The known values of local stress and strain and nominal stress and nominal strain for fifteen points on the initial loading cycle were used in

$$K_t^2 = \frac{\sigma \varepsilon}{Se}$$

to obtain stress concentration factors for averaging (Table 16). The stress concentration factors thus obtained were $K_t = 2.62$ for strain gage (1) and $K_t = 2.67$ for strain gage (2) based on monotonic stress-strain data from the cyclic stress-strain test, and $K_t = 2.61$ for strain gage (1) and $K_t = 2.65$ for strain gage (2) for data based on the single amplitude cyclic loading test.

The maximum variation of the stress concentration factors calculated by Neuber's theory was 3.75 percent. The average of the four values was $K_t = 2.64$, within 2.56 percent of the theoretical and identical to the average $K_t$ found in the single amplitude cyclic loading test. The conclusion made from the data of the single amplitude cyclic loading test, that Neuber's theory is valid as a basis for calculating local stress, was reinforced by the close correlation of experimental stress concentration factors with the theoretical in this test.

Local stress vs. nominal strain curves were constructed for comparison with those obtained in the previous test. The stress concentration factors were calculated using data from both local strain gages and two moduli of elasticity: one from the dual amplitude cyclic test, and one from the monotonic stress-strain curve of the cyclic stress-strain test, and nominal strain data from this test. The Neuber method was applied to these data to obtain local stress which was plotted against the corresponding nominal strain (Fig. 20

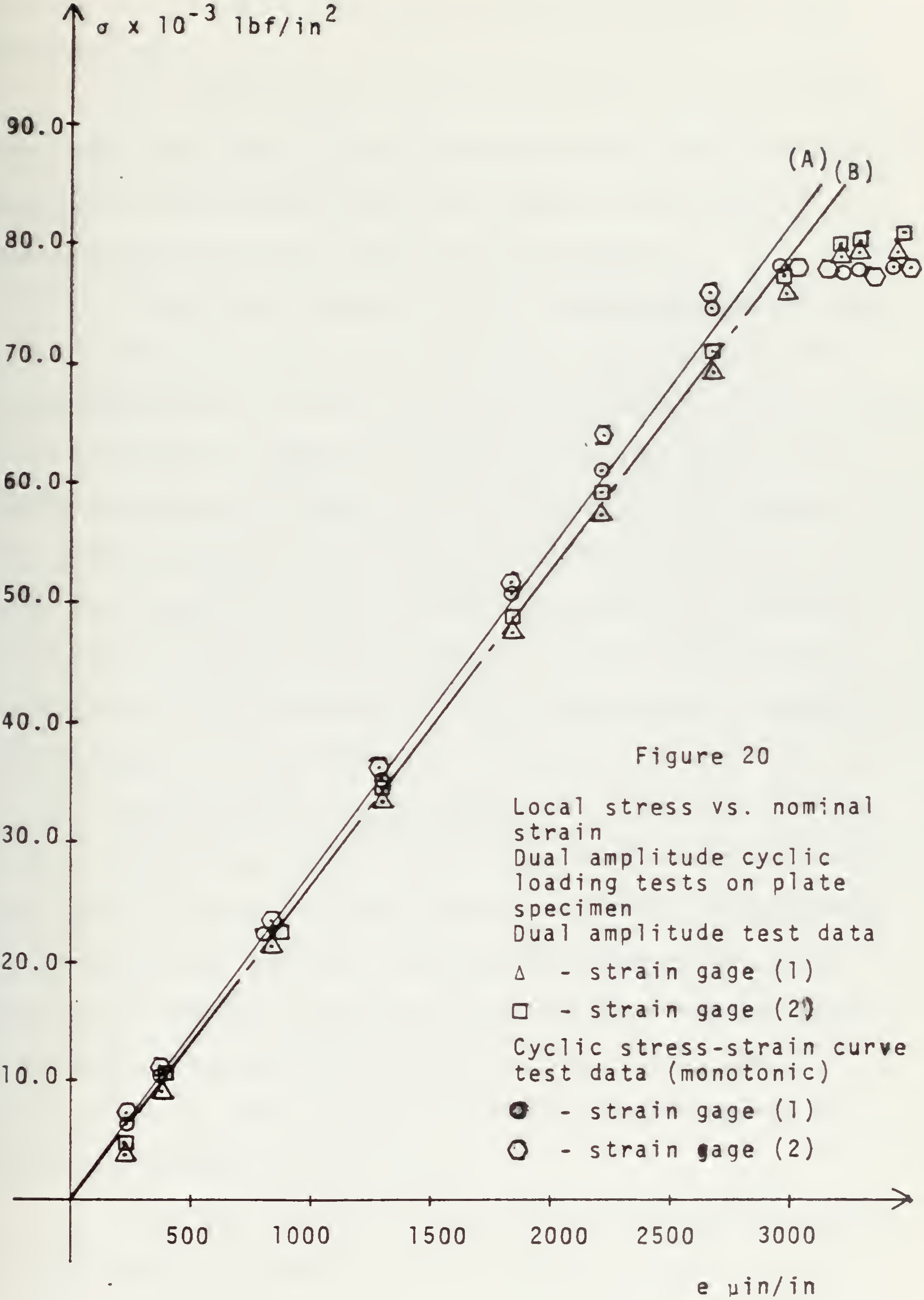


Figure 20

Local stress vs. nominal strain
Dual amplitude cyclic loading tests on plate specimen
Dual amplitude test data
△ - strain gage (1)
□ - strain gage (2)
Cyclic stress-strain curve test data (monotonic)
● - strain gage (1)
⬡ - strain gage (2)

and Table 17) to give four local stress vs. nominal strain relationships.

The theoretical and average curves constructed in the single amplitude cyclic loading test were applied to the data points of this test also and, again, a low scatter of points about the curves (A) and (B) was noted.

A variation between curves constructed on one data base with those of the other data base was noted. Maximum variation between the two sets of data points was 11.94 percent for strain gage (1) and 13.24 percent for strain gage (2), with the average variation for all points of 6.32 percent and 7.71 percent, respectively (Table 17). Due to the low scatter about the theoretical and average curves and the relatively low variation between the two sets, a curve constructed on the basis of either averaged or published material properties would be good for practical use.

### 5. Discussion of Test Results

The primary objective of the single and dual amplitude cyclic loading tests on plates with central holes were: (1) evaluation of Neuber's relationship, for validity as a basis for a method to calculate local stress from knowledge of nominal strain and material properties alone; and (2) construction of local stress vs. nominal strain curves for comparison between the two tests.

The calculation of the stress concentration factors for evaluation of Neuber's relationship, and the construction of the monotonic local stress vs. nominal strain curves, were carried out using two sets of stress and strain data for each plate test, the first set being the uniaxial specimen test data corresponding to the particular loading test being applied to the plate, and the second set being based on the monotonic stress-strain curve obtained in the cyclic stress-strain curve test on the uniaxial specimen. The latter data base provided a commonality to the calculations made for the two different plate tests. As would be expected from the close correlation of the stress-strain curves obtained in the uniaxial specimen single and dual amplitude cyclic loading tests, the stress concentration factor calculated in one plate test was approximately equal to that calculated in the other test for a particular strain gage. As further evidence of the consistency of the basic data used in the two tests, the stress concentration factors calculated in each test, based on the monotonic stress-strain curve from the cyclic stress-strain curve test on a uniaxial specimen, were approximately equal from test to test for a particular strain gage. Finally, the maximum variation between any two of the eight stress concentration factors calculated for both tests was 3.63 percent, indicating that the stress concentration factor was essentially consistent from test to test. The average value of all eight factors was $K_t = 2.64$, within 2.56 percent of the theoretical value.

The high correlation of the stress concentration factors obtained experimentally from both tests with the theoretical value calculated from plate dimensions indicated that Neuber's theory is a valid basis for computation of local stress using only nominal strain.

The low scatter of the local stress vs. nominal strain data points about the theoretical and average curves in both tests, and the relatively low variation between sets of data points within a test, led to the conclusion that a single local stress vs. nominal strain curve based on either average or theoretical data would provide an accurate, practical relationship for determining monotonic local stress at a stress concentration in a structure with only nominal strain and the readily available material property.

## E. STRESS RELAXATION BEHAVIOR

### 1. Introduction and Theory

The stress relaxation behavior of the plate specimens under two different conditions of loading was required for comparison with the behavior of the uniaxial specimens in order to determine the effects, if any, of geometry on the behavior.

Local strain and nominal stress and strain data were obtained from the unloading portion of the stress-strain curve, and from the point of maximum strain on each cycle of the single and dual amplitude cyclic loading tests on plates. The calculation of local stress from these data was required.

The method used to calculate the local stress for the monotonic local stress vs. nominal strain relationships was expanded upon to obtain local stress for relaxation behavior under cyclic loading conditions, where stress concentration factors could differ from those of the monotonic case. After initial tensile yielding, the stress-strain behavior shifts to the right on the stress-strain curve (Fig. 21), and further cycling is along curve (A-B). This can be thought of as loading from a new origin. Designating quantities which originate from there with a subscript, u, for unloading, and noting that the modulus of elasticity along curve (A-B) is approximately equal to that along curve (O-A), the following quantities are defined:

- $\sigma_m$ - initial maximum local stress in the material
- $\varepsilon_m$ - initial maximum local strain in the material
- $\sigma_u$ - difference in initial maximum local stress and maximum local stress on a given cycle
- $\varepsilon_u$ - difference in initial maximum local strain and maximum local strain in a given cycle
- $S_m$ - initial maximum nominal stress
- $S_u$ - difference in initial maximum nominal stress and maximum nominal stress in a given cycle
- $K_{tu}$ - stress concentration factor associated with curve A-B

From these definitions and Figure 21 several general equations can be set forth:

$$\sigma_u = E\,\varepsilon_u$$

$$\sigma_u = K_{tu} S_u$$

$$\varepsilon_u = \varepsilon_m - \varepsilon$$

$$\sigma_u = \sigma_m - \sigma$$

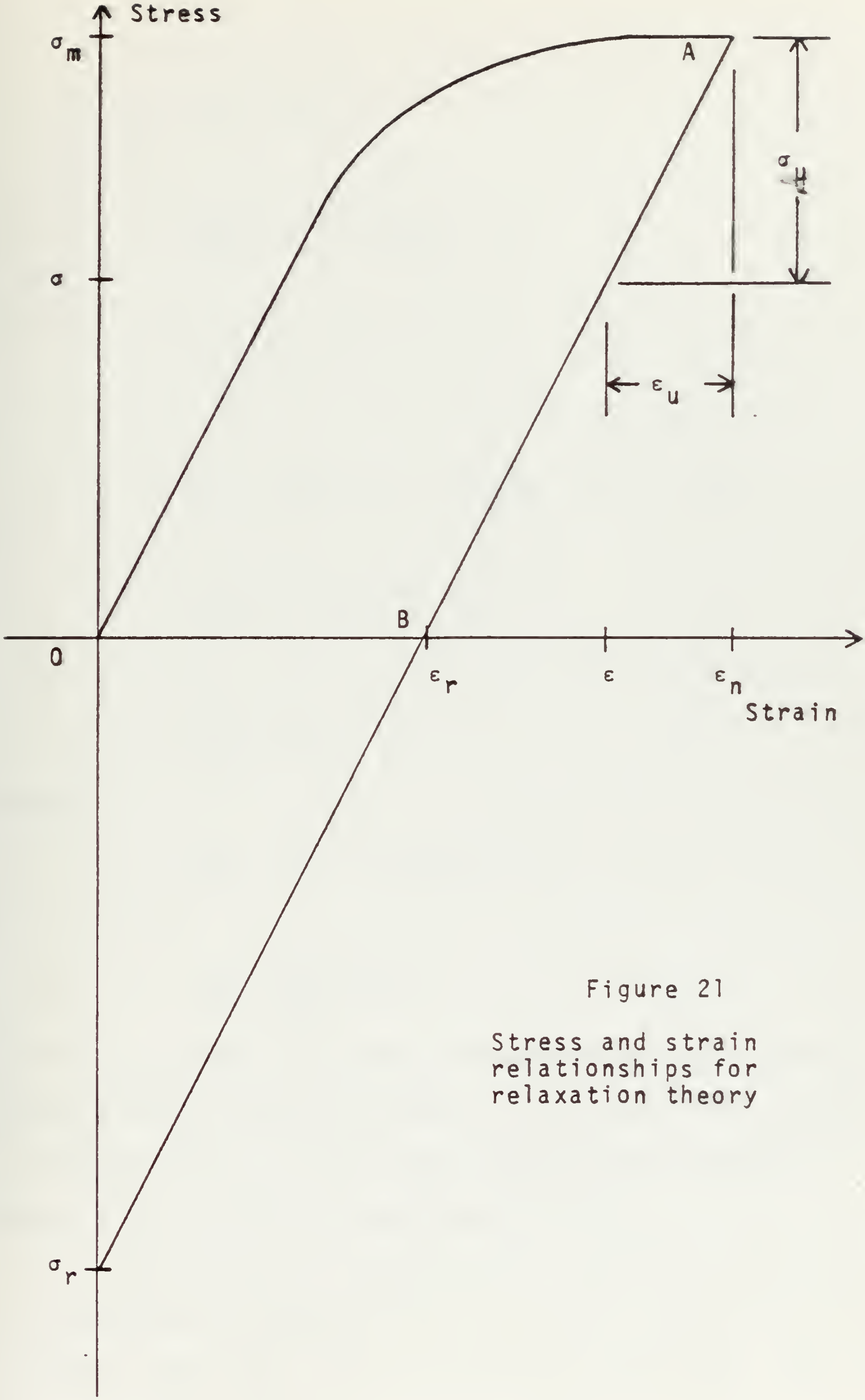


Figure 21

Stress and strain relationships for relaxation theory

Then

$$E\varepsilon_u = K_{tu}S_u$$

or

$$E[\varepsilon_m - \varepsilon] = K_{tu}S_u$$

If

$$\sigma_u = \sigma_m - \sigma$$

and

$$\sigma_u = K_{tu}S_u, \ \sigma_m = K_{tu}S_m, \text{ and } \sigma = K_{tu}S$$

then

$$K_{tu}S_u = K_{tu}S_m - K_{tu}S$$

or

$$S_u = S_m - S$$

Therefore

$$E[\varepsilon_m - \varepsilon] = K_{tu}[S_m - S]$$

or

$$K_{tu} = \frac{\varepsilon_m - \varepsilon}{S_m - S} E .$$

Using this equation the stress concentration factor can be obtained from the measured nominal stress and local strain data from curve A-B. The stress concentration factor is assumed to be constant on subsequent cycles.

Once a stress concentration factor for the curve A-B is obtained, an equation for the local stress at the point of maximum strain in a given cycle may be derived. From the above equations

$$\sigma = \sigma_m - \sigma_u$$

can be written. Then

$$\sigma = \sigma_m - K_{tu} S_u$$

and

$$\sigma = \sigma_m - K_{tu}[S_m - S]$$

follows. For relaxation tests, the nominal stress, S, is a function of the cycle number, N, such that

$$\sigma = \sigma_m - K_{tu} \, [S_m - S(N)] \; .$$

Stress relaxation appears to occur only after the material is yielded; therefore, the maximum local stress, $\sigma_m$, will be assumed to be the yield stress of the material. Thus the local stress at the point of maximum strain in a given cycle can be determined as a function of initial maximum local stress, stress concentration factor, initial maximum nominal stress, and the maximum nominal stress of a given cycle.

This method eliminates the requirement for a cyclic stress-strain curve in local stress calculations. Consideration of Figure 21 indicates that after initial yield the specimen does not follow the monotonic nor the cyclic stress-strain curves as constructed from the uniaxial specimen tests, but rather follows one which is shifted to the right and which is not identical to one passing through the origin of the stress-strain coordinates.

Utilizing the above procedure, a satisfactory calculation can be made of local stress at the hole to provide data for the study of the effects of geometry on local stress relaxation behavior. Thus, with local stress available as a function of the cycle number, N, a determination of stress relaxation behavior could be made.

2. Single Amplitude Cyclic Loading Test Results

The stress and strain data obtained from eleven points on the initial unloading portion of the stress-strain curve of the single amplitude cyclic loading test on a plate (Table 18) were used to calculate stress concentration factors by the method previously developed. Moduli of elasticity of $E = 10.67 \times 10^6$ lbf/in$^2$ from the monotonic stress-strain curve obtained in the cyclic stress-strain curve test, and $E = 10.0 \times 10^6$ lbf/in$^2$ from the single amplitude cyclic loading test on uniaxial specimens, were used. Local strains from both strain gages were also used to provide four stress concentration factors for comparison (Table 19). The values thus calculated were averaged to produce $K_t = 2.85$ for strain gage (1) and $K_t = 2.94$ for strain gage (2), using data from the monotonic stress-strain curve obtained in the uniaxial specimen cyclic stress-strain curve test; and $K_t = 2.66$ for strain gage (1) and $K_t$ 2.76 for strain gage (2), from data obtained in the uniaxial specimen single amplitude cyclic loading test.

Due to the significant variation of the stress concentration factors from the theoretical value, $K_t = 2.57$, and from each other, a more classical method of calculation of stress concentration factors was used to evaluate the results. From the previous development

$$\sigma(N) = \sigma_m - \sigma_u$$

$$\varepsilon(N) = \varepsilon_m - \varepsilon_u$$

and

$$\sigma_u = E\varepsilon_u$$

were obtained and the equation

$$\sigma_u = \sigma_m - \sigma(N) = E\varepsilon_u$$

or

$$\sigma(N) - \sigma_m = - E[\varepsilon_m - \varepsilon(N)]$$

could be written. Then,

$$\sigma(N) = \sigma_m - E[\varepsilon_m - \varepsilon(N)].$$

In order for $K_t = \frac{\sigma}{S}$ to be valid on the unloading portion of the curve, residual stress, $\sigma_r$, must be accounted for such that

$$K_t = \frac{\sigma(N) - \sigma_r}{S(N)}$$

or

$$K_t = \frac{\sigma_m - E[\varepsilon_m - \varepsilon(N)] - \sigma_r}{S(N)}$$

where $\sigma_r = \sigma_m - E[\varepsilon_m - \varepsilon_r]$ and $\varepsilon_r$ is the value of residual strain when nominal stress is zero. The average values calculated by this method (Table 20) were only slightly higher (1%) than those calculated by the orignal method in every case, lending validity to the first calculations. Subsequent data reduction was made using the first stress concentration factors obtained.

Because of the wide variation in stress concentration factors, the calculation of local stresses for the relaxation behavior study was done using all the factors obtained, in order to determine what effects the differences would have in this area.

To calculate local stress for determination of the local stress relaxation behavior, the equation

$$\sigma = \sigma_m - K_t[S_m - S(N)]$$

was applied to the maximum nominal stress in a given cycle. The nominal stress was computed from the recorder plot of output voltage for the initial cycle, to obtain S and $\sigma_m$, and for every fifth cycle throughout the test run (Table 21). The stress concentration factors found using the two moduli of elasticity and data from the two local strain gages were applied to the equation to obtain four values of local stress (Fig. 22 and 23 and Table 22) for a given cycle number, N. As in the uniaxial specimen tests, a least squares exponential curve fit routine was applied to the data to determine equations of the form $\sigma = \sigma_0 e^{-bN}$, describing the local stress

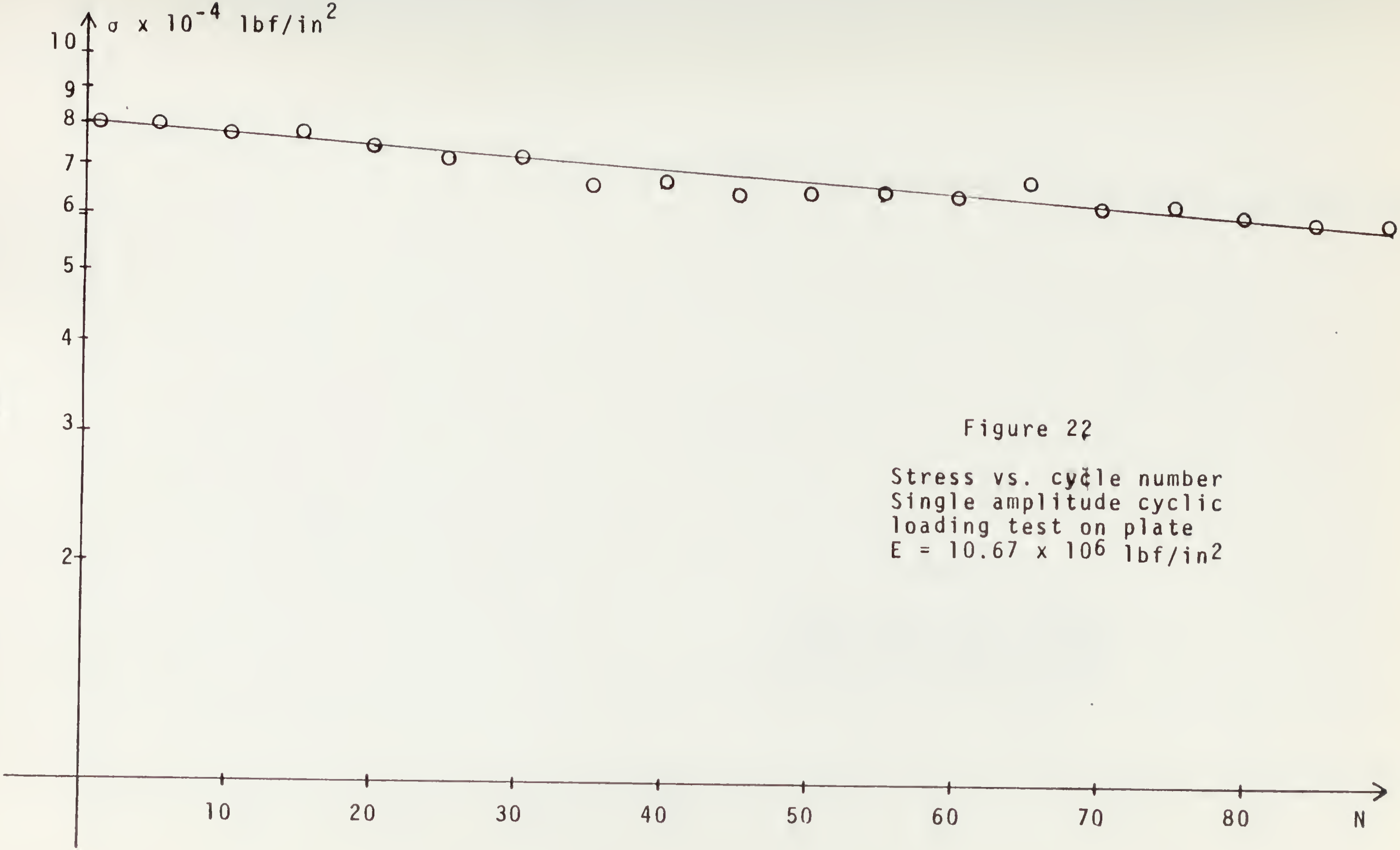


Figure 22

Stress vs. cycle number
Single amplitude cyclic
loading test on plate
$E = 10.67 \times 10^6$ lbf/in$^2$

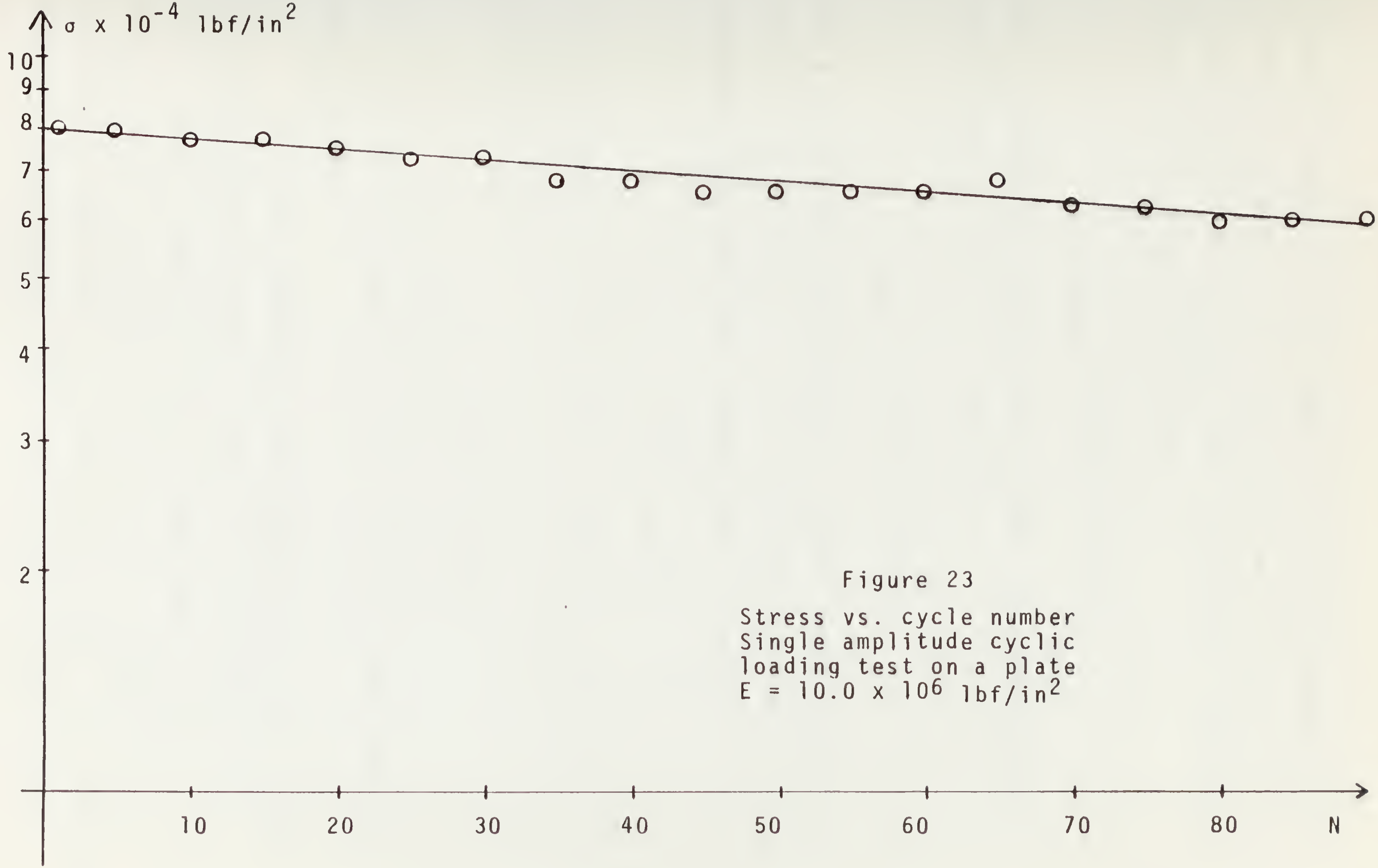


Figure 23

Stress vs. cycle number
Single amplitude cyclic
loading test on a plate
$E = 10.0 \times 10^6$ lbf/in$^2$

relaxation behavior. All four sets of data were used to provide a comparison. For data based on the monotonic stress-strain curve obtained from the uniaxial specimen cyclic stress-strain curve test:

$$\sigma = 79220\ 3^{-(3.427 \times 10^{-3})N}$$

with a correlation coefficient of 0.9650 , was obtained for strain gage (1) with $K_t = 2.85$; and

$$\sigma = 79470\ e^{-(3.752 \times 10^{-3})N}$$

with a correlation coefficient of 0.9650, was obtained for strain page (2) with $K_t = 2.95$. Data based on the uniaxial specimen single amplitude loading test produced

$$\sigma = 79470\ e^{-(3.324 \times 10^{-3})N}$$

with a correlation coefficient of 0.964 for strain gage (1) with $K_t = 2.66$ and

$$\sigma = 79470\ e^{-(3.470 \times 10^{-3})N}$$

with a correlation coefficient of 0.9650 for strain gage (2) with $K_t = 2.76$.

The four equations thus obtained for local stress relaxation behavior exhibit little or no variation in the initial stress; however, there is a maximum variation of 7.52 percent between the stress relaxation rate parameters for a single strain gage but of different data bases. The variation of stress concentration factors was considered the cause of this result. Because the data used to calculate the stress relaxation behavior equations were from equally valid tests, no further conclusions were drawn at this point as to the accuracy of one equation over the other.

3. Dual Amplitude Cyclic Loading Test Results

The stress and strain data taken from six points on the initial unloading portion of the stress-strain curve of the dual amplitude cyclic loading test on a plate (Table 23) were used to calculate stress concentration factors by the previously outlined method.

The moduli of elasticity of $E = 10.67 \times 10^6$ $lbf/in^2$, from the monotonic stress-strain curve obtained in the cyclic stress-strain curve test, and $E = 10.19 \times 10^6$ $lbf/in^2$, from the dual amplitude loading test on uniaxial specimens were used. Local strains from both strain gages were also used (Table 24). Again, the values of stress concentration factors calculated for individual points along the stress-strain curve were averaged. Stress concentration factors thus obtained were $K_t = 3.09$ for strain gage (1), and $K_t = 3.17$ for strain gage (2), based on data from the monotonic stress-strain curve of the cyclic stress-strain curve test on the uniaxial specimen test; and $K_t = 2.95$ for strain gage (1), and $K_t = 3.03$ for strain gage (2), based on the uniaxial specimen dual amplitude loading test.

As in the single amplitude cyclic loading test, significant variation of the calculated stress concentration factors from the theoretical value, $K_t = 2.57$, and between each other, was noted. The classical method previously described was applied to the data of this test (Table 25). The values obtained by this method were also varied, but considerably lower and much closer to the theoretical value of $K_t = 2.57$. Because the results of the single amplitude cyclic loading test exhibited close correlation between $K_t$ values calculated by both methods, and the opposite was found in this test, additional tests were indicated prior to forming a definite conclusion as to the actual value of stress concentration factor on the unloading portion of the stress-strain curve.

In the interests of uniformity of method, the first set of stress concentration factors obtained was used in the calculation of the stress relaxation behavior equations, as was done in the single amplitude cyclic loading test. Due to the variation within this set, all values were used in subsequent calculations.

Local stresses for the determination of local stress relaxation behavior were calculated according to

$$\sigma = \sigma_m - K_t \left[S_m - S(N)\right]$$

Because high stress cycles were alternated with low stress cycles in this test, two sets of stress relaxation behavior data were obtained. Maximum nominal stress and local strain were computed for the peak of the initial cycle to provide $S_m$ and $\sigma_m$ for both high and low stress calculations and then for every fourth cycle thereafter on both high and low peak local strain amplitudes (Table 26). The stress concentration factors found using two moduli of elasticity and data from two local strain gages were applied to the equation to obtain four values of local stress for each cycle number, N (Fig. 24 and 25 and Tables 27 and 21).

To obtain an equation of the form $\sigma = \sigma_0 e^{-bN}$ describing the local stress relaxation behavior of the high stress data, the least squares exponential curve fit routine used in the other uniaxial and plate specimen tests was applied to the data. For data based on the monotonic stress-strain curve obtained in the uniaxial specimen cyclic stress-strain curve test

$$\sigma = 78080\ e^{-(2.593 \times 10^{-3})n}$$

was calculated for local strain gage (1), based on twenty-nine data points with a correlation coefficient of 0.947; and

$$\sigma = 78100\ e^{-(2.671 \times 10^{-3})N}$$

was calculated for strain gage (2), based on twenty-nine data points with a correlation coefficient of 0.947. For data based on the monotonic stress-strain curve obtained in the uniaxial specimen dual amplitude loading test

$$\sigma = 78060\ e^{-(2.457 \times 10^{-3})N}$$

was calculated for strain gage (1), based on twenty-nine points with a correlation coefficient of 0.947; and

$$\sigma = 78070\ e^{-(2.534 \times 10^{-3})N}$$

was calculated for strain gage (2), based on twenty-nine points with a correlation coefficient of 0.947.

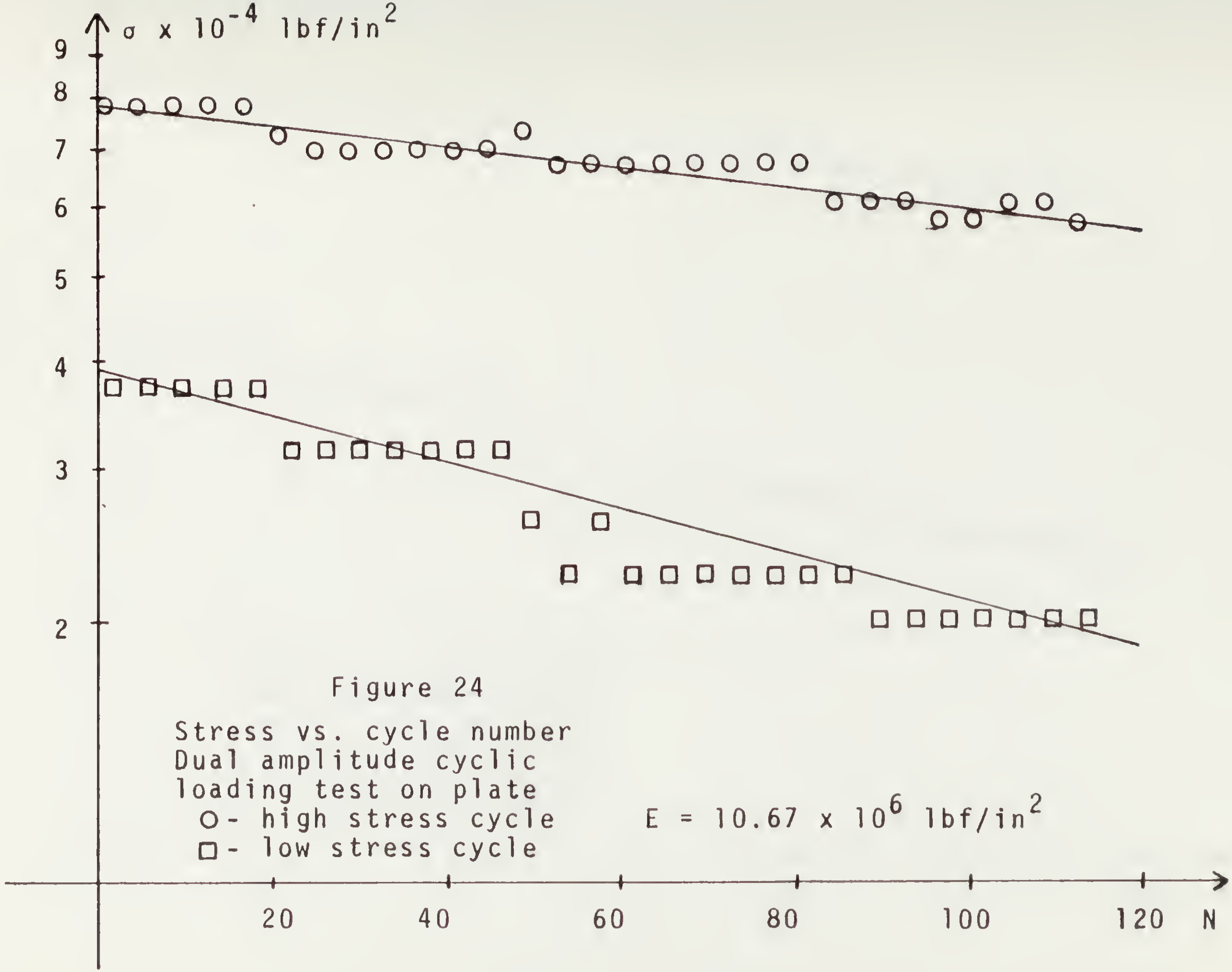


Figure 24

Stress vs. cycle number
Dual amplitude cyclic
loading test on plate
O - high stress cycle
□ - low stress cycle

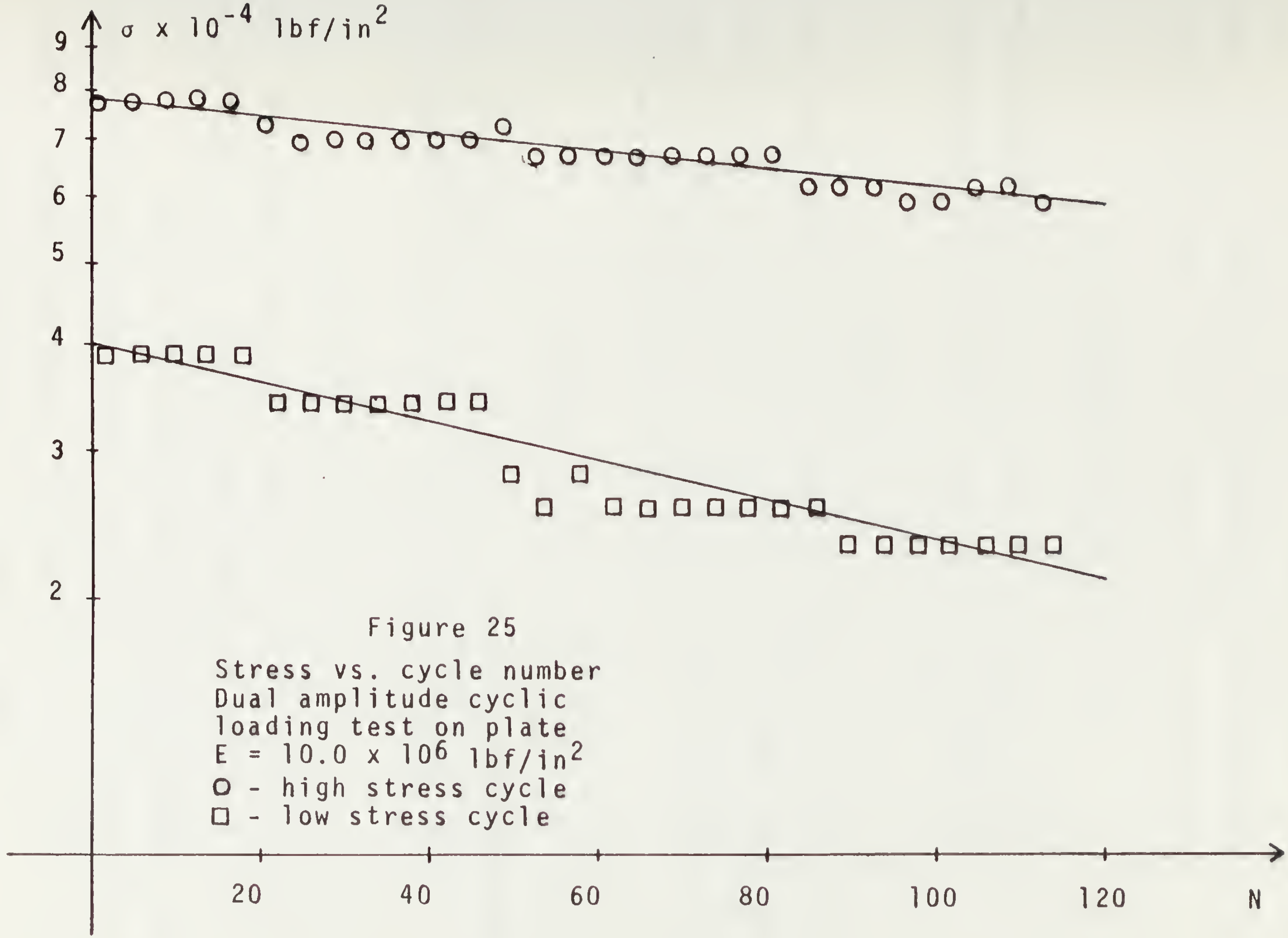


Figure 25

Stress vs. cycle number
Dual amplitude cyclic
loading test on plate
$E = 10.0 \times 10^6$ lbf/in$^2$
O - high stress cycle
□ - low stress cycle

The least squares exponential curve fit routine was also applied to the data for the low stress cycles to obtain the low local stress relaxation behavior. For data based on the monotonic stress-strain curve obtained in the uniaxial specimen cyclic stress-strain curve test

$$\sigma = 39030\ e^{-(6.350 \times 10^{-3})N}$$

was calculated for strain gage (1), based on twenty-nine points with a correlation coefficient of 0.960 and

$$\sigma = 38120\ e^{-(6.843 \times 10^{-3})N}$$

was calculated for strain gage (2), based on twenty-nine points with a correlation coefficient of 0.961. For data based on the monotonic stress-strain curve from the uniaxial specimen dual amplitude loading test

$$\sigma = 40650\ e^{-(5.594 \times 10^{-3})N}$$

was calcualted for strain gage (1), based on twenty-nine points with a correlation coefficient of 0.959 and

$$\sigma = 39720\ e^{-(6.010 \times 10^{-3})N}$$

was calculated for strain gage (2), based on twenty-nine points with a correlation coefficient of 0.960.

The four equations obtained for the stress relaxation behavior of the high stress cycles show excellent correlation between initial stresses. Maximum variation in the relaxation rate parameter was found to be 5.24 percent.

The four equations obtained for the low stress cycles exhibited greater variation in the initial stresses, where a maximum variation of 4.03 percent was found. Maximum variation between the relaxation rate parameters was found to be 12.17 percent.

The variations noted in the stress relaxation behavior equations were considered due, in part, to the variations in stress concentration factors used in the calculations. No further conclusions were drawn due to the equally valid tests from which the data bases were drawn.

4. Discussion of Test Results

The stress concentration factors calculated for local stress computation in the study of stress relaxation behavior were found to be significantly greater than those calculated for the construction of local stress vs. nominal strain curves and the theoretical value computed from plate geometry. An attempt to verify the validity of the stress concentration factors based on data from the unloading portion of the stress-strain curve, by comparison with those calculated by a more classical method, provided two divergent results in two tests, and no conclusion could be offered as to the validity of one stress concentration factor over another at this point. Further tests are warranted to resolve this inconsistency.

Because of the wide variation in stress concentration factors, the calculation of local stresses for the relaxation behavior study was done using all factors obtained in order to determine what effects the differences would have in this area. The local stress relaxation behavior described by equations of the form

$$\sigma = \sigma_0 e^{-bN}$$

can best be compared by considering Table 29, where the parameters $\sigma_0$ and b are listed according to test and data source, and Figure 26, where these parameters are plotted against each other, along with those values obtained in the uniaxial specimen tests under single and dual amplitude cylic loading. Of particular interest is the symmetric grouping of the high stress data points, with respect to the relaxation rate parameter, b, based on plate tests around the points for the uniaxial specimen tests' data points. The average relaxation rate parameter for all ten data points is 3.06, with single amplitude plate test values tending to be higher, dual amplitude plate test values somewhat lower. Although only two plate tests were run, the four values for each plate test, differing due to data base used in the calculations, appear to indicate that the local high stress relaxation rate tends to follow that of the uniaxial specimen rate. The local low stress and uniaxial low stress relaxation parameters are widely scattered and will require additional data to delineate the correct behavior description.

The accuracy of the stress relaxation behavior noted in this study is subject to the variation of the calculated stress concentration factors. This effect would be equally

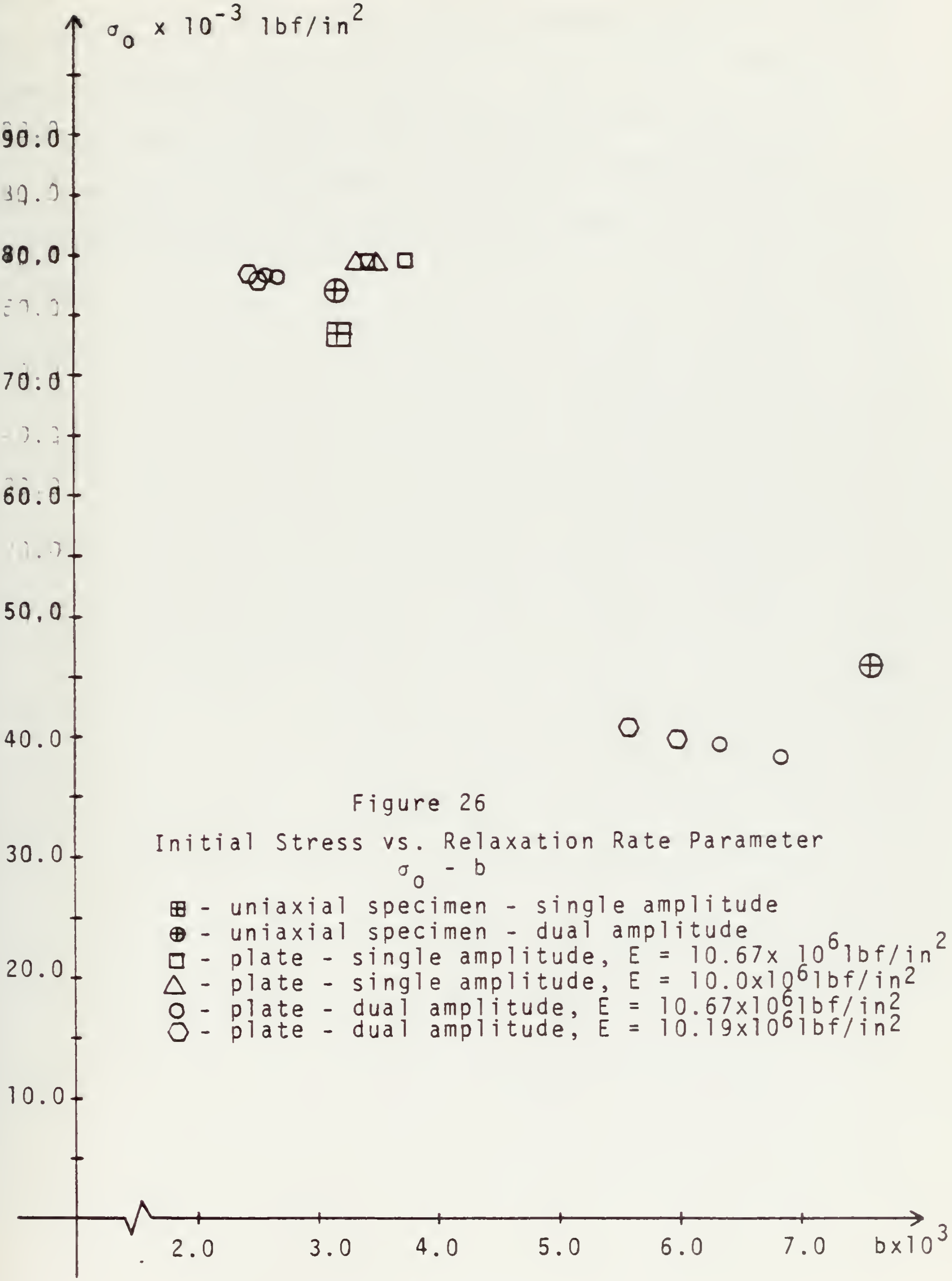


Figure 26

Initial Stress vs. Relaxation Rate Parameter

$\sigma_o$ - b

⊞ - uniaxial specimen - single amplitude
⊕ - uniaxial specimen - dual amplitude
□ - plate - single amplitude, $E = 10.67 \times 10^6 \text{ lbf/in}^2$
△ - plate - single amplitude, $E = 10.0 \times 10^6 \text{ lbf/in}^2$
○ - plate - dual amplitude, $E = 10.67 \times 10^6 \text{ lbf/in}^2$
⬡ - plate - dual amplitude, $E = 10.19 \times 10^6 \text{ lbf/in}^2$

applied to both single and dual amplitude cyclic loading test results, and therefore would not affect the relationship between the data points of the two on the initial stress vs. relaxation rate parameter curve significantly (Fig. 26). Thus, the conclusions drawn on relaxation behavior were considered to be qualitatively, if not quantitatively, accurate.

# IV. CONCLUSIONS ON TEST RESULTS

The major areas of interest in the tests on uniaxial and plate specimens were: (1) determining whether Neuber's relationship, $K_t^2 = \frac{\sigma\varepsilon}{Se}$ , would provide an accurate, practical method of calculating local stress, with knowledge of the material properties and nominal strain alone, in a structure subject to stress concentrations, (2) determining whether the local stress in the specimen follows the stress relaxation behavior of the uniaxial specimen; and (3) determining whether the type of loading applied to both uniaxial and plate specimens alters the stress relaxation behavior.

Neuber's theory proved to be a valid basis on which to establish a method of calculating local stress at a stress concentration in a structure based on knowledge of nominal strain and material properties alone. Because Neuber's method provides only the initial monotonic local stress in the structure, the stress relaxation behavior must be known to utilize the method in practical fatigue life determination.

In the area of stress relaxation behavior, further study of the calculation of stress concentration factors is warranted. The values calculated for the unloading portion of the stress-strain curve ranged from $K_t$ = 2.66 to $K_t$ = 3.17, significantly different than the corresponding stress concentration factors calulated on the initial loading cycle of the same curve, and varying greatly among themselves.

Additional tests are recommended to determine whether the stress concentration factor does vary from the loading to unloading portions of the curve, and whether a consistent factor can be obtained for the unloading segment. However, because a number of stress concentration factors were calculated and used to determine stress relaxation behavior, it is felt that essentially valid conclusions can be drawn regarding relaxation behavior without detrimental influence due to possibly incorrect values of stress concentration factors.

A comparison of stress relaxation behavior obtained in the uniaxial and plate specimen tests indicated that, when the material is cycled repeatedly into the yield stress range, the relaxation rate tends to be low, in the area of $b = 3.00 \times 10^{-3}$ (Fig. 26), regardless of the loading situation of the geometric configuration. Due to the scatter realized in the low stress relaxation behavior data, no conclusion could be drawn in this area other than the fact that relaxation rate parameters are significantly higher than those of the high stress behaviors. Additionally, it appeared that the type of loading situation had some influence on the high stress relaxation rates found in the plate tests. This was not the case in the uniaxial specimens, which would indicate that some combined influences of geometric effects and loading history were present in the plates with regard to stress relaxation behavior. Further tests are necessary to establish the stress relaxation behavior throughout the range of stress from high to low values. These tests would hopefully reveal a relationship between initial stress and relaxation rate which would reduce the present form of the equation.

$$\sigma = \sigma_o e^{-bN}$$

to a form involving relaxation rate, b, as a function of initial stress, $\sigma_o$, or

$$\sigma = \sigma_o e^{-a(\sigma_o)N}$$

thus facilitating the calculation of local stress at a given cycle. The establishment of a valid stress relaxation behavior equation is necessary in order to extend the use of Neuber's method to practical situations where structures do, in fact, cycle repeatedly. The local stress-nominal strain relationships are valid only for the initial cycle, after which stress relaxation behavior must be applied to obtain accurate knowledge of local stress for fatigue life studies.

In general, the conclusions were that the initial local stress in a structure subject to geometric effects can be obtain readily and accurately by applying Neuber's method; and that once the initial local stress is known, a stress relaxation behavior equation of the form

$$\sigma = \sigma_o e^{-bN}$$

can be applied to obtain local stresses at given cycles for studies involving fatigue life estimation in aircraft structures.

## APPENDIX A - TABULAR DATA

TABLE 1

Strain data and percent bending moment from alignment evaluation on MTS test system's load cell.

| Strain Gage 1 | Strain Gage 2 | Strain Gage 3 | Strain Gage 4 | Maximum Strain | Average Strain | Percent Bending Moment |
|---|---|---|---|---|---|---|
| 0 | 0 | 0 | 0 | 0 | 0 | 0 |
| 939 | 983 | 966 | 920 | 983 | 952 | 3.26 |
| 1404 | 1479 | 1458 | 1385 | 1479 | 1431.5 | 3.32 |
| 1780 | 1875 | 1849 | 1757 | 1875 | 1815.25 | 3.29 |
| 1399 | 1481 | 1455 | 1380 | 1481 | 1428.75 | 3.66 |
| 928 | 990 | 965 | 909 | 990 | 948.00 | 4.43 |
| - 979 | - 931 | - 920 | - 959 | - 979 | - 947.25 | 3.35 |
| -1452 | -1404 | -1381 | -1417 | -1452 | -1413.50 | 2.72 |
| -1828 | -1781 | -1749 | -1785 | -1828 | -1785.75 | 2.37 |
| -1450 | -1390 | -1384 | -1431 | -1450 | -1413.75 | 2.56 |
| - 972 | - 910 | - 925 | - 974 | - 974 | - 945.25 | 3.04 |

Percent bending moment calculated according to:

$$\% \text{ Bending} = \frac{\text{max - avg}}{\text{avg}} \times 100$$

Strain gages were calibrated such that
1.0 VDC = 1000 $\mu$in/in

TABLE 2

Cyclic stress and strain data from cyclic stress-strain curve test on a uniaxial specimen.

TENSILE LOADS

| Cycle | Load Volts | Strain Volts | Stress $lbf/in^2$ | Strain $\mu in/in$ |
|---|---|---|---|---|
| 1 | 7.80 | 7.40 | 78000 | 11339 |
| 2 | 7.75 | 7.05 | 77500 | 10802 |
| 3 | 7.60 | 6.80 | 76000 | 10419 |
| 4 | 7.50 | 6.50 | 75000 | 9960 |
| 5 | 7.30 | 6.00 | 73000 | 9193 |
| 6 | 7.00 | 5.30 | 70000 | 8121 |
| 7 | 6.40 | 4.50 | 64000 | 6895 |
| 8 | 5.30 | 3.55 | 53000 | 5439 |
| 9 | 3.90 | 2.50 | 39000 | 3831 |

COMPRESSIVE LOADS

| Cycle | Load Volts | Strain Volts | Stress $lbf/in^2$ | Strain $\mu in/in$ |
|---|---|---|---|---|
| 1 | -7.90 | -7.35 | -79000 | -11262 |
| 2 | -7.90 | -7.25 | -79000 | -11109 |
| 3 | -7.85 | -7.00 | -78500 | -10726 |
| 4 | -7.80 | -6.55 | -78000 | -10036 |
| 5 | -7.60 | -5.95 | -76000 | - 9117 |
| 6 | -7.20 | -5.20 | -72000 | - 7968 |
| 7 | -6.30 | -4.30 | -63000 | - 6589 |
| 8 | -5.00 | -3.35 | -50000 | - 5133 |

CALIBRATION

Load: 1.0 VDC = 10,000 lbf

Strain: 1.0 VDC = 153223 $\mu in/in$

Stress: 1.0 VDC = 10,000 $lbf/in^2$

TABLE 3

Monotonic stress and strain data from cyclic stress-strain curve test on a uniaxial specimen.

| Load Volts | Strain Volts | Stress $lbf/in^2$ | Strain $\mu in/in$ |
|---|---|---|---|
| 1.00 | 0.60 | 10000 | 919 |
| 2.00 | 1.20 | 20000 | 1839 |
| 3.00 | 1.85 | 30000 | 2835 |
| 4.00 | 2.5 | 40000 | 3831 |
| 5.00 | 3.10 | 50000 | 4750 |
| 6.00 | 3.80 | 60000 | 5822 |
| 7.00 | 4.20 | 70000 | 6435 |
| 7.80 | 5.20 | 78000 | 7968 |
| 7.80 | 7.40 | 78000 | 11339 |

CALIBRATION

Load: 1.0 VDC = 10,000 lbf

Strain: 1.0 VDC = 1532.23 $\mu in/in$

Stress: 1.0 VDC = 10,000 $lbf/in^2$

TABLE 4

Monotonic stress and strain data from single amplitude cyclic loading test on a uniaxial specimen.

| Load Volts | Strain Volts | Stress $lbf/in^2$ | Strain $\mu in/in$ | Stress x Strain $lbf/in^2$ |
|---|---|---|---|---|
| 0.00 | 0.00 | 0.00 | 0.00 | 0.00 |
| 0.50 | 0.40 | 5000 | 612 | 3.062 |
| 1.00 | 0.70 | 10000 | 1071 | 10.71 |
| 1.50 | 1.10 | 15000 | 1683 | 25.25 |
| 2.00 | 1.35 | 20000 | 2065 | 41.30 |
| 2.50 | 1.65 | 25000 | 2524 | 63.10 |
| 3.00 | 2.00 | 30000 | 3060 | 91.80 |
| 3.50 | 2.30 | 35000 | 3518 | 123.13 |
| 4.00 | 2.60 | 40000 | 3977 | 159.08 |
| 4.50 | 2.95 | 45000 | 4513 | 203.09 |
| 5.00 | 3.25 | 50000 | 4972 | 248.60 |
| 5.50 | 3.58 | 55000 | 5477 | 301.24 |
| 6.00 | 3.90 | 60000 | 5966 | 357.96 |
| 6.50 | 4.25 | 65000 | 6501 | 422.57 |
| 7.00 | 4.60 | 70000 | 7037 | 492.59 |
| 7.20 | 4.75 | 72000 | 7266 | 523.15 |
| 7.40 | 4.90 | 74000 | 7496 | 554.70 |
| 7.50 | 5.00 | 75000 | 7649 | 573.68 |
| 7.60 | 5.10 | 76000 | 7802 | 592.95 |
| 7.70 | 5.30 | 77000 | 8108 | 624.32 |
| 7.80 | 5.40 | 78000 | 8261 | 644.36 |
| 7.90 | 6.00 | 79000 | 9179 | 725.14 |
| 8.00 | 6.60 | 80000 | 10096 | 807.68 |
| 8.00 | 7.30 | 80000 | 11167 | 893.36 |

CALIBRATION

Load: 1.0 VDC = 10,000 lbf

Strain: 1.0 VDC = 1529.76 $\mu in/in$

Stress: 1.0 VDC = 10,000 $lbf/in^2$

TABLE 5

Stress and cycle number data from single amplitude cyclic loading test on a uniaxial specimen.

| Cycle | Load Volts | Stress $lbf/in^2$ | Cycle | Load Volts | Stress $lbf/in^2$ |
|---|---|---|---|---|---|
| 1 | 8.00 | 80000 | 45 | 6.30 | 63000 |
| 2 | 7.85 | 78500 | 46 | 6.30 | 63000 |
| 3 | 7.80 | 78000 | 47 | 6.30 | 63000 |
| 4 | 7.75 | 77500 | 48 | 6.25 | 62500 |
| 5 | 7.70 | 77000 | 49 | 6.25 | 62500 |
| 6 | 7.65 | 76500 | 50 | 6.20 | 62000 |
| 7 | 7.60 | 76000 | 51 | 6.20 | 62000 |
| 8 | 7.60 | 76000 | 52 | 6.15 | 61500 |
| 9 | 7.55 | 75500 | 53 | 6.10 | 61000 |
| 10 | 7.50 | 75000 | 54 | 6.05 | 60500 |
| 11 | 7.50 | 75000 | 55 | 6.05 | 60500 |
| 12 | 7.45 | 74500 | 56 | 6.00 | 60000 |
| 13 | 7.40 | 74000 | 57 | 6.00 | 60000 |
| 14 | 7.40 | 74000 | 58 | 6.00 | 60000 |
| 15 | 7.35 | 73500 | 59 | 5.95 | 59000 |
| 16 | 7.30 | 73000 | 60 | 5.90 | 59000 |
| 17 | 7.25 | 72500 | 61 | 5.90 | 59000 |
| 18 | 7.20 | 72000 | 62 | 5.90 | 59000 |
| 19 | 7.15 | 71500 | 63 | 5.85 | 58500 |
| 20 | 7.10 | 71000 | 64 | 5.85 | 58500 |
| 21 | 7.10 | 71000 | 65 | 5.80 | 58000 |
| 22 | 7.05 | 70500 | 66 | 5.80 | 58000 |
| 23 | 7.05 | 70500 | 67 | 5.80 | 58000 |
| 24 | 7.00 | 70000 | 68 | 5.75 | 57500 |
| 25 | 6.95 | 69500 | 69 | 5.70 | 57000 |
| 26 | 6.90 | 69000 | 70 | 5.70 | 57000 |
| 27 | 6.90 | 69000 | 71 | 5.70 | 57000 |
| 28 | 6.85 | 68500 | 72 | 5.65 | 56500 |
| 29 | 6.80 | 68000 | 73 | 5.65 | 56500 |
| 30 | 6.80 | 68000 | 74 | 5.60 | 56000 |
| 31 | 6.75 | 67500 | 75 | 5.60 | 56000 |
| 32 | 6.70 | 67000 | 76 | 5.60 | 56000 |
| 33 | 6.70 | 67000 | 77 | 5.55 | 55500 |
| 34 | 6.65 | 66500 | 78 | 5.55 | 55500 |
| 35 | 6.60 | 66000 | 79 | 5.50 | 55000 |
| 36 | 6.60 | 66000 | 80 | 5.50 | 55000 |
| 37 | 6.55 | 65500 | 81 | 5.50 | 55000 |
| 38 | 6.50 | 65000 | 82 | 5.45 | 54500 |
| 39 | 6.50 | 65000 | 83 | 5.45 | 54500 |
| 40 | 6.45 | 64500 | 84 | 5.40 | 54000 |
| 41 | 6.40 | 64000 | 85 | 5.40 | 54000 |
| 42 | 6.40 | 64000 | 86 | 5.40 | 54000 |
| 43 | 6.40 | 64000 | 87 | 5.40 | 54000 |
| 44 | 6.35 | 63500 | 88 | 5.35 | 53500 |

| Cycle | Load Volts | Stress lbf/in$^2$ | Cycle | Load Volts | Stress lbf/in$^2$ |
|---|---|---|---|---|---|
| 89 | 5.35 | 53500 | 140 | 4.60 | 46000 |
| 90 | 5.30 | 53000 | 141 | 4.60 | 46000 |
| 91 | 5.30 | 53000 | 142 | 4.60 | 46000 |
| 92 | 5.30 | 53000 | 143 | 4.60 | 46000 |
| 93 | 5.30 | 53000 | 144 | 4.60 | 46000 |
| 94 | 5.30 | 53000 | 145 | 4.55 | 45500 |
| 95 | 5.25 | 52500 | 146 | 4.55 | 45500 |
| 96 | 5.25 | 52500 | 147 | 4.55 | 45000 |
| 97 | 5.20 | 52000 | 148 | 4.50 | 45000 |
| 98 | 5.20 | 52000 | 149 | 4.50 | 45000 |
| 99 | 5.20 | 52000 | 150 | 4.50 | 45000 |
| 100 | 5.20 | 52000 | 151 | 4.50 | 45000 |
| 101 | 5.15 | 51500 | 152 | 4.50 | 45000 |
| 102 | 5.15 | 51500 | 153 | 4.50 | 45000 |
| 103 | 5.10 | 51000 | 154 | 4.45 | 44500 |
| 104 | 5.10 | 51000 | 155 | 4.45 | 44500 |
| 105 | 5.10 | 51000 | 156 | 4.40 | 44000 |
| 106 | 5.10 | 51000 | 157 | 4.40 | 44000 |
| 107 | 5.05 | 50500 | 158 | 4.40 | 44000 |
| 108 | 5.05 | 50500 | 159 | 4.40 | 44000 |
| 109 | 5.00 | 50000 | 160 | 4.40 | 44000 |
| 110 | 5.00 | 50000 | 161 | 4.35 | 43500 |
| 111 | 5.00 | 50000 | 162 | 4.35 | 43500 |
| 112 | 5.00 | 50000 | 163 | | |
| 113 | 5.00 | 50000 | 164 | | |
| 114 | 4.95 | 49500 | 165 | | |
| 115 | 4.95 | 49500 | 166 | | |
| 116 | 4.95 | 49500 | 167 | | |
| 117 | 4.90 | 49000 | 168 | | |
| 118 | 4.90 | 49000 | 169 | | |
| 119 | 4.90 | 49000 | 170 | | |
| 120 | 4.90 | 49000 | 171 | | |
| 121 | 4.85 | 48500 | 172 | | |
| 122 | 4.85 | 48500 | 173 | | |
| 123 | 4.85 | 48500 | 174 | | |
| 124 | 4.80 | 48000 | 175 | | |
| 125 | 4.80 | 48000 | 176 | | |
| 126 | 4.80 | 48000 | 177 | 4.15 | 41500 |
| 127 | 4.80 | 48000 | 178 | 4.15 | 41500 |
| 128 | 4.75 | 47500 | 179 | 4.10 | 41000 |
| 129 | 4.75 | 47500 | 180 | 4.10 | 41000 |
| 130 | 4.75 | 47500 | 181 | 4.10 | 41000 |
| 131 | 4.75 | 47500 | 182 | 4.05 | 40500 |
| 132 | 4.70 | 47000 | 183 | 4.05 | 40500 |
| 133 | 4.70 | 47000 | 184 | 4.05 | 40500 |
| 134 | 4.70 | 47000 | 185 | 4.00 | 40000 |
| 135 | 4.70 | 47000 | 186 | 4.00 | 40000 |
| 136 | 4.65 | 46500 | 187 | 4.00 | 40000 |
| 137 | 4.65 | 46500 | 188 | 4.00 | 40000 |
| 138 | 4.65 | 46500 | 189 | 4.00 | 40000 |
| 139 | 4.60 | 46000 | 190 | 4.00 | 40000 |

| Cycle | Load Volts | Stress $lbf/in^2$ | Cycle | Load Volts | Stress $lbf/in^2$ |
|---|---|---|---|---|---|
| 191 | 4.00 | 40000 | 234 | 3.60 | 36000 |
| 192 | 3.95 | 39500 | 235 | 3.55 | 35500 |
| 193 | 3.95 | 39500 | 236 | 3.55 | 35500 |
| 194 | 3.95 | 39500 | 237 | 3.55 | 35500 |
| 195 | 3.90 | 39000 | 238 | 3.55 | 35500 |
| 196 | 3.90 | 39000 | 239 | 3.55 | 35500 |
| 197 | 3.90 | 39000 | 240 | 3.50 | 35000 |
| 198 | 3.90 | 39000 | 241 | 3.50 | 35000 |
| 199 | 3.90 | 39000 | 242 | 3.50 | 35000 |
| 200 | 3.85 | 38500 | 243 | 3.50 | 35000 |
| 201 | 3.85 | 38500 | 244 | 3.50 | 35000 |
| 202 | 3.85 | 38500 | 245 | 3.50 | 35000 |
| 203 | 3.80 | 38000 | 246 | 3.45 | 34500 |
| 204 | 3.80 | 38000 | 247 | 3.45 | 34500 |
| 205 | 3.80 | 38000 | 248 | 3.45 | 34500 |
| 206 | 3.80 | 38000 | 249 | 3.45 | 34500 |
| 207 | 3.80 | 38000 | 250 | 3.40 | 34000 |
| 208 | 3.80 | 38000 | 251 | 3.40 | 34000 |
| 209 | 3.75 | 37500 | 252 | 3.40 | 34000 |
| 210 | 3.75 | 37500 | 253 | 3.40 | 34000 |
| 211 | 3.75 | 37500 | 254 | 3.35 | 33500 |
| 212 | 3.75 | 37500 | 255 | 3.35 | 33500 |
| 213 | 3.75 | 37500 | 256 | 3.30 | 33000 |
| 214 | 3.75 | 37500 | 257 | 3.30 | 33000 |
| 215 | 3.75 | 37500 | 258 | 3.30 | 33000 |
| 216 | 3.70 | 37000 | 259 | 3.25 | 32500 |
| 217 | 3.70 | 37000 | 260 | 3.20 | 32000 |
| 218 | 3.70 | 37000 | 261 | 3.20 | 32000 |
| 219 | 3.70 | 37000 | 262 | 3.20 | 32000 |
| 220 | 3.70 | 37000 | 263 | 3.15 | 31500 |
| 221 | 3.70 | 37000 | 264 | 3.10 | 31000 |
| 222 | 3.70 | 37000 | 265 | 3.10 | 31000 |
| 223 | 3.70 | 37000 | 266 | 3.05 | 30500 |
| 224 | 3.70 | 37000 | 267 | 3.00 | 30000 |
| 225 | 3.65 | 36500 | 268 | 3.00 | 30000 |
| 226 | 3.65 | 36500 | 269 | 2.85 | 28500 |
| 227 | 3.65 | 36500 | 270 | 2.65 | 26500 |
| 228 | 3.60 | 36000 | 271 | 2.50 | 25000 |
| 229 | 3.60 | 36000 | 272 | 2.40 | 24000 |
| 230 | 3.60 | 36000 | 273 | 2.40 | 24000 |
| 231 | 3.60 | 36000 | 274 | 2.35 | 23500 |
| 232 | 3.60 | 36000 | 275 | 2.30 | 23000 |
| 233 | 3.60 | 36000 | | | |

CALIBRATION

Load: 1.0 VDC = 10,000 lbf

Stress: 1.0 VDC = 10,000 $lbf/in^2$

TABLE 6

Monotonic stress and strain data from dual amplitude cyclic loading test on a uniaxial specimen.

| Load Volts | Strain Volts | Stress $lbf/in^2$ | Strain $\mu in/in$ | Stress x Strain $lbf/in^2$ |
|---|---|---|---|---|
| 0 | 0 | 0 | 0 | 0 |
| 0.50 | 0.30 | 5000 | 459 | 2.295 |
| 1.00 | 0.60 | 10000 | 918 | 9.180 |
| 1.50 | 0.90 | 15000 | 1377 | 20.655 |
| 2.00 | 1.25 | 20000 | 1912 | 38.240 |
| 2.50 | 1.55 | 25000 | 2371 | 59.275 |
| 3.00 | 1.90 | 30000 | 2907 | 87.210 |
| 3.50 | 2.20 | 35000 | 3365 | 117.78 |
| 4.00 | 2.55 | 40000 | 3901 | 156.04 |
| 4.50 | 2.90 | 45000 | 4436 | 199.62 |
| 5.00 | 3.20 | 50000 | 4895 | 244.75 |
| 5.50 | 3.55 | 55000 | 5431 | 298.71 |
| 6.00 | 3.90 | 60000 | 5966 | 357.96 |
| 6.50 | 4.30 | 65000 | 6578 | 427.57 |
| 6.60 | 4.40 | 66000 | 6731 | 444.25 |
| 6.70 | 4.55 | 67000 | 6960 | 466.32 |
| 6.80 | 4.60 | 68000 | 7037 | 478.52 |
| 6.90 | 4.70 | 69000 | 7190 | 496.11 |
| 7.00 | 4.80 | 70000 | 7343 | 514.01 |
| 7.10 | 5.00 | 71000 | 7649 | 543.08 |
| 7.20 | 5.05 | 72000 | 7729 | 556.49 |
| 7.30 | 5.25 | 73000 | 8031 | 586.26 |
| 7.40 | 5.40 | 74000 | 8261 | 611.31 |
| 7.50 | 5.50 | 75000 | 8414 | 631.05 |
| 7.60 | 5.60 | 76000 | 8567 | 651.09 |
| 7.70 | 5.80 | 77000 | 8873 | 683.22 |
| 7.80 | 6.00 | 78000 | 9179 | 715.96 |
| 7.80 | 6.20 | 78000 | 9485 | 739.83 |
| 7.80 | 7.10 | 78000 | 10861 | 847.16 |

CALIBRATION

Load: 1.0 VDC = 10,000 lbf

Strain: 1.0 VDC = 1532.23 $\mu in/in$

Stress: 1.0 VDC = 10,000 $lbf/in^2$

TABLE 7

Stress and cyclic number data from dual amplitude cyclic loading test on a uniaxial specimen test.

| Cycle | Load Volts | Stress $lbf/in^2$ | Cycle | Load Volts | Stress $lbf/in^2$ |
|---|---|---|---|---|---|
| 1 | 7.80 | 78000 | 47 | 6.60 | 66000 |
| 2 | 4.60 | 46000 | 48 | 3.20 | 32000 |
| 3 | 7.65 | 765000 | 49 | 6.60 | 66000 |
| 4 | 4.40 | 44000 | 50 | 3.15 | 31500 |
| 5 | 7.60 | 76000 | 51 | 6.55 | 66500 |
| 6 | 4.30 | 43000 | 52 | 3.10 | 31000 |
| 7 | 7.50 | 75000 | 53 | 6.50 | 65000 |
| 8 | 4.20 | 42000 | 54 | 3.05 | 30500 |
| 9 | 7.45 | 74500 | 55 | 6.45 | 64500 |
| 10 | 4.15 | 41500 | 56 | 3.05 | 30500 |
| 11 | 7.40 | 74000 | 57 | 6.40 | 64000 |
| 12 | 4.05 | 40500 | 58 | 3.00 | 30000 |
| 13 | 7.40 | 74000 | 59 | 6.40 | 64000 |
| 14 | 4.00 | 40000 | 60 | 2.95 | 29500 |
| 15 | 7.30 | 73000 | 61 | 6.35 | 63500 |
| 16 | 3.95 | 39500 | 62 | 2.90 | 29000 |
| 17 | 7.30 | 73000 | 63 | 6.30 | 63000 |
| 18 | 3.90 | 39000 | 64 | 2.85 | 28500 |
| 19 | 7.25 | 72500 | 65 | 6.25 | 62500 |
| 20 | 3.85 | 38500 | 66 | 2.85 | 28500 |
| 21 | 7.20 | 72000 | 67 | 6.20 | 62000 |
| 22 | 3.80 | 38000 | 68 | 2.80 | 28000 |
| 23 | 7.15 | 71500 | 69 | 6.20 | 62000 |
| 24 | 3.75 | 37500 | 70 | 2.75 | 27500 |
| 25 | 7.10 | 71000 | 71 | 6.15 | 61500 |
| 26 | 3.70 | 37000 | 72 | 2.70 | 27000 |
| 27 | 7.05 | 70500 | 73 | 6.10 | 61000 |
| 28 | 3.65 | 36500 | 74 | 2.65 | 26500 |
| 29 | 7.00 | 70000 | 75 | 6.05 | 60500 |
| 30 | 3.60 | 36000 | 76 | 2.60 | 26000 |
| 31 | 7.00 | 70000 | 77 | 6.00 | 60000 |
| 32 | 3.55 | 35500 | 78 | 2.60 | 26000 |
| 33 | 6.95 | 69500 | 79 | 6.00 | 60000 |
| 34 | 3.50 | 35000 | 80 | 2.55 | 25500 |
| 35 | 6.90 | 69000 | 81 | 5.95 | 59500 |
| 36 | 3.45 | 34500 | 82 | 2.45 | 24500 |
| 37 | 6.85 | 68500 | 83 | 5.90 | 59000 |
| 38 | 3.40 | 34000 | 84 | 2.45 | 24500 |
| 39 | 6.80 | 68000 | 85 | 5.85 | 58500 |
| 40 | 3.35 | 33500 | 86 | 2.40 | 24000 |
| 41 | 6.75 | 67500 | 87 | 5.80 | 58000 |
| 42 | 3.30 | 33000 | 88 | 2.35 | 23500 |
| 43 | 6.70 | 67000 | 89 | 5.80 | 58000 |
| 44 | 3.30 | 33000 | 90 | 2.30 | 23000 |
| 45 | 6.65 | 66500 | 91 | 5.75 | 57500 |
| 46 | 3.25 | 32500 | 92 | 2.30 | 23000 |

| Cycle | Load Volts | Stress $lbf/in^2$ | Cycle | Load Volts | Stress $lbf/in^2$ |
|---|---|---|---|---|---|
| 93 | 5.70 | 57000 | 117 | 5.30 | 53000 |
| 94 | 2.25 | 22500 | 118 | 1.85 | 18500 |
| 95 | 5.70 | 57000 | 119 | 5.25 | 52500 |
| 96 | 2.25 | 22500 | 120 | 1.80 | 18000 |
| 97 | 5.65 | 56500 | 121 | 5.20 | 52000 |
| 98 | 2.20 | 22000 | 122 | 1.80 | 18000 |
| 99 | 5.60 | 56000 | 123 | 5.20 | 52000 |
| 100 | 2.20 | 22000 | 124 | 1.75 | 17500 |
| 101 | 5.60 | 56000 | 125 | 5.20 | 52000 |
| 102 | 2.15 | 21500 | 126 | 1.75 | 17500 |
| 103 | 5.60 | 56000 | 127 | 5.15 | 51500 |
| 104 | 2.10 | 21000 | 128 | 1.70 | 17000 |
| 105 | 5.50 | 55000 | 129 | 5.10 | 51000 |
| 106 | 2.05 | 20500 | 130 | 1.65 | 16500 |
| 107 | 5.50 | 55000 | 131 | 5.10 | 51000 |
| 108 | 2.00 | 20000 | 132 | 1.65 | 16500 |
| 109 | 5.45 | 54500 | 133 | 5.05 | 50500 |
| 110 | 2.00 | 20000 | 134 | 1.60 | 16000 |
| 111 | 5.40 | 54000 | 135 | 5.05 | 50500 |
| 112 | 2.00 | 20000 | 136 | 1.60 | 16000 |
| 113 | 5.40 | 54000 | 137 | 5.00 | 50000 |
| 114 | 1.95 | 19500 | 138 | 1.55 | 15500 |
| 115 | 5.35 | 53500 | 139 | 5.00 | 50000 |
| 116 | 1.90 | 19000 | 140 | 1.50 | 15000 |

CALIBRATION

Load: 1.0 VDC = 10,000 lbf

Stress: 1.0 VDC = 10,000 $lbf/in^2$

TABLE 8

Stress and strain data from strain gage placement tests and comparison of experimental and theoretical strains, maximum and average. Run number 1

| Load Volts | Average Strain Volts | Maximum Strain Volts | Stress $lbf/in^2$ | Exp. Avg. Strain $\mu in/in$ | Theo. Avg Strain $\mu in/in$ | Exp. Max. Strain $\mu in/in$ | Theo. Max. Strain $\mu in/in$ |
|---|---|---|---|---|---|---|---|
| 1.054 | 2.018 | 2.195 | 1996 | 496 | 507 | 587 | 565 |
| 2.000 | 2.479 | 2.725 | 3788 | 957 | 962 | 1117 | 1072 |
| 2.993 | 2.957 | 3.278 | 5669 | 1435 | 1440 | 1670 | 1604 |
| 4.021 | 3.459 | 3.853 | 7615 | 1937 | 1934 | 2245 | 2155 |
| 5.021 | 3.948 | 4.413 | 9509 | 2426 | 2416 | 2805 | 2691 |
| 6.011 | 4.435 | 4.971 | 11384 | 2913 | 2892 | 3363 | 3222 |
| 7.004 | 4.924 | 5.532 | 13265 | 3402 | 3369 | 3924 | 3754 |
| 7.998 | 5.417 | 6.096 | 15148 | 3895 | 3848 | 4488 | 4287 |
| 9.014 | 5.923 | 6.680 | 17072 | 4401 | 4336 | 5072 | 4832 |
| 8.017 | 5.423 | 6.099 | 15184 | 3901 | 3857 | 4491 | 4297 |
| 7.006 | 4.917 | 5.518 | 13269 | 3395 | 3370 | 3910 | 3755 |
| 6.032 | 4.433 | 4.956 | 11424 | 2911 | 2902 | 3348 | 3233 |
| 5.018 | 3.934 | 4.383 | 9504 | 2412 | 2414 | 2775 | 2690 |
| 4.018 | 3.441 | 3.816 | 7610 | 1919 | 1933 | 2208 | 2154 |
| 3.008 | 2.949 | 3.250 | 5697 | 1427 | 1447 | 1642 | 1612 |
| 2.004 | 2.462 | 2.687 | 3795 | 940 | 964 | 1079 | 1074 |
| 1.008 | 1.982 | 2.130 | 1909 | 460 | 485 | 522 | 540 |

CALIBRATION

Load: 1.0 VDC = 10,000 lbf

Strain: 1.0 VDC = 245.76 $\mu in/in$

Stress: 1.0 VDC = 1893.74 $lbf/in^2$

TABLE 9

Stress and strain data from strain gage placement tests and comparison of experimental and theoretical strains, maximum and average. Run number 2

| Load Volts | Average Strain Volts | Maximum Strain Volts | Stress $lbf/in^2$ | Exp. Avg. Strain μin/in | Theo. Avg Strain μin/in | Exp. Max. Strain μin/in | Theo. Max. Strain μin/in |
|---|---|---|---|---|---|---|---|
| 1.013 | 2.011 | 2.135 | 1919 | 477 | 487 | 564 | 543 |
| 1.998 | 2.486 | 2.691 | 3784 | 952 | 961 | 1120 | 1071 |
| 3.018 | 2.978 | 3.264 | 5716 | 1444 | 1452 | 1693 | 1618 |
| 4.010 | 3.464 | 3.823 | 7595 | 1930 | 1929 | 2252 | 2149 |
| 5.010 | 3.952 | 4.387 | 9489 | 2418 | 2410 | 2816 | 2685 |
| 6.015 | 4.444 | 4.953 | 11392 | 2910 | 2894 | 3382 | 3224 |
| 7.014 | 4.936 | 5.519 | 13284 | 3402 | 3374 | 3948 | 3760 |
| 8.040 | 5.445 | 6.099 | 15227 | 3911 | 3868 | 4528 | 4310 |
| 9.001 | 5.921 | 6.647 | 17047 | 4387 | 4330 | 5076 | 4825 |
| 8.023 | 5.435 | 6.083 | 15195 | 3901 | 3860 | 4512 | 4301 |
| 7.026 | 4.942 | 5.514 | 13307 | 3408 | 3380 | 3943 | 3766 |
| 6.015 | 4.442 | 4.936 | 11392 | 2908 | 2894 | 3365 | 3224 |
| 5.025 | 3.954 | 4.372 | 9517 | 2420 | 2417 | 2801 | 2694 |
| 4.015 | 3.458 | 3.802 | 7604 | 1924 | 1932 | 2231 | 2152 |
| 3.012 | 2.970 | 3.238 | 5705 | 1436 | 1449 | 1667 | 1615 |
| 2.008 | 2.483 | 2.677 | 3803 | 949 | 966 | 1106 | 1076 |
| 1.021 | 2.011 | 2.128 | 1934 | 477 | 491 | 557 | 547 |

CALIBRATION

Load: 1.0 VDC = 10,000 lbf

Strain: 1.0 VDC = 245.76 μin/in

Stress: 1.0 VDC = 1893.74 $lbf/in^2$

## TABLE 10

Stress and strain data from strain gage placement tests and comparison of experimental and theoretical strains, maximum and average. Run number 3

| Load Volts | Average Strain Volts | Maximum Strain Volts | Stress $lbf/in^2$ | Exp. Avg. Strain μin/in | Theo. Avg. Strain μin/in | Exp. Max. Strain μin/in | Theo. Max. Strain μin/in |
|---|---|---|---|---|---|---|---|
| 1.011 | 1.990 | 1.752 | 1915 | 474 | 486 | 559 | 542 |
| 2.015 | 2.474 | 2.317 | 3816 | 958 | 969 | 1124 | 1080 |
| 3.023 | 2.962 | 2.885 | 5725 | 1446 | 1454 | 1692 | 1620 |
| 4.035 | 3.455 | 3.455 | 7642 | 1939 | 1941 | 2262 | 2163 |
| 5.023 | 3.937 | 4.013 | 9513 | 2421 | 2416 | 2820 | 2692 |
| 6.033 | 4.433 | 4.583 | 11426 | 2917 | 2902 | 3390 | 3234 |
| 7.020 | 4.919 | 5.143 | 13295 | 3403 | 3377 | 3950 | 3763 |
| 8.017 | 5.412 | 5.708 | 15184 | 3896 | 3857 | 4515 | 4297 |
| 9.023 | 5.911 | 6.283 | 17089 | 4395 | 4341 | 5090 | 4837 |
| 8.032 | 5.418 | 5.707 | 15212 | 3902 | 3864 | 4514 | 4305 |
| 7.021 | 4.919 | 5.132 | 13297 | 3403 | 3378 | 3939 | 3763 |
| 6.017 | 4.423 | 4.557 | 11396 | 3907 | 2895 | 3364 | 3225 |
| 5.037 | 3.939 | 4.000 | 9540 | 2423 | 2423 | 2807 | 2670 |
| 4.014 | 3.439 | 3.424 | 7602 | 1923 | 1931 | 2231 | 2152 |
| 3.028 | 2.959 | 2.867 | 5735 | 1443 | 1457 | 1674 | 1623 |
| 2.006 | 2.465 | 2.297 | 3799 | 949 | 965 | 1104 | 1075 |
| 1.014 | 1.991 | 1.746 | 1920 | 475 | 488 | 553 | 544 |

CALIBRATION

Load: 1.0 VDC = 10,000 lbf

Strain: 1.0 VDC = 245.76 μin/in

Stress: 1.0 VDC = 1893.74 $lbf/in^2$

TABLE 11

Percent differences in experimental and theoretical strain valves from strain gage placement tests.

| Run No. 1 | | | Run No. 2 | | | Run No. 3 | | |
|---|---|---|---|---|---|---|---|---|
| Stress $lbf/in^2$ | Avg. Strain Diff. percent | Max. Strain Diff. percent | Stress $lbf/in^2$ | Avg. Strain Diff. percent | Max. Strain Diff. percent | Stress $lbf/in^2$ | Avg. Strain Diff. percent | Max. Strain Diff. percent |
| 1966 | 2.18 | 3.90 | 1919 | 2.12 | 3.87 | 1915 | 2.54 | 3.15 |
| 3788 | 0.54 | 4.19 | 3784 | 0.96 | 4.58 | 3816 | 1.17 | 4.07 |
| 5669 | 0.34 | 4.09 | 5716 | 0.54 | 4.65 | 5725 | 0.57 | 4.42 |
| 7616 | 0.13 | 4.16 | 7595 | 0.05 | 4.77 | 7642 | 0.11 | 4.58 |
| 9509 | 0.43 | 4.22 | 9489 | 0.32 | 4.86 | 9513 | 0.19 | 4.74 |
| 11384 | 0.73 | 4.38 | 11392 | 0.56 | 4.89 | 11426 | 0.50 | 4.83 |
| 13265 | 0.97 | 4.52 | 13284 | 0.82 | 5.01 | 13295 | 0.76 | 4.97 |
| 15148 | 1.23 | 4.69 | 15227 | 1.11 | 5.07 | 15184 | 1.02 | 5.07 |
| 17072 | 1.49 | 4.97 | 17047 | 1.31 | 5.21 | 17089 | 1.25 | 5.24 |
| 15184 | 1.15 | 4.51 | 15195 | 1.07 | 4.92 | 15212 | 0.98 | 4.85 |
| 13269 | 0.73 | 4.12 | 13307 | 0.83 | 4.70 | 13297 | 0.75 | 4.67 |
| 11424 | 0.31 | 3.55 | 11392 | 0.49 | 4.37 | 11396 | 0.43 | 4.30 |
| 9504 | 0.09 | 3.17 | 9517 | 0.11 | 3.99 | 9540 | 0.01 | 3.96 |
| 7610 | 0.72 | 2.52 | 7604 | 0.39 | 3.66 | 7602 | 0.42 | 3.69 |
| 5697 | 1.39 | 1.84 | 5705 | 0.90 | 3.25 | 5735 | 0.94 | 3.14 |
| 3795 | 2.50 | 0.45 | 3803 | 1.76 | 2.76 | 3799 | 1.66 | 2.67 |
| 1909 | 5.14 | 3.39 | 1934 | 2.89 | 1.78 | 1920 | 2.63 | 1.74 |

$$\text{Percent difference} = 1 - \frac{\text{avg x } 100}{\text{max}}$$

TABLE 12

Monotonic stress and strain data from single amplitude cyclic loading test on a plate.

| Nominal Load Volts | Nominal Stress Volts | Local Strain (1) Volts | Local Strain (2) Volts | Nominal Stress $lbf/in^2$ | Nominal Strain $\mu$in/in | Local Strain (1) $\mu$in/in | Local Strain (2) $\mu$in/in |
|---|---|---|---|---|---|---|---|
| 0.048 | 0.019 | 0.006 | 0.001 | 444 | 29 | 9 | 2 |
| 0.10 | 0.15 | 0.250 | 0.250 | 926 | 229 | 382 | 401 |
| 0.30 | 0.25 | 0.650 | 0.650 | 2778 | 381 | 994 | 1041 |
| 0.70 | 0.55 | 1.250 | 1.250 | 6481 | 839 | 1912 | 2003 |
| 1.40 | 0.85 | 2.05 | 1.95 | 12963 | 1296 | 3136 | 3124 |
| 2.00 | 1.20 | 3.00 | 2.85 | 18519 | 1830 | 4589 | 4566 |
| 2.40 | 1.45 | 3.75 | 3.65 | 22222 | 2211 | 5737 | 5848 |
| 3.00 | 1.75 | 4.65 | 4.55 | 27778 | 2668 | 7113 | 7290 |
| 3.40 | 1.95 | 5.00 | 5.00 | 31481 | 2973 | 7649 | 8011 |
| 3.40 | 2.10 | 5.45 | 5.50 | 31481 | 3202 | 8337 | 8812 |
| 3.60 | 2.10 | 5.85 | 5.85 | 33333 | 3202 | 8949 | 9373 |
| 3.60 | 2.15 | 6.15 | 6.25 | 33333 | 3278 | 9408 | 10014 |
| 3.80 | 2.25 | 6.50 | 6.55 | 35185 | 3431 | 9943 | 10495 |
| 3.80 | 2.25 | 6.75 | 6.85 | 35185 | 3431 | 10326 | 10975 |
| 4.00 | 2.25 | 6.95 | 7.05 | 37037 | 3431 | 10632 | 11296 |
| 4.00 | 2.35 | 7.15 | 7.25 | 37037 | 3583 | 10938 | 11616 |
| 4.00 | 2.35 | 7.25 | 7.35 | 37037 | 3583 | 11091 | 11776 |
| 4.00 | 2.35 | 7.35 | 7.45 | 37037 | 3583 | 11244 | 11937 |

CALIBRATION

Load: 1.0 VDC=10,000 lbf
Strain: 1.0 VDC=1532.23 $\mu$in/in
Stress: 1.0 VDC=9259.26 $lbf/in^2$

TABLE 13

Data for calculation of average stress concentration factors for monotonic local stress vs. nominal strain curves at single amplitude cyclic loading test on a plate.

Strain Gage (1)

| Nominal Stress $lbf/in^2$ | Nominal Strain $\mu in/in$ | Local Strain $\mu in/in$ | SAL Data Local Stress $lbf/in^2$ | SAL Data $K_t$ | Mono. Data Local Stress $lbf/in^2$ | Mono. Data $K_t$ |
|---|---|---|---|---|---|---|
| 444 | 29 | 9 | 0 | 0 | 0 | 0 |
| 926 | 229 | 382 | 4000 | 2.68 | 4000 | 2.68 |
| 2778 | 381 | 994 | 9750 | 3.03 | 10000 | 3.06 |
| 6481 | 839 | 1912 | 19000 | 2.58 | 20500 | 2.68 |
| 12936 | 1296 | 3136 | 31000 | 2.41 | 32750 | 2.47 |
| 18519 | 1830 | 4589 | 45000 | 2.47 | 49000 | 2.58 |
| 22222 | 2211 | 5737 | 56500 | 2.57 | 60000 | 2.65 |
| 27778 | 2668 | 7113 | 70500 | 2.60 | 74000 | 2.67 |
| 31481 | 2973 | 7649 | 74750 | 2.47 | 76500 | 2.50 |
| 31481 | 3202 | 8337 | 77500 | 2.53 | 78000 | 2.54 |
| 33333 | 3202 | 8949 | 79000 | 2.57 | 78000 | 2.56 |
| 33333 | 3278 | 9408 | 79500 | 2.62 | 78000 | 2.59 |
| 35185 | 3431 | 9943 | 80000 | 2.57 | 78000 | 2.53 |
| 35185 | 3431 | 10326 | 80000 | 2.62 | 78000 | 2.58 |
| 37037 | 3431 | 10632 | 80000 | 2.59 | 78000 | 2.55 |
| 37037 | 3583 | 10938 | 80000 | 2.57 | 78000 | 2.54 |
| 37037 | 3583 | 11091 | 80000 | 2.59 | 78000 | 2.55 |
| 37037 | 3583 | 11244 | 80000 | 2.60 | 78000 | 2.57 |
| | | AVERAGE $K_t$ | | 2.59 | | 2.61 |

Strain Gage (2)

| Nominal Stress $lbf/in^2$ | Nominal Strain $\mu in/in$ | Local Strain $\mu in/in$ | SAL Data Local Stress $lbf/in^2$ | SAL Data $K_t$ | Mono. Data Local Stress $lbf/in^2$ | Mono. Data $K_t$ |
|---|---|---|---|---|---|---|
| 444 | 29 | 2 | 0 | 0 | 0 | 0 |
| 926 | 229 | 401 | 4000 | 2.75 | 4000 | 2.75 |
| 2778 | 381 | 1041 | 10500 | 3.21 | 11000 | 3.29 |
| 6481 | 839 | 2003 | 19750 | 2.70 | 21000 | 2.78 |
| 12936 | 1296 | 3124 | 31000 | 2.40 | 33000 | 2.48 |
| 18519 | 1830 | 4566 | 44750 | 2.46 | 47750 | 2.54 |
| 22222 | 2211 | 5848 | 57750 | 2.62 | 61250 | 2.70 |
| 27778 | 2668 | 7290 | 72000 | 2.66 | 75000 | 2.72 |
| 31481 | 2973 | 8011 | 76500 | 2.56 | 77000 | 2.57 |
| 31481 | 3202 | 8812 | 79000 | 2.63 | 78000 | 2.61 |
| 33333 | 3202 | 9373 | 79500 | 2.64 | 78000 | 2.62 |
| 33333 | 3278 | 10014 | 80000 | 2.71 | 78000 | 2.67 |

Table 13 (Cont'd)

Strain Gage (2)

| Nominal Stress lbf/in$^2$ | Nominal Strain $\mu$in/in | Local Strain $\mu$in/in | SAL Data Local Stress lbf/in$^2$ | SAL Data $K_t$ | Mono. Data Local Stress lbf/in$^2$ | Mono. Data $K_t$ |
|---|---|---|---|---|---|---|
| 35185 | 3431 | 10495 | 80000 | 2.64 | 78000 | 2.60 |
| 35185 | 3431 | 10975 | 80000 | 2.70 | 78000 | 2.66 |
| 37037 | 3431 | 11296 | 80000 | 2.67 | 78000 | 2.63 |
| 37037 | 3583 | 11616 | 80000 | 2.65 | 78000 | 2.61 |
| 37037 | 3583 | 11776 | 80000 | 2.66 | 78000 | 2.63 |
| 37037 | 3583 | 11937 | 80000 | 2.68 | 78000 | 2.65 |
| | AVERAGE $K_t$ | | | 2.67 | | 2.68 |

SAL - Single amplitude cyclic loading test data

Mono. - Monotonic stress-strain data from cyclic stress-strain curve test.

TABLE 14

Stress and strain data for construction of local stress vs. nominal strain curves of single amplitude cyclic loading test on a plate and percent variation between stresses calculated from one strain gage and two data bases.

Single Amplitude Cyclic Leading Test Data

| Nominal Strain $\mu$in/in | Stress x Strain lbf/in$^2$ | Local Stress (1) lbf/in$^2$ | Stress x Strain | Local Stress (2) lbf/in$^2$ | Percent Variation Strain gage (1) |
|---|---|---|---|---|---|
| 29 | 0.0564 | 0 | 0.05995 | 0 | 0 |
| 229 | 3.518 | 4000 | 3.7385 | 4500 | 38.46 |
| 381 | 9.738 | 9250 | 10.348 | 9500 | 11.90 |
| 839 | 47.22 | 21500 | 50.182 | 22000 | 4.44 |
| 1296 | 112.67 | 33500 | 119.74 | 34500 | 4.29 |
| 1830 | 224.65 | 47500 | 238.74 | 48750 | 6.40 |
| 2211 | 327.93 | 57500 | 348.50 | 59250 | 7.63 |
| 2668 | 477.50 | 69250 | 507.45 | 71000 | 7.36 |
| 2973 | 592.91 | 76000 | 630.10 | 77500 | 2.56 |
| 3202 | 687.77 | 79000 | 730.91 | 79750 | 1.27 |
| 3202 | 687.77 | 79000 | 730.91 | 79750 | 1.27 |
| 3278 | 720.80 | 79500 | 766.02 | 80000 | 1.89 |
| 3431 | 789.66 | 80000 | 839.20 | 80000 | 2.50 |
| 3431 | 789.66 | 80000 | 839.20 | 80000 | 2.50 |
| 3431 | 789.66 | 80000 | 839.20 | 80000 | 2.50 |
| 3583 | 861.18 | 80000 | 915.20 | 80000 | 2.50 |
| 3583 | 861.18 | 80000 | 915.20 | 80000 | 2.50 |
| 3583 | 861.18 | 80000 | 915.20 | 80000 | 2.50 |
| | | | | | 6.03 |
| | | | | | (4.00) |

Table 14 (Cont'd)

Monotonic data from cyclic stress-strain curve test

| Nominal Strain $\mu$in/in | Stress x Strain lbf/in$^2$ | Local Stress (1) lbf/in$^2$ | Stress x Strain | Local Stress (2) lbf/in$^2$ | Percent Variation Strain gage (2) |
|---|---|---|---|---|---|
| 29 | 0.0611 | 0 | 0.0644 | 0 | 0 |
| 229 | 3.8117 | 6500 | 4.0189 | 7250 | 37.93 |
| 381 | 10.551 | 10500 | 11.125 | 11000 | 13.64 |
| 839 | 51.165 | 22500 | 53.946 | 23250 | 5.38 |
| 1296 | 122.08 | 35000 | 128.72 | 36000 | 4.17 |
| 1830 | 243.42 | 50750 | 256.65 | 52250 | 6.70 |
| 2211 | 355.32 | 62250 | 374.64 | 64000 | 7.42 |
| 2668 | 517.39 | 74750 | 545.51 | 76000 | 6.58 |
| 2973 | 642.44 | 78000 | 677.37 | 78000 | 0.64 |
| 3202 | 745.23 | 78000 | 785.74 | 78000 | 2.19 |
| 3202 | 745.23 | 78000 | 785.74 | 78000 | 2.19 |
| 3278 | 781.02 | 78000 | 823.48 | 78000 | 2.50 |
| 3431 | 855.63 | 78000 | 902.14 | 78000 | 2.50 |
| 3431 | 855.63 | 78000 | 902.14 | 78000 | 2.50 |
| 3431 | 855.63 | 78000 | 902.14 | 78000 | 2.50 |
| 3583 | 933.12 | 78000 | 983.85 | 78000 | 2.50 |
| 3583 | 933.12 | 78000 | 983.85 | 78000 | 2.50 |
| 3583 | 933.12 | 78000 | 983.85 | 78000 | 2.50 |
| | | | | | 6.14 |
| | | | | | (4.15) |

TABLE 15

Monotonic stress and strain data from dual amplitude cyclic loading test on a plate.

| Nominal Load Volts | Nominal Strain Volts | Local Strain (1) Volts | Local Strain (2) Volts | Nominal Stress $lbf/in^2$ | Nominal Strain $\mu in/in$ | Local Strain (1) $\mu in/in$ | Local Strain (2) $\mu in/in$ |
|---|---|---|---|---|---|---|---|
| 0.00 | 0.10 | 0.20 | 0.20 | 0 | 152 | 306 | 319 |
| 0.20 | 0.25 | 0.55 | 0.55 | 1852 | 381 | 841 | 878 |
| 0.70 | 0.40 | 1.15 | 1.15 | 6481 | 610 | 1759 | 1836 |
| 1.10 | 0.75 | 1.75 | 1.75 | 10185 | 1144 | 2677 | 2794 |
| 1.50 | 0.95 | 2.40 | 2.35 | 13889 | 1449 | 3671 | 3753 |
| 2.00 | 1.35 | 3.35 | 3.25 | 18519 | 2059 | 5125 | 5190 |
| 3.20 | 2.00 | 5.50 | 5.35 | 29630 | 3050 | 8414 | 8543 |
| 3.50 | 2.10 | 5.95 | 5.80 | 32407 | 3202 | 9102 | 9262 |
| 3.70 | 2.20 | 6.50 | 6.40 | 34259 | 3355 | 9943 | 10220 |
| 3.90 | 2.20 | 6.60 | 6.60 | 36111 | 3355 | 10096 | 10539 |
| 3.90 | 2.20 | 6.70 | 6.60 | 36111 | 3355 | 10249 | 10539 |
| 3.90 | 2.20 | 6.80 | 6.70 | 36111 | 3355 | 10402 | 10699 |
| 3.90 | 2.35 | 7.00 | 6.95 | 36111 | 3583 | 10708 | 11098 |
| 3.90 | 2.30 | 7.10 | 7.00 | 36111 | 3507 | 10861 | 11178 |
| 3.90 | 2.35 | 7.15 | 7.10 | 36111 | 3583 | 10938 | 11337 |
| 3.90 | 2.30 | 7.15 | 7.10 | 36111 | 3507 | 10938 | 11337 |

CALIBRATION

Load: 1.0 VDC= 10,000 lbf
Strain: 1.0 VDC= 1532.23 $\mu in/in$
Stress: 1.0 VDC= 9259.26 $lbf/in^2$

TABLE 16

Data for calculation of average stress concentration factors for monotonic local stress vs. nominal strain curves of dual amplitude cyclic loading test on a plate.

Strain Gage (1)

| Nominal Stress $lbf/in^2$ | Nominal Strain $\mu in/in$ | Local Strain $\mu in/in$ | DAL Data Local Stress $lbf/in^2$ | DAL Data $K_t$ | Mono. Data Local Stress $lbf/in^2$ | Mono. Data $K_t$ |
|---|---|---|---|---|---|---|
| 0 | 152 | 306 | 3500 | 0 | 3250 | 0 |
| 1852 | 381 | 841 | 9000 | 3.28 | 8750 | 3.23 |
| 6481 | 610 | 1759 | 13000 | 2.83 | 13500 | 2.87 |
| 10185 | 1144 | 2677 | 27500 | 2.51 | 23000 | 2.54 |
| 13889 | 1449 | 3671 | 37500 | 2.62 | 38500 | 2.65 |
| 18519 | 2059 | 5125 | 52500 | 2.66 | 53750 | 2.69 |
| 29630 | 3050 | 8414 | 75500 | 2.65 | 77750 | 2.69 |
| 32407 | 3202 | 9102 | 77750 | 2.61 | 78000 | 2.62 |
| 34259 | 3355 | 9943 | 78000 | 2.60 | 78000 | 2.60 |
| 36111 | 3355 | 10096 | 78000 | 2.55 | 78000 | 2.55 |
| 36111 | 3355 | 10249 | 78000 | 2.57 | 78000 | 2.57 |
| 36111 | 3355 | 10402 | 78000 | 2.59 | 78000 | 2.59 |
| 36111 | 3583 | 10708 | 78000 | 2.54 | 78000 | 2.54 |
| 36111 | 3507 | 10861 | 78000 | 2.59 | 78000 | 2.59 |
| 36111 | 3583 | 10938 | 78000 | 2.57 | 78000 | 2.57 |
| 36111 | 3507 | 10938 | 78000 | 2.60 | 78000 | 2.60 |
| | | AVERAGE $K_t$ | | 2.61 | | 2.62 |

Strain Gage (2)

| Nominal Stress $lbf/in^2$ | Nominal Strain $\mu in/in$ | Local Strain $\mu in/in$ | DAL Data Local Stress $lbf/in^2$ | DAL Data $K_t$ | Mono. Data Local Stress $lbf/in^2$ | Mono. Data $K_t$ |
|---|---|---|---|---|---|---|
| 0 | 152 | 319 | 3500 | 0 | 3250 | 0 |
| 1852 | 381 | 878 | 9250 | 3.39 | 9250 | 3.39 |
| 6481 | 610 | 1836 | 19000 | 2.97 | 19250 | 2.99 |
| 10185 | 1144 | 2794 | 28500 | 2.61 | 29000 | 2.64 |
| 13889 | 1449 | 3753 | 38500 | 2.68 | 39500 | 2.71 |
| 18519 | 2059 | 5190 | 52750 | 2.68 | 54250 | 2.72 |
| 29630 | 3050 | 8543 | 76000 | 2.68 | 78000 | 2.72 |
| 32407 | 3202 | 9262 | 78000 | 2.64 | 78000 | 2.64 |
| 34259 | 3355 | 10220 | 78000 | 2.63 | 78000 | 2.63 |
| 36111 | 3355 | 10539 | 78000 | 2.60 | 78000 | 2.60 |
| 36111 | 3355 | 10539 | 78000 | 2.60 | 78000 | 2.60 |
| 36111 | 3355 | 10699 | 78000 | 2.62 | 78000 | 2.62 |
| 36111 | 3583 | 11098 | 78000 | 2.59 | 78000 | 2.59 |
| 36111 | 3507 | 11178 | 78000 | 2.62 | 78000 | 2.62 |
| 36111 | 3583 | 11337 | 78000 | 2.61 | 78000 | 2.61 |
| 36111 | 3507 | 11337 | 78000 | 2.64 | 78000 | 2.64 |
| | | AVERAGE $K_t$ | | 2.66 | | 2.67 |

DAL - Dual amplitude cyclic loading test data

Mono. - Monotonic stress-strain data from cyclic stress-strain curve test.

TABLE 17

Stress and strain data for construction of local stress vs. nominal strain curves of dual amplitude cyclic loading test on a plate and percent variation between stresses calculated from one strain gage and two data bases.

Dual amplitude cyclic loading test data

| Nominal Strain $\mu$in/in | Stress x Strain lbf/in$^2$ | Local Stress (1) lbf/in | Stress x Strain lbf/in$^2$ | Local Stress (2) lbf/in$^2$ | Percent Variation Strain gage (1) |
|---|---|---|---|---|---|
| 381 | 10.076 | 9500 | 10.466 | 9750 | 11.63 |
| 610 | 25.829 | 14750 | 26.829 | 14750 | 11.94 |
| 1144 | 90.846 | 29750 | 94.360 | 30250 | 3.25 |
| 1449 | 145.74 | 35750 | 151.38 | 36500 | 9.49 |
| 2059 | 294.28 | 55000 | 305.67 | 55000 | 4.76 |
| 3050 | 635.74 | 76000 | 670.71 | 76750 | 2.56 |
| 3202 | 711.70 | 77500 | 739.23 | 78000 | 0.64 |
| 3355 | 781.34 | 78000 | 811.56 | 78000 | 6.32 |
| 3355 | 781.34 | 78000 | 811.56 | 78000 | |
| 3355 | 781.34 | 78000 | 811.56 | 78000 | |
| 3355 | 781.34 | 78000 | 811.56 | 78000 | |
| 3583 | 891.15 | 78000 | 925.62 | 78000 | |
| 3507 | 853.74 | 78000 | 886.77 | 78000 | |
| 3583 | 891.15 | 78000 | 925.62 | 78000 | |
| 3507 | 853.74 | 78000 | 886.77 | 78000 | |

Table 17 (Cont'd)

Monotonic data from cyclic stress-strain curve test

| Nominal Strain $\mu$in/in | Stress x Strain lbf/in$^2$ | Local Stress (1) lbf/in | Stress x Strain lbf/in$^2$ | Local Stress (2) lbf/in$^2$ | Percent Variation Strain gage (2) |
|---|---|---|---|---|---|
| 381 | 10.632 | 10750 | 11.042 | 11000 | 11.36 |
| 610 | 27.254 | 16750 | 28.304 | 17000 | 13.24 |
| 1144 | 95.856 | 30750 | 99.549 | 31500 | 3.97 |
| 1449 | 153.78 | 39500 | 159.71 | 40250 | 9.32 |
| 2059 | 310.51 | 57750 | 322.48 | 59000 | 6.78 |
| 3050 | 681.34 | 78000 | 707.60 | 78000 | 1.60 |
| 3202 | 750.95 | 78000 | 779.88 | 78000 | 7.71 |
| 3355 | 824.43 | 78000 | 856.19 | 78000 | |
| 3355 | 824.43 | 78000 | 856.19 | 78000 | |
| 3355 | 824.43 | 78000 | 856.19 | 78000 | |
| 3355 | 824.43 | 78000 | 856.19 | 78000 | |
| 3583 | 940.29 | 78000 | 976.52 | 78000 | |
| 3507 | 900.82 | 78000 | 935.53 | 78000 | |
| 3583 | 940.29 | 78000 | 976.52 | 78000 | |
| 3507 | 900.82 | 78000 | 935.53 | 78000 | |

TABLE 18

Stress and strain data from unloading portion of initial loading cycle from single amplitude cyclic loading test on a plate.

| Nominal Load Volts | Local Strain (1) Volts | Local Strain (2) Volts | Nominal Stress $lbf/in^2$ | Local Strain (1) $\mu in/in$ | Local Strain (2) $\mu in/in$ |
|---|---|---|---|---|---|
| 4.0 | 7.35 | 7.45 | 37037 | 11244 | 11937 |
| 4.0 | 7.25 | 7.35 | 37037 | 11091 | 11776 |
| 3.8 | 7.00 | 7.05 | 35185 | 10708 | 11296 |
| 3.4 | 6.45 | 6.60 | 31481 | 9867 | 10575 |
| 3.0 | 5.75 | 6.00 | 27778 | 8796 | 9613 |
| 2.6 | 5.00 | 5.20 | 24074 | 7649 | 8332 |
| 2.0 | 4.05 | 4.25 | 18519 | 6196 | 6809 |
| 1.4 | 3.20 | 3.40 | 12963 | 4895 | 5448 |
| 0.8 | 2.35 | 2.50 | 7407 | 3595 | 4006 |
| 0.4 | 1.55 | 1.80 | 3704 | 2371 | 2884 |
| 0.0 | 0.90 | 1.10 | 0 | 1377 | 1762 |

CALIBRATION

Load: 1.0 VDC= 10,000 lbf
Strain: 1.0 VDC= 1532.23 $\mu in/in$
Stress: 1.0 VDC= 9259.26 $lbf/in^2$

TABLE 19

Data for calculation of average stress concentration factors for stress relaxation behavior study of single amplitude cyclic loading test on a plate.

| Nominal Stress $lbf/in^2$ | Local Strain (1) $\mu$in/in | Local Strain (2) $\mu$in/in | $K_t$ (1) Mono. Data | $K_t$ (1) SAL Data | $K_t$ (2) Mono. Data | $K_t$ (2) SAL Data |
|---|---|---|---|---|---|---|
| 37037 | 11244 | 11937 | 0 | 0 | 0 | 0 |
| 37037 | 11091 | 11776 | 0 | 0 | 0 | 0 |
| 35185 | 10708 | 11296 | 3.09 | 2.89 | 3.69 | 3.46 |
| 31481 | 9867 | 10575 | 2.64 | 2.48 | 2.62 | 2.45 |
| 27778 | 8796 | 9316 | 2.82 | 2.64 | 2.68 | 2.51 |
| 24074 | 7649 | 8332 | 2.96 | 2.77 | 2.97 | 2.78 |
| 18519 | 6196 | 6809 | 2.91 | 2.73 | 2.95 | 2.77 |
| 12963 | 4895 | 5448 | 2.81 | 2.64 | 2.88 | 2.70 |
| 7407 | 3595 | 4006 | 2.75 | 2.44 | 2.86 | 2.68 |
| 3704 | 2371 | 2884 | 2.84 | 2.66 | 2.90 | 2.72 |
| 0 | 1377 | 1762 | 2.84 | 2.66 | 2.93 | 2.75 |
| | | AVERAGE $K_t$ | 2.85 | 2.66 | 2.94 | 2.76 |

SAL - Single amplitude cyclic loading test data. .
Mono. Monotonic data from cyclic stress-strain curve test.

TABLE 20

Data from alternate calculation of average stress concentration factors for unloading portion of initial loading cycle in single amplitude cyclic loading test on a plate.

| Nominal Stress $lbf/in^2$ | Local Strain (1) $\mu in/in$ | Local Stress (2) $\mu in/in$ | Mono. Data $K_t(1)$ | Mono. Data $K_t(2)$ | SAL Data $K_t(1)$ | SAL Data $K_t(2)$ |
|---|---|---|---|---|---|---|
| 37037 | 11244 | 11937 | 2.84 | 2.93 | 2.66 | 2.75 |
| 37037 | 11091 | 11776 | 2.89 | 2.88 | 2.62 | 2.70 |
| 35185 | 10708 | 11296 | 2.83 | 2.89 | 2.65 | 2.71 |
| 31481 | 9867 | 10575 | 2.88 | 2.99 | 2.70 | 2.80 |
| 27778 | 8796 | 9613 | 2.85 | 3.02 | 2.67 | 2.83 |
| 24074 | 7649 | 8332 | 2.78 | 2.91 | 2.61 | 2.73 |
| 18519 | 6196 | 6809 | 2.78 | 2.91 | 2.60 | 2.73 |
| 12963 | 4895 | 5448 | 2.90 | 3.03 | 2.71 | 2.84 |
| 7407 | 3595 | 4006 | 3.20 | 3.23 | 2.99 | 3.03 |
| 3704 | 2371 | 2884 | 2.86 | 3.23 | 2.68 | 3.03 |
| 0 | 1377 | 1762 | 0 | 0 | 0 | 0 |
| | | | 2.88 | 3.00 | 2.69 | 2.82 |

Mono. - Monotonic data from cyclid stress-strain curve test
$\sigma_m$ = 78,000 $lbf/in^2$ E= 10.67 x $10^6$ $lbf/in^2$

SAL - Single amplitude cyclic loading test data
$\sigma_m$ = 80,000 $lbf/in^2$ E= 10.0 x $10^6$ $lbf/in^2$

TABLE 21

Stress, strain, and cycle number data from single amplitude cyclic loading test on a plate.

| Cycle | Nominal Load Volts | Nominal Strain Volts | Local Strain(1) Volts | Local Strain(2) Volts | Nominal Stress $lbf/in^2$ | Nominal Strain $\mu in/in$ | Local Strain(1) $\mu in/in$ | Local Strain(2) $\mu in/in$ |
|---|---|---|---|---|---|---|---|---|
| 1 | 4.0 | 2.40 | 7.35 | 7.45 | 37037 | 3660 | 11244 | 11937 |
| 2 | 4.0 | 2.40 | 7.35 | 7.45 | 37037 | 3660 | 11244 | 11937 |
| 3 | 4.0 | 2.40 | 7.35 | 7.45 | 37037 | 3660 | 11244 | 11937 |
| 4 | 4.0 | 2.30 | 7.40 | 7.50 | 37037 | 3507 | 11320 | 12017 |
| 5 | 4.0 | 2.30 | 7.40 | 7.50 | 37037 | 3507 | 11320 | 12017 |
| 6 | 3.9 | 2.30 | 7.40 | 7.50 | 36111 | 3507 | 11320 | 12017 |
| 7 | 3.9 | 2.30 | 7.40 | 7.45 | 36111 | 3507 | 11320 | 11937 |
| 8 | 3.9 | 2.30 | 7.40 | 7.50 | 36111 | 3507 | 11320 | 12017 |
| 9 | 3.9 | 2.30 | 7.40 | 7.50 | 36111 | 3507 | 11320 | 12017 |
| 10 | 3.9 | 2.30 | 7.40 | 7.50 | 36111 | 3507 | 11320 | 12017 |
| 11 | 3.9 | 2.30 | 7.40 | 7.50 | 36111 | 3507 | 11320 | 12017 |
| 12 | 3.9 | 2.30 | 7.40 | 7.50 | 36111 | 3507 | 11320 | 12017 |
| 13 | 3.9 | 2.30 | 7.40 | 7.50 | 36111 | 3507 | 11320 | 12017 |
| 14 | 3.9 | 2.30 | 7.40 | 7.50 | 36111 | 3507 | 11320 | 12017 |
| 15 | 3.9 | 2.30 | 7.40 | 7.50 | 36111 | 3507 | 11320 | 12017 |
| 16 | 3.8 | 2.30 | 7.40 | 7.50 | 35185 | 3507 | 11320 | 12017 |
| 17 | 3.8 | 2.30 | 7.40 | 7.50 | 35185 | 3507 | 11320 | 12017 |
| 18 | 3.9 | 2.30 | 7.40 | 7.50 | 36111 | 3507 | 11320 | 12017 |
| 19 | 3.9 | 2.30 | 7.40 | 7.50 | 36111 | 3507 | 11320 | 12017 |
| 20 | 3.8 | 2.20 | 7.40 | 7.50 | 35185 | 3355 | 11320 | 12017 |
| 21 | 3.8 | 2.20 | 7.40 | 7.50 | 35185 | 3355 | 11320 | 12017 |
| 22 | 3.8 | 2.20 | 7.40 | 7.50 | 35185 | 3355 | 11320 | 12017 |
| 23 | 3.8 | 2.20 | 7.35 | 7.45 | 35185 | 3355 | 11244 | 11937 |

Table 21 (Cont'd)

| Cycle | Nominal Load Volts | Nominal Strain Volts | Local Strain(1) Volts | Local Strain(2) Volts | Nominal Stress $lbf/in^2$ | Nominal Strain $\mu in/in$ | Local Strain(1) $\mu in/in$ | Local Strain(2) $\mu in/in$ |
|---|---|---|---|---|---|---|---|---|
| 24 | 3.8 | 2.20 | 7.35 | 7.45 | 35185 | 3355 | 11244 | 11937 |
| 25 | 3.7 | 2.20 | 7.35 | 7.45 | 34259 | 3355 | 11244 | 11937 |
| 26 | 3.6 | 2.20 | 7.35 | 7.45 | 34259 | 3355 | 11244 | 11937 |
| 27 | 3.6 | 2.15 | 7.35 | 7.45 | 34259 | 3278 | 11244 | 11937 |
| 28 | 3.6 | 2.15 | 7.35 | 7.45 | 34259 | 3278 | 11244 | 11937 |
| 29 | 3.6 | 2.15 | 7.35 | 7.45 | 34259 | 3278 | 11244 | 11937 |
| 30 | 3.6 | 2.15 | 7.35 | 7.45 | 34259 | 3278 | 11244 | 11937 |
| 31 | 3.6 | 2.15 | 7.35 | 7.45 | 34259 | 3278 | 11244 | 11937 |
| 32 | 3.5 | 2.15 | 7.35 | 7.45 | 32407 | 3278 | 11244 | 11937 |
| 33 | 3.6 | 2.15 | 7.35 | 7.45 | 34259 | 3278 | 11244 | 11937 |
| 34 | 3.6 | 2.15 | 7.35 | 7.35 | 34259 | 3278 | 11244 | 11937 |
| 35 | 3.5 | 2.15 | 7.35 | 7.45 | 32407 | 3278 | 11244 | 11937 |
| 36 | 3.5 | 2.05 | 7.35 | 7.45 | 32407 | 3126 | 11244 | 11937 |
| 37 | 3.5 | 2.10 | 7.35 | 7.45 | 32407 | 3202 | 11244 | 11937 |
| 38 | 3.5 | 2.10 | 7.35 | 7.45 | 32407 | 3202 | 11244 | 11937 |
| 39 | 3.6 | 2.10 | 7.35 | 7.45 | 34259 | 3202 | 11244 | 11937 |
| 40 | 3.5 | 2.10 | 7.35 | 7.45 | 32407 | 3202 | 11244 | 11937 |
| 41 | 3.6 | 2.10 | 7.40 | 7.45 | 34259 | 3202 | 11320 | 11937 |
| 42 | 3.5 | 2.05 | 7.35 | 7.45 | 32407 | 3126 | 11244 | 11937 |
| 43 | 3.6 | 2.10 | 7.35 | 7.35 | 34259 | 3202 | 11244 | 11937 |
| 44 | 3.4 | 2.10 | 7.35 | 7.45 | 31481 | 3202 | 11244 | 11937 |
| 45 | 3.4 | 2.10 | 7.35 | 7.45 | 31481 | 3202 | 11244 | 11937 |
| 46 | 3.4 | 2.10 | 7.35 | 7.50 | 31481 | 3202 | 11244 | 12017 |
| 47 | 3.4 | 2.10 | 7.35 | 7.50 | 31481 | 3202 | 11244 | 12017 |
| 48 | 3.4 | 2.10 | 7.40 | 7.50 | 31481 | 3202 | 11320 | 12017 |

Table 21 (Cont'd)

| Cycle | Nominal Load Volts | Nominal Strain Volts | Local Strain(1) Volts | Local Strain(2) Volts | Nominal Stress $lbf/in^2$ | Nominal Strain $\mu in/in$ | Local Strain(1) $\mu in/in$ | Local Strain(2) $\mu in/in$ |
|---|---|---|---|---|---|---|---|---|
| 49 | 3.4 | 2.10 | 7.40 | 7.50 | 31481 | 3202 | 11320 | 12017 |
| 50 | 3.4 | 2.10 | 7.40 | 7.50 | 31481 | 3202 | 11320 | 12017 |
| 51 | 3.4 | 2.10 | 7.40 | 7.50 | 31481 | 3202 | 11320 | 12017 |
| 52 | 3.4 | 2.10 | 7.40 | 7.50 | 31481 | 3202 | 11320 | 12017 |
| 53 | 3.4 | 2.10 | 7.40 | 7.50 | 31481 | 3202 | 11320 | 12017 |
| 54 | 3.4 | 2.00 | 7.40 | 7.50 | 31481 | 3050 | 11320 | 12017 |
| 55 | 3.4 | 2.00 | 7.40 | 7.50 | 31481 | 3050 | 11320 | 12017 |
| 56 | 3.5 | 2.00 | 7.40 | 7.50 | 32407 | 3050 | 11320 | 12017 |
| 57 | 3.5 | 2.00 | 7.40 | 7.50 | 32407 | 3050 | 11320 | 12017 |
| 58 | 3.5 | 2.00 | 7.40 | 7.50 | 32407 | 3050 | 11320 | 12017 |
| 59 | 3.5 | 2.00 | 7.40 | 7.50 | 32407 | 3050 | 11320 | 12017 |
| 60 | 3.4 | 2.00 | 7.40 | 7.50 | 21481 | 3050 | 11320 | 12017 |
| 61 | 3.4 | 2.00 | 7.40 | 7.50 | 31481 | 3050 | 11320 | 12017 |
| 62 | 3.5 | 2.00 | 7.40 | 7.50 | 32407 | 3050 | 11320 | 12017 |
| 63 | 3.4 | 2.00 | 7.40 | 7150 | 31481 | 3050 | 11320 | 12017 |
| 64 | 3.5 | 2.00 | 7.40 | 7.50 | 32407 | 3050 | 11320 | 12017 |
| 65 | 3.5 | 2.00 | 7.40 | 7.50 | 32407 | 3050 | 11320 | 12017 |
| 66 | 3.3 | 2.00 | 7.40 | 7.50 | 30556 | 3050 | 11320 | 10217 |
| 67 | 3.3 | 2.00 | 7.40 | 7.50 | 30556 | 3050 | 11320 | 12017 |
| 68 | 3.3 | 2.00 | 7.40 | 7.50 | 30556 | 3050 | 11320 | 12017 |
| 69 | 3.2 | 2.00 | 7.40 | 7.50 | 29630 | 3050 | 11320 | 12017 |
| 70 | 3.3 | 2.00 | 7.40 | 7.50 | 30556 | 3050 | 11320 | 12017 |
| 71 | 3.3 | 2.00 | 7.40 | 7.55 | 30556 | 3050 | 11320 | 12097 |
| 72 | 3.3 | 2.00 | 7.40 | 7.55 | 30556 | 3050 | 11320 | 12097 |
| 73 | 3.3 | 2.00 | 7.40 | 7.55 | 30556 | 3050 | 11320 | 12097 |

Table 21 (Cont'd)

| Cycle | Nominal Load Volts | Nominal Strain Volts | Local Strain(1) Volts | Local Strain(2) Volts | Nominal Stress $lbf/in^2$ | Nominal Strain $\mu in/in$ | Local Strain(1) $\mu in/in$ | Local Strain(2) $\mu in/in$ |
|---|---|---|---|---|---|---|---|---|
| 74 | 3.3 | 2.00 | 7.40 | 7.50 | 30556 | 3050 | 11320 | 12017 |
| 75 | 3.3 | 1.95 | 7.40 | 7.50 | 30556 | 2973 | 11320 | 12017 |
| 76 | 3.3 | 1.95 | 7.40 | 7.50 | 30556 | 2973 | 11320 | 12017 |
| 77 | 3.2 | 1.90 | 7.40 | 7.50 | 29630 | 2897 | 11320 | 12017 |
| 78 | 3.3 | 1.95 | 7.40 | 7.50 | 30556 | 2973 | 11320 | 12017 |
| 79 | 3.2 | 1.90 | 7.40 | 7.50 | 29630 | 2897 | 11320 | 12017 |
| 80 | 3.2 | 1.95 | 7.40 | 7.50 | 29630 | 2973 | 11320 | 12017 |
| 81 | 3.2 | 1.95 | 7.40 | 7.50 | 29630 | 2973 | 11320 | 12017 |
| 82 | 3.3 | 1.95 | 7.40 | 7.50 | 30556 | 2973 | 11320 | 12017 |
| 83 | 3.3 | 1.95 | 7.40 | 7.50 | 30556 | 2973 | 11320 | 12017 |
| 84 | 3.2 | 1.95 | 7.40 | 7.50 | 29630 | 2973 | 11320 | 12017 |
| 85 | 3.2 | 1.90 | 7.40 | 7.50 | 29630 | 2897 | 11320 | 12017 |
| 86 | 3.2 | 1.95 | 7.40 | 7.50 | 29630 | 2973 | 11320 | 12017 |
| 87 | 3.2 | 1.95 | 7.40 | 7.50 | 29630 | 2973 | 11320 | 12017 |
| 88 | 3.2 | 1.90 | 7.40 | 7.50 | 29630 | 2897 | 11320 | 12017 |
| 89 | 3.3 | 1.95 | 7.40 | 7.45 | 30556 | 2973 | 11320 | 11937 |
| 90 | 3.2 | 1.90 | 7.40 | 7.40 | 29630 | 2897 | 11320 | 11857 |

CALIBRATION

Load: 1.0 VDC= 10,000 lbf

Strain: Strain Gage (1) 1.0 VDC= 1529.76 $\mu in/in$

Strain Gage (2) 1.0 VDC= 1602.23 $\mu in/in$

Nominal Strain Gage 1.0 VDC= 1524.83 $\mu in/in$

Stress: 1.0 VDC= 9559.26 $lbf/in^2$

TABLE 22

Data for calculation of local stress for stress relaxation behavior of single amplitude cyclic loading test on a plate.

| | | Mono. Data | | SAL Data | |
|---|---|---|---|---|---|
| Cycle | Nominal Stress $lbf/in^2$ | Local Stress(1) $lbf/in^2$ | Local Stress(2) $lbf/in^2$ | Local Stress(1) $lbf/in^2$ | Local Stress(2) $lbf/in^2$ |
| 1 | 37037 | 80000 | 80000 | 80000 | 80000 |
| 5 | 37037 | 80000 | 80000 | 80000 | 80000 |
| 10 | 36111 | 77361 | 77268 | 77537 | 77444 |
| 15 | 36111 | 77361 | 77268 | 77537 | 77444 |
| 20 | 35185 | 74722 | 74537 | 75074 | 74888 |
| 25 | 34259 | 72083 | 71805 | 72611 | 72333 |
| 30 | 34259 | 72083 | 71805 | 72611 | 72333 |
| 35 | 32407 | 66805 | 66342 | 67684 | 67221 |
| 40 | 32407 | 66805 | 66342 | 67684 | 67221 |
| 45 | 31481 | 64165 | 63610 | 65221 | 64665 |
| 50 | 31481 | 64165 | 63610 | 65221 | 64665 |
| 55 | 31481 | 64165 | 63610 | 65221 | 64665 |
| 60 | 31481 | 64165 | 63610 | 65221 | 64665 |
| 65 | 32407 | 66805 | 66342 | 67684 | 67221 |
| 70 | 30556 | 61529 | 60881 | 62761 | 62112 |
| 75 | 30556 | 61529 | 60881 | 62761 | 62112 |
| 80 | 29630 | 58890 | 58149 | 60297 | 59557 |
| 85 | 29630 | 58890 | 58149 | 60297 | 59557 |
| 90 | 29630 | 58890 | 58149 | 60297 | 59557 |

$Sm = 37037\ lbf/in^2$
$\sigma m = 80{,}000\ lbf/in^2$
Mono.-Monotonic data from cyclic stress-strain curve test
$E = 10.67 \times 10^6\ lbf/in^2$
SAL - Single amplitude cyclic loading test data
$E = 10.0 \times 10^6\ lbf/in^2$

TABLE 23

Stress and strain data from unloading portion of initial loading cycle from dual amplitude cyclic loading test on a plate.

| Nominal Load Volts | Local Strain(1) Volts | Local Strain(2) Volts | Nominal Stress $lbf/in^2$ | Local Strain(1) $\mu in/in$ | Strain(2) $\mu in/in$ |
|---|---|---|---|---|---|
| 3.8 | 7.10 | 7.10 | 35185 | 10861 | 11337 |
| 3.4 | 6.30 | 6.30 | 31481 | 9637 | 10060 |
| 2.3 | 4.50 | 4.50 | 21296 | 6884 | 7186 |
| 1.1 | 2.30 | 2.50 | 10185 | 3518 | 3992 |
| 0.10 | 1.00 | 1.10 | 926 | 1530 | 1757 |
| 0.0 | 0.90 | 0.95 | 0 | 1377 | 1517 |
| -0.60 | 0.0 | 0.20 | -5556 | 0 | 319 |

CALIBRATION

Load: 1.0 VDC= 10,000 lbf
Strain: 1.0 VDC= 1523.23 $\mu$ in/in
Stress: 1.0 VDC= 9259.26 $lbf/in^2$

TABLE 24

Data for calculation of average stress concentration factors for stress relaxation behavior study of dual amplitude cyclic loading test on a plate.

| Nominal Stress $lbf/in^2$ | Local Strain (1) $\mu in/in$ | Local Strain (2) $\mu in/in$ | $K_t(1)$ Mono. Data | $K_t(1)$ DAL Data | $K_t(2)$ Mono. Data | $K_t(2)$ DAL Data |
|---|---|---|---|---|---|---|
| 35185 | 10861 | 11337 | 0 | 0 | 0 | 0 |
| 31481 | 9637 | 10060 | 3.53 | 3.37 | 3.68 | 3.51 |
| 21296 | 6884 | 7186 | 3.06 | 2.92 | 3.19 | 3.05 |
| 10185 | 3518 | 3992 | 3.13 | 2.99 | 3.13 | 2.99 |
| 926 | 1530 | 1757 | 2.91 | 2.78 | 2.98 | 2.85 |
| 0 | 1377 | 1517 | 0 | 0 | 0 | 0 |
| -5556 | 0 | 319 | 2.84 | 2.72 | 2.89 | 2.76 |
| | | AVERAGE $K_t$ | 3.09 | 2.95 | 3.17 | 3.03 |

DAL - Dual amplitude cyclic loading test data.

Mono. - Monotonic data from cyclic stress-strain curve test.

TABLE 25

Data from alternate calculation of average stress concentration factors for unloading portion of initial loading cycle in dual amplitude cyclic loading test on a plate.

| Nominal Stress $lbf/in^2$ | Local Strain (1) $\mu in/in$ | Local Strain (2) $\mu in/in$ | Mono. Data $K_t(1)$ | Mono. Data $K_t(2)$ | DAL Data $K_t(1)$ | DAL Data $K_t(2)$ |
|---|---|---|---|---|---|---|
| 35185 | 10861 | 11337 | 2.88 | 2.84 | 2.75 | 2.84 |
| 31481 | 9637 | 10060 | 2.80 | 2.77 | 2.67 | 2.77 |
| 21296 | 6884 | 7186 | 2.76 | 2.71 | 2.64 | 2.71 |
| 10185 | 3518 | 3992 | 2.24 | 2.48 | 2.14 | 2.48 |
| 926 | 1530 | 1757 | 1.76 | 2.64 | 1.68 | 2.64 |
| 0 | 1377 | 1517 | 0 | 0 | 0 | 0 |
| -5556 | 0 | 319 | $\frac{2.64}{2.51}$ | $\frac{2.20}{2.61}$ | $\frac{2.53}{2.40}$ | $\frac{2.20}{2.61}$ |

Mono.-Monotonic data from cyclic stress-strain curve test.
$\sigma_m = 78{,}000\ lbf/in^2$  $E = 10.67 \times 10^6\ lbf/in^2$

DAL - Dual Amplitude cyclic loading test data.
$\sigma_m = 78{,}000\ lbf/in^2$  $E = 10.19 \times 10^6\ lbf/in^2$

TABLE 26

Stress, strain, and cycle number data from dual amplitude cyclic loading test on a plate.

| Cycle | Nominal Load Volts | Nominal Strain Volts | Local Strain (1) Volts | Local Strain (2) Volts | Nominal Stress $lbf/in^2$ | Nominal Strain $\mu in/in$ | Local Strain (1) $\mu in/in$ | Local Strain (2) $\mu in/in$ |
|---|---|---|---|---|---|---|---|---|
| 1 | 3.9 | 2.25 | 7.05 | 7.05 | 36111 | 3431 | 10785 | 11258 |
| 2 | 2.5 | 1.45 | 4.85 | 4.85 | 23148 | 2211 | 7419 | 7745 |
| 3 | 3.9 | 2.30 | 7.15 | 7.05 | 36111 | 3507 | 10938 | 11258 |
| 4 | 2.5 | 1.35 | 4.85 | 4.85 | 23148 | 2059 | 7419 | 7745 |
| 5 | 3.9 | 2.15 | 7.15 | 7.25 | 36111 | 3278 | 10938 | 11258 |
| 6 | 2.5 | 1.35 | 4.85 | 4.85 | 23148 | 2059 | 7419 | 7745 |
| 7 | 3.9 | 2.15 | 7.15 | 7.05 | 36111 | 3278 | 10938 | 11258 |
| 8 | 2.5 | 1.35 | 4.85 | 4.85 | 23148 | 2059 | 7419 | 7745 |
| 9 | 3.9 | 2.15 | 7.15 | 7.05 | 36111 | 3278 | 10938 | 11258 |
| 10 | 2.5 | 1.35 | 4.85 | 4.85 | 23148 | 2059 | 7419 | 7745 |
| 11 | 3.9 | 2.15 | 7.15 | 7.05 | 36111 | 3278 | 10938 | 11258 |
| 12 | 2.5 | 1.35 | 4.85 | 4.75 | 23148 | 2059 | 7419 | 7585 |
| 13 | 3.9 | 2.15 | 7.15 | 7.05 | 36111 | 3278 | 10938 | 11258 |
| 14 | 2.5 | 1.30 | 4.85 | 4.75 | 23148 | 1982 | 7419 | 7585 |
| 15 | 3.9 | 2.15 | 7.05 | 7.05 | 36111 | 3278 | 10785 | 11258 |
| 16 | 2.5 | 1.30 | 4.75 | 4.75 | 23148 | 1982 | 7266 | 7585 |
| 17 | 3.9 | 2.15 | 7.05 | 7.05 | 36111 | 3278 | 10785 | 11258 |
| 18 | 2.5 | 1.30 | 4.75 | 4.75 | 23148 | 1982 | 7266 | 7585 |
| 19 | 3.7 | 2.15 | 7.05 | 7.05 | 34259 | 3278 | 10785 | 11258 |
| 20 | 2.5 | 1.30 | 4.75 | 4.75 | 23148 | 1982 | 7266 | 7585 |
| 21 | 3.7 | 2.05 | 7.05 | 7.05 | 34259 | 3126 | 10785 | 11258 |
| 22 | 2.3 | 1.25 | 4.75 | 4.75 | 21296 | 1906 | 7266 | 7585 |
| 23 | 3.6 | 2.05 | 7.05 | 7.05 | 33333 | 3126 | 10785 | 11258 |

Table 26 (Cont'd)

| Cycle | Nominal Load Volts | Nominal Strain Volts | Local Strain (1) Volts | Local Strain (2) Volts | Nominal Stress $lbf/in^2$ | Nominal Strain $\mu in/in$ | Local Strain (1) $\mu in/in$ | Local Strain (2) $\mu in/in$ |
|---|---|---|---|---|---|---|---|---|
| 24 | 2.3 | 1.30 | 4.75 | 4.75 | 21296 | 1982 | 7266 | 7585 |
| 25 | 3.6 | 2.05 | 7.05 | 7.05 | 33333 | 3126 | 10785 | 11258 |
| 26 | 2.3 | 1.30 | 4.75 | 4.75 | 21296 | 1982 | 7266 | 7585 |
| 27 | 3.6 | 2.05 | 7.05 | 7.05 | 33333 | 3126 | 10785 | 11258 |
| 28 | 2.3 | 1.25 | 4.75 | 4.75 | 21296 | 1906 | 7266 | 7585 |
| 29 | 3.6 | 2.05 | 7.05 | 7.05 | 33333 | 3126 | 10785 | 11258 |
| 30 | 2.3 | 1.30 | 4.75 | 4.75 | 21296 | 1982 | 7266 | 7585 |
| 31 | 3.6 | 2.05 | 7.05 | 6.95 | 33333 | 3126 | 10785 | 11098 |
| 32 | 2.3 | 1.30 | 4.75 | 4.75 | 21296 | 1982 | 7266 | 7585 |
| 33 | 3.6 | 2.05 | 7.05 | 6.95 | 33333 | 3126 | 10785 | 11098 |
| 34 | 2.3 | 1.25 | 4.75 | 4.75 | 21296 | 1906 | 7266 | 7585 |
| 35 | 3.6 | 2.10 | 7.05 | 6.95 | 33333 | 3203 | 10785 | 11098 |
| 36 | 2.3 | 1.25 | 4.75 | 4.75 | 21296 | 1906 | 7266 | 7585 |
| 37 | 3.6 | 2.05 | 7.05 | 6.95 | 33333 | 3126 | 10785 | 11098 |
| 38 | 2.3 | 1.15 | 4.75 | 4.75 | 21296 | 1754 | 7266 | 7585 |
| 39 | 3.6 | 2.05 | 7.05 | 6.95 | 33333 | 3126 | 10785 | 11098 |
| 40 | 2.3 | 1.15 | 4.75 | 4.75 | 21296 | 1754 | 7266 | 7585 |
| 41 | 3.6 | 2.10 | 7.05 | 6.95 | 33333 | 3202 | 10785 | 11098 |
| 42 | 2.3 | 1.15 | 4.75 | 4.75 | 21296 | 1754 | 7266 | 7585 |
| 43 | 3.6 | 2.05 | 7.05 | 6.95 | 33333 | 3126 | 10785 | 11098 |
| 44 | 2.3 | 1.15 | 4.75 | 4.75 | 21296 | 1754 | 7266 | 7585 |
| 45 | 3.6 | 2.05 | 7.05 | 6.95 | 33333 | 3126 | 10785 | 11098 |
| 46 | 2.3 | 1.15 | 4.75 | 4.75 | 21296 | 1754 | 7266 | 7585 |
| 47 | 3.7 | 1.95 | 7.05 | 6.95 | 34259 | 2973 | 10785 | 11098 |
| 48 | 2.3 | 1.15 | 4.75 | 4.75 | 21296 | 1754 | 7266 | 7585 |

Table 26 (Cont'd)

| Cycle | Nominal Load Volts | Nominal Strain Volts | Local Strain (1) Volts | Local Strain (2) Volts | Nominal Stress $lbf/in^2$ | Nominal Strain $\mu in/in$ | Local Strain (1) $\mu in/in$ | Local Strain (2) $\mu in/in$ |
|---|---|---|---|---|---|---|---|---|
| 49 | 3.7 | 1.95 | 7.05 | 6.95 | 34259 | 2973 | 10785 | 11098 |
| 50 | 2.1 | 1.15 | 4.75 | 4.75 | 19444 | 1754 | 7266 | 7585 |
| 51 | 3.5 | 1.95 | 7.95 | 6.95 | 32407 | 2973 | 10785 | 11098 |
| 52 | 2.1 | 1.15 | 4.75 | 4.75 | 19444 | 1754 | 7266 | 7585 |
| 53 | 3.5 | 1.95 | 7.05 | 6.95 | 32407 | 2973 | 10785 | 11098 |
| 54 | 2.0 | 1.15 | 4.75 | 4.75 | 18519 | 1754 | 7266 | 7585 |
| 55 | 3.5 | 1.95 | 7.95 | 6.95 | 32407 | 2973 | 10785 | 11098 |
| 56 | 2.1 | 1.20 | 4.75 | 4.75 | 19444 | 1830 | 7266 | 7585 |
| 57 | 3.5 | 1.95 | 7.05 | 6.95 | 32407 | 2973 | 10785 | 11098 |
| 58 | 2.1 | 1.15 | 4.75 | 4.75 | 19444 | 1754 | 7266 | 7585 |
| 59 | 3.5 | 1.95 | 7.05 | 6.95 | 32407 | 2973 | 10785 | 11098 |
| 60 | 2.0 | 1.15 | 4.75 | 4.75 | 18519 | 1754 | 7266 | 7585 |
| 61 | 3.5 | 1.95 | 7.05 | 6.95 | 32407 | 2973 | 10785 | 11098 |
| 62 | 2.0 | 1.15 | 4.75 | 4.75 | 18519 | 1754 | 7266 | 7585 |
| 63 | 3.5 | 1.95 | 7.05 | 6.95 | 32407 | 2973 | 10785 | 11098 |
| 64 | 2.0 | 1.15 | 4.75 | 4.75 | 18519 | 1754 | 7266 | 7585 |
| 65 | 3.5 | 1.95 | 6.95 | 6.95 | 32407 | 2973 | 10632 | 11098 |
| 66 | 2.0 | 1.15 | 4.75 | 4.75 | 18519 | 1754 | 7266 | 7585 |
| 67 | 3.5 | 1.95 | 7.95 | 6.95 | 32407 | 2973 | 10785 | 11098 |
| 68 | 2.0 | 1.15 | 4.75 | 4.75 | 18519 | 1754 | 7266 | 7585 |
| 69 | 3.5 | 1.95 | 6.95 | 6.95 | 32407 | 2973 | 10632 | 11098 |
| 70 | 2.0 | 1.05 | 4.75 | 4.75 | 18519 | 1601 | 7266 | 7585 |
| 71 | 3.5 | 1.95 | 6.95 | 6.95 | 32407 | 2973 | 10632 | 11098 |
| 72 | 2.0 | 1.05 | 4.75 | 4.75 | 18519 | 1601 | 7266 | 7585 |
| 73 | 3.5 | 1.95 | 6.95 | 6.95 | 32407 | 2973 | 10632 | 11098 |

Table 26 (Cont'd)

| Cycle | Nominal Load Volts | Nominal Strain Volts | Local Strain (1) Volts | Local Strain (2) Volts | Nominal Stress $lbf/in^2$ | Nominal Strain $\mu in/in$ | Local Strain (1) $\mu in/in$ | Local Strain (2) $\mu in/in$ |
|---|---|---|---|---|---|---|---|---|
| 74 | 2.0 | 1.05 | 4.75 | 4.75 | 18519 | 1601 | 7266 | 7585 |
| 75 | 3.5 | 1.95 | 6.95 | 6.95 | 32407 | 2973 | 10632 | 11098 |
| 76 | 2.0 | 1.05 | 4.75 | 4.75 | 18519 | 1601 | 7266 | 7585 |
| 77 | 3.5 | 1.90 | 6.95 | 6.95 | 32407 | 2897 | 10632 | 11098 |
| 78 | 2.0 | 1.05 | 4.75 | 4.75 | 18519 | 1601 | 7266 | 7585 |
| 79 | 3.5 | 1.90 | 6.95 | 6.95 | 32407 | 2897 | 10632 | 11098 |
| 80 | 2.0 | 1.05 | 4.75 | 4.75 | 18519 | 1601 | 7266 | 7585 |
| 81 | 3.5 | 1.85 | 6.95 | 6.95 | 32407 | 2821 | 10632 | 11098 |
| 82 | 2.0 | 1.05 | 4.75 | 4.75 | 18519 | 1601 | 7266 | 7585 |
| 83 | 3.5 | 1.85 | 6.95 | 6.95 | 32407 | 2821 | 10632 | 11098 |
| 84 | 2.0 | 1.05 | 4.75 | 4.75 | 18519 | 1601 | 7266 | 7585 |
| 85 | 3.3 | 1.90 | 6.95 | 6.95 | 30556 | 2897 | 10632 | 11098 |
| 86 | 2.0 | 1.05 | 4.75 | 4.75 | 18519 | 1601 | 7266 | 7585 |
| 87 | 3.3 | 1.90 | 6.95 | 6.95 | 30556 | 2897 | 10632 | 11098 |
| 88 | 1.9 | 1.10 | 4.75 | 4.75 | 17593 | 1677 | 7266 | 7585 |
| 89 | 3.3 | 1.85 | 6.95 | 6.95 | 30556 | 2821 | 10632 | 11098 |
| 90 | 1.9 | 1.10 | 4.75 | 4.75 | 17593 | 1677 | 7266 | 7585 |
| 91 | 3.3 | 1.85 | 6.95 | 6.95 | 30556 | 2821 | 10632 | 11098 |
| 92 | 1.9 | 1.05 | 4.65 | 4.75 | 17593 | 1601 | 7113 | 7585 |
| 93 | 3.3 | 1.90 | 6.95 | 6.95 | 30556 | 2897 | 10632 | 11098 |
| 94 | 1.9 | 1.05 | 4.65 | 4.75 | 17593 | 1601 | 7113 | 7585 |
| 95 | 3.3 | 1.85 | 6.95 | 6.95 | 30556 | 2821 | 10632 | 11098 |
| 96 | 1.9 | 1.05 | 4.65 | 4.75 | 17593 | 1601 | 7113 | 7585 |
| 97 | 3.2 | 1.85 | 6.95 | 6.95 | 29630 | 2821 | 10632 | 11098 |
| 98 | 1.9 | 1.05 | 4.70 | 4.75 | 17593 | 1601 | 7190 | 7585 |

Table 26 (Cont'd)

| Cycle | Nominal Load Volts | Nominal Strain Volts | Local Strain (1) Volts | Local Strain (2) Volts | Nominal Stress lbf/in$^2$ | Nominal Strain $\mu$in/in | Local Strain (1) $\mu$in/in | Local Strain (2) $\mu$in/in |
|---|---|---|---|---|---|---|---|---|
| 99 | 3.3 | 1.90 | 6.95 | 6.90 | 30556 | 2897 | 10632 | 11018 |
| 100 | 1.9 | 1.05 | 4.65 | 4.75 | 17593 | 1601 | 7113 | 7585 |
| 101 | 3.20 | 1.85 | 6.95 | 6.85 | 29630 | 2821 | 10632 | 10938 |
| 102 | 1.9 | 1.05 | 4.65 | 4.75 | 17593 | 1601 | 7113 | 7585 |
| 103 | 3.3 | 1.90 | 6.95 | 6.85 | 30556 | 2897 | 10632 | 10938 |
| 104 | 1.9 | 1.00 | 4.65 | 4.75 | 17593 | 1525 | 7113 | 7585 |
| 105 | 3.3 | 1.85 | 6.95 | 6.85 | 30556 | 2821 | 10632 | 10938 |
| 106 | 1.9 | 0.95 | 4.65 | 4.75 | 17593 | 1449 | 7113 | 7585 |
| 107 | 3.3 | 1.85 | 6.95 | 6.85 | 30556 | 2821 | 10632 | 10938 |
| 108 | 1.9 | 0.95 | 4.65 | 4.75 | 17593 | 1449 | 7113 | 7585 |
| 109 | 3.3 | 1.85 | 6.95 | 6.85 | 30556 | 2821 | 10632 | 10938 |
| 110 | 1.9 | 0.95 | 4.65 | 4.75 | 17593 | 1449 | 7113 | 7585 |
| 111 | 3.3 | 1.85 | 6.85 | 6.85 | 30556 | 2821 | 10479 | 10938 |
| 112 | 1.9 | 0.95 | 4.65 | 4.75 | 17593 | 1449 | 7113 | 7585 |
| 113 | 3.2 | 1.75 | 6.85 | 6.85 | 29630 | 2668 | 10479 | 10938 |
| 114 | 1.9 | 0.95 | 4.65 | 4.75 | 17593 | 1449 | 7113 | 7585 |

CALIBRATION

Load: 1.0 VDC = 10,000 lbf

Strain: Strain Gage (1) 1.0 VDC = 1529.76 $\mu$in/in

Strain Gage (2) 1.0 VDC = 1596.83 $\mu$in/in

Stress: 1.0 VDC = 9259.26 lbf/in$^2$

TABLE 27

Data for calculation of local stress for stress relaxation behavior of dual amplitude cyclic loading test on a plate on high stress cycles.

| | | Mono. Data | | DAL Data | |
|---|---|---|---|---|---|
| Cycle | Nominal Stress $lbf/in^2$ | Local Stress(1) $lbf/in^2$ | Local Stress(2) $lbf/in^2$ | Local Stress(1) $lbf/in^2$ | Local Stress(2) $lbf/in^2$ |
| 1 | 36111 | 78000 | 78000 | 78000 | 78000 |
| 5 | 36111 | 78000 | 78000 | 78000 | 78000 |
| 9 | 36111 | 78000 | 78000 | 78000 | 78000 |
| 13 | 36111 | 78000 | 78000 | 78000 | 78000 |
| 17 | 36111 | 78000 | 78000 | 78000 | 78000 |
| 21 | 34259 | 72277 | 72129 | 72537 | 72388 |
| 25 | 33333 | 69416 | 69194 | 69805 | 69853 |
| 29 | 33333 | 69416 | 69194 | 69805 | 69853 |
| 33 | 33333 | 69416 | 69194 | 69805 | 69853 |
| 37 | 33333 | 69416 | 69194 | 69805 | 69853 |
| 41 | 33333 | 69416 | 69194 | 69805 | 69853 |
| 45 | 33333 | 69416 | 69194 | 69805 | 69853 |
| 49 | 34259 | 72277 | 72129 | 72537 | 72388 |
| 53 | 32407 | 56555 | 66258 | 67073 | 66777 |
| 57 | 32407 | 56555 | 66258 | 67073 | 66777 |
| 61 | 32407 | 56555 | 66258 | 67073 | 66777 |
| 65 | 32407 | 56555 | 66258 | 67073 | 66777 |
| 69 | 32407 | 56555 | 66258 | 67073 | 66777 |
| 73 | 32407 | 56555 | 66258 | 67073 | 66777 |
| 77 | 32407 | 56555 | 66258 | 67073 | 66777 |
| 81 | 32407 | 56555 | 66258 | 67073 | 66777 |
| 85 | 30556 | 60835 | 60391 | 61613 | 61168 |
| 89 | 30556 | 60835 | 60391 | 61613 | 61168 |
| 93 | 30556 | 60835 | 60391 | 61613 | 61168 |
| 97 | 29630 | 57974 | 57455 | 58881 | 58363 |
| 101 | 29630 | 57974 | 57455 | 58881 | 58363 |
| 105 | 30556 | 60835 | 60391 | 61613 | 61168 |
| 109 | 30556 | 60835 | 60391 | 61613 | 61168 |
| 113 | 29630 | 57974 | 57455 | 58881 | 58363 |

Sm = 36111 $lbf/in^2$
$\sigma$m = 78,000 $lbf/in^2$
Mono. - Monotonic data from cyclic stress-strain curve test.
E = $10.67 \times 10^6$ $lbf/in^2$
DAL - Dual amplitude cyclic loading test data.
E = $10.19 \times 10^6$ $lbf/in^2$

TABLE 28

Data for calculation of local stress for stress relaxation behavior of dual amplitude cyclic loading test on a plate on low stress cycles.

| Cycle | Nominal Stress lbf/in$^2$ | Mono. Data Local Stress(1) lbf/in$^2$ | Mono. Data Local Stress(2) lbf/in$^2$ | DAL Data Local Stress(1) lbf/in$^2$ | DAL Data Local Stress(2) lbf/in$^2$ |
|---|---|---|---|---|---|
| 2 | 23148 | 37944 | 36907 | 39759 | 38722 |
| 6 | 23148 | 37944 | 36907 | 39759 | 38722 |
| 10 | 23148 | 37944 | 36907 | 39759 | 38772 |
| 14 | 23148 | 37944 | 36907 | 39759 | 38772 |
| 18 | 23148 | 37944 | 36907 | 39759 | 38772 |
| 22 | 21296 | 32222 | 31036 | 34296 | 33111 |
| 26 | 21296 | 32222 | 31306 | 34296 | 33111 |
| 30 | 21296 | 32222 | 31036 | 34296 | 33111 |
| 34 | 21296 | 32222 | 31036 | 34296 | 33111 |
| 38 | 21296 | 32222 | 31036 | 34296 | 33111 |
| 42 | 21296 | 32222 | 31036 | 34296 | 33111 |
| 46 | 21296 | 32222 | 31036 | 34296 | 33111 |
| 50 | 19444 | 26499 | 25166 | 28832 | 27499 |
| 54 | 18519 | 23641 | 22233 | 26104 | 24696 |
| 58 | 19444 | 26499 | 25166 | 28832 | 27499 |
| 62 | 18519 | 23641 | 22233 | 26104 | 24696 |
| 66 | 18519 | 23641 | 22233 | 26104 | 24696 |
| 70 | 18519 | 23641 | 22233 | 26104 | 24696 |
| 74 | 18519 | 23641 | 22233 | 26104 | 24696 |
| 78 | 18519 | 23641 | 22233 | 26104 | 24696 |
| 82 | 18519 | 23641 | 22233 | 26104 | 24696 |
| 86 | 18519 | 23641 | 22233 | 25104 | 24696 |
| 90 | 17593 | 20779 | 19298 | 23372 | 21890 |
| 94 | 17593 | 20779 | 19298 | 23372 | 21890 |
| 98 | 17593 | 20779 | 19298 | 23372 | 21890 |
| 102 | 17593 | 20779 | 19298 | 23372 | 21890 |
| 106 | 17593 | 20779 | 19298 | 23372 | 21890 |
| 110 | 17593 | 20779 | 19298 | 23372 | 21890 |
| 114 | 17593 | 20779 | 19298 | 23372 | 21890 |

Sm = 78,000 lbf/in$^2$
$\sigma$m = 36111 lbf/in$^2$
Mono. - Monotonic data from cyclic stress-strain curve test.
E = 10.67 x 10$^6$ lbf/in$^2$
DAL - Dual amplitude cyclic loading test data.
E = 10.19 x 10$^6$ lbf/in$^2$

TABLE 29

Initial stress and stress relaxation rate parameter data from uniaxial and plate cyclic loading tests.

| Initial Stress $lbf/in^2$ | Relaxation Rate Parameter | Type of Test | | |
|---|---|---|---|---|
| | | **Plate** | | |
| | | Data from uniaxiam specimen cyclic stress-strain curve test | | |
| 79220 | 3.427 | SAL test | $K_t$=2.85 | Strain Gage (1) |
| 79470 | 3.752 | | $K_t$=2.95 | Strain Gage (2) |
| 78080 | 2.593 | DAL test | $K_t$=3.09 | Strain Gage (1) |
| 78100 | 2.671 | (high) | $K_t$=3.17 | Strain Gage (2) |
| 39030 | 6.350 | DAL test | $K_t$=3.09 | Strain Gage (1) |
| 38120 | 6.843 | (low) | $K_t$=3.17 | Strain Gage (2) |
| | | Data from uniaxiam specimen single amplitude test | | |
| 79470 | 3.324 | | $K_t$=2.66 | Strain Gage (1) |
| 79470 | 3.470 | | $K_t$=2.76 | Strain Gage (2) |
| | | Data from uniaxial specimen dual amplitude tests | | |
| 78060 | 2.457 | (High) | $K_t$=2.95 | Strain Gage (1) |
| 78070 | 2.534 | (High) | $K_t$=3.03 | Strain Gage (2) |
| 70650 | 5.594 | (Low) | $K_t$=2.95 | Strain Gage (1) |
| 39720 | 6.010 | (Low) | $K_t$=3.03 | Strain Gage (2) |
| | | **Uniaxial Specimen** | | |
| 73160 | 3.177 | SAL | | |
| 76930 | 3.168 | DAL (High) | | |
| 45600 | 7.572 | DAL (Low) | | |

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